

NASA  
N62-15296  
\$1.75

# TECHNICAL NOTE

D-1393

A TABULATION OF WIND-TUNNEL PRESSURE DATA AND  
SECTION AERODYNAMIC CHARACTERISTICS AT MACH NUMBERS  
OF 1.61 AND 2.01 FOR A REFLEX CAMBERED WING AND A  
CAMBERED AND TWISTED WING HAVING  
THE SAME SWEPT PLANFORM

By Emma Jean Landrum

Langley Research Center  
Langley Station, Hampton, Va.

NATIONAL AERONAUTICS AND SPACE ADMINISTRATION  
WASHINGTON

September 1962

## NATIONAL AERONAUTICS AND SPACE ADMINISTRATION

## TECHNICAL NOTE D-1393

A TABULATION OF WIND-TUNNEL PRESSURE DATA AND  
SECTION AERODYNAMIC CHARACTERISTICS AT MACH NUMBERS  
OF 1.61 AND 2.01 FOR A REFLEX CAMBERED WING AND A  
CAMBERED AND TWISTED WING HAVING  
THE SAME SWEPT PLANFORM

By Emma Jean Landrum

## SUMMARY

The pressure, section normal-force, and section pitching-moment coefficients for a cambered and twisted wing and a reflex cambered wing both having the same swept planform are tabulated. Both wings had an NACA 65A005 thickness distribution,  $50^\circ$  sweepback of the quarter chord, a taper ratio of 0.20, and an aspect ratio of 3.5. The cambered and twisted wing had at each spanwise station an  $a = 0$  mean line modified to have a maximum height of 4-percent chord and a linear spanwise twist variation with  $6^\circ$  of washout at the tip. The reflex cambered wing had a 1-wave-length sinusoidal mean line with a leading-edge angle of attack of  $-6^\circ$ . The wings were tested at Mach numbers of 1.61 and 2.01 with fixed transition at Reynolds numbers of  $3.6 \times 10^6$  and  $3.1 \times 10^6$ , respectively, through an angle-of-attack range from  $-20^\circ$  to  $20^\circ$ . For convenience, previously published data for a flat wing, a cambered wing, and a linearly twisted wing of the same planform as the present wings have been included.

## INTRODUCTION

The prediction of the changes in aerodynamic characteristics of wings when they distort under variable flight loads is of considerable interest in the design of efficient wings for supersonic aircraft. As part of a general investigation at low supersonic speeds of the effects of arbitrary camber and twist, a series of sweptback wings having the same planform but with systematic variations in surface shape has been tested. The tabulated results of a pressure investigation of the separate effects of camber and twist on the aerodynamic characteristics of sweptback wings at Mach numbers of 1.61 and 2.01 are presented in

reference 1, and a limited analysis of some of these results is presented in reference 2. The results of a force study of the same wings are given in reference 3. The section normal-force and section pitching-moment coefficients for the flat and twisted wings of reference 1 are given in reference 4.

The purpose of this report is to present the pressure, section normal-force, and section pitching-moment coefficients obtained at Mach numbers of 1.61 and 2.01 for two additional swept wings of the same planform, one with camber and twist and one with reflex camber. For convenience, some of the previously published data for a flat wing, a cambered wing, and a linearly twisted wing of the same planform as the present wings have been included. No analysis of the data has been made.

### SYMBOLS

$b/2$	semispan
$C_p$	pressure coefficient
$c$	wing chord
$c_n$	section normal-force coefficient
$c_m$	section pitching-moment coefficient (taken about midchord of wing mean aerodynamic chord)
$M$	Mach number
$x$	chordwise distance from wing leading edge
$y$	spanwise distance from wing root chord
$\alpha$	angle of attack of wing root, deg

### MODELS AND MODEL MOUNTING

Two semispan sweptback wings were tested, one with camber and twist and one with reflex camber. Both wings had an NACA 65A005 thickness distribution,  $50^\circ$  sweepback of the quarter chord, a taper ratio of 0.20, and an aspect ratio of 3.5. The cambered and twisted wing had at each spanwise station an  $a = 0$  mean line modified to have a maximum height of 4-percent chord and a linear spanwise twist variation with  $6^\circ$  of

washout at the tip. The reflex cambered wing had a 1-wave-length sinusoidal mean line with a leading-edge angle of attack of  $-6^{\circ}$ . A plan view of the models is shown in figure 1.

Each wing had seven streamwise rows of orifices located at 0.05, 0.20, 0.35, 0.50, 0.70, 0.825, and 0.95 semispan. Orifices were located on both surfaces of the wings. Chordwise orifice locations for the cambered and twisted wing are given in table I in terms of a nominal location. For the reflex cambered wing the chordwise orifice locations are 0.0125, 0.025, 0.050, 0.075, 0.100, 0.150, 0.200, 0.250, 0.300, 0.400, 0.500, 0.600, 0.700, 0.800, and 0.900 local chord at 0.050, 0.200, 0.350, and 0.500 semispan and are 0.025, 0.075, 0.150, 0.250, 0.350, 0.450, 0.550, 0.650, 0.750, and 0.850 local chord at 0.700, 0.825, and 0.950 semispan.

The semispan wings were mounted horizontally in the tunnel from a turntable in a boundary-layer bypass plate which was located vertically in the test section about 10 inches from the tunnel wall.

### TESTS AND TEST PROCEDURES

The tests were conducted in the Langley 4- by 4-foot supersonic pressure tunnel at Mach numbers of 1.61 and 2.01. Transition was fixed about  $1/2$  inch from the wing leading edge by No. 60 carborundum grains.

Angle of attack was changed through a range from  $-20^{\circ}$  to  $20^{\circ}$  by manual rotation of the turntable on which the models were mounted and was measured by a vernier scale outside the tunnel. The tests were conducted at a tunnel stagnation pressure of 15 pounds per square inch absolute which provided Reynolds numbers of  $3.6 \times 10^6$  and  $3.1 \times 10^6$  at Mach numbers of 1.61 and 2.01, respectively.

### PRESENTATION OF RESULTS

For the convenience of the user of the data for the cambered and twisted wing and the reflex cambered wing, tables at the same Reynolds number and transition condition from previously published reports for a flat wing, a cambered wing, and a linearly twisted wing have been included in this report. These wings have the same planform as those of the present tests. The cambered wing has the same camber as the cambered and twisted wing and the linearly twisted wing has the same spanwise twist variation. The flat wing is designated wing F in references 1, 3, and 4; the cambered wing is designated wing C in references 1 and 3; the twisted wing is designated wing 1 in references 1, 3, and 4.

The pressure coefficients are presented in tables II to VI and the section normal-force and section pitching-moment coefficients are given in tables VII to XI. Data are presented in tables II and VII for the cambered and twisted wing, in tables III and VIII for the reflex cambered wings, in tables IV (from ref. 1) and IX (from ref. 4) for the flat wing, in tables V (from ref. 1) and X for the cambered wings, and in tables VI (from ref. 1) and XI (from ref. 4) for the linearly twisted wing.

Langley Research Center,  
National Aeronautics and Space Administration,  
Langley Station, Hampton, Va., June 18, 1962.

L  
3  
0  
0  
0

#### REFERENCES

1. Grant, Frederick C.: A Tabulation of Wind-Tunnel Pressure Data at Mach Numbers of 1.61 and 2.01 for Five Swept Wings of the Same Plan Form but Different Surface Shapes. NACA RM L58D23, 1958.
2. Grant, Frederick C., and Mugler, John P., Jr.: Span Loadings Due to Wing Twist at Transonic and Supersonic Speeds. NACA RM L57D24a, 1957.
3. Landrum, Emma Jean, and Czarnecki, K. R.: Effects at Mach Numbers of 1.61 and 2.01 of Camber and Twist on the Aerodynamic Characteristics of Three Swept Wings Having the Same Planform. NASA TN D-929, 1961.
4. Landrum, Emma Jean: A Tabulation of Section Aerodynamic Characteristics at Mach Numbers of 1.61 and 2.01 for Four Swept Wings Having the Same Planform but Different Surface Shapes. NASA TN D-1244, 1962.

TABLE I.- CHORDWISE ORIFICE LOCATIONS FOR CAMBERED AND TWISTED WING

x/c, nominal	x/c at $y/b$ of:													
	0.05		0.20		0.35		0.50		0.70		0.825		0.95	
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower
0.0125	0.0125	0.0125	0.0125	0.0125	0.0125	0.0125	0.0125	0.0125						
.025	.0250	.0250	.0250	.0250	.0250	.0250	.0250	.0250	0.0250	0.0250	0.0250	0.0250	0.0250	0.0250
.050	.0500	.0500	.0500	.0500	.0500	.0500	.0500	.0500						
.075	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750	.0750
.100	.1000	.1000	.1000	.1000	.1000	.1000	.1000	.1000						
.150	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500	.1500
.200	.2000	.2000	.2000	.2000	.2000	.2000	.2000	.2000	.1500	.1500	.1500	.1500	.1500	.1500
.250	.2500	.2500	.2500	.2500	.2500	.2500	.2500	.2500	.2445	.2445	.2500	.2500	.2500	.2500
.300	.3000	.3000	.3000	.3000	.3000	.3000	.3000	.3012	.3533	.3533	.3500	.3500	.3500	.3500
.350														
.400	.4000	.3988	.4000	.4000	.4000	.4000	.4000	.4000	.4500	.4500	.4535	.4535	.4500	.4500
.450														
.500	.5000	.5000	.5000	.5000	.5000	.5000	.5000	.5070	.5579	.5579	.5500	.5500	.5500	.5500
.550														
.600	.6000	.6000	.6000	.6000	.6000	.6000	.6000	.6000	.6500	.6500	.6500	.6500	.6500	.6500
.650														
.700	.7000	.7000	.7000	.7000	.7000	.7000	.7000	.7000	.7500	.7500	.7500	.7500	.7500	.7500
.750														
.800	.8000	.8000	.8000	.8000	.8000	.8000	.8000	.8000	.8500	.8500	.8500	.8500	.8500	.8500
.850														
.900	.9000	.9000	.9000	.9000	.9000	.9000	.9000	.9000						

TABLE II  
PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

x/c, nominal	Cp at $y/\frac{1}{2}$ of:																x/c, nominal
	.05		.20		.35		.50		.70		.825		.95				
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower			
$\alpha = -20$																	
.0125	.810	-.483	.680	-.483	.625	-.477	.572	-.464									.0125
.025	.831	-.489	.722	-.483	.674	-.479	.611	-.467									.025
.050	.785	-.504	.716	-.479	.680	-.477		-.465									.050
.075	.738	-.497	.690	-.484	.666	-.478	.640		.629		.561	-.415	.682	-.235			.075
.100	.684	-.493	.636	-.484	.619	-.473	.609		-.443								.100
.150	.604	-.473	.587	-.487	.587	-.479		-.468	.571	-.463	.533	-.386	.676	-.247	.150		
.200	.550	-.384	.554	-.491	.536	-.479	.563	-.469									.200
.250	.544	-.301	.482	-.490	.494	-.483	.520	-.467	.520	-.466	.508	-.377	.657	-.231	.250		
.300	.486	-.285	.444	-.485	.455	-.485	.474	-.467									.300
.350									.438	-.463	.442	-.371	.613	-.241	.350		
.400	.437	-.228	.372	-.499	.385	-.486	.396	-.473									.400
.450									.372	-.444	.423	-.360	.566	-.289	.450		
.500	.344	-.233	.329	-.487	.328	-.486	.333	-.468	.348	-.434	.472	-.367	.527	-.325	.500		
.550																	.550
.600	.306	-.226	.274	-.473	.307	-.479	.319	-.456	.343	-.439	.602	-.371	.480	-.314	.600		
.650																	.650
.700	.264	-.234	.275	-.435	.280	-.469	.308	-.449									.700
.750									.385	-.365	.558	-.369	.408	-.247	.750		
.800	.258	-.212	.275	-.360	.300	-.467	.328	-.462	.546	-.434	.508	-.375	.372	-.249	.800		
.850																	.850
.900	.296	-.238			.332	-.461	.383										.900
.950																	.950
$\alpha = -18$																	
.0125	.781	-.479	.668	-.484	.613	-.478	.572	-.468									.0125
.025	.793	-.490	.696	-.486	.650	-.478	.587	-.468									.025
.050	.734	-.506	.684	-.480	.644	-.476		-.468	.569	-.461	.519	-.454	.505	-.271	.050		
.075	.684	-.505	.651	-.487	.624	-.478	.598		.590		.534	-.441	.558	-.252	.075		
.100	.633	-.509	.595	-.488	.571	-.474	.561		-.444								.100
.150	.553	-.462	.536	-.494	.540	-.478		-.470	.521	-.461	.474	-.425	.508	-.233	.150		
.200	.496	-.354	.498	-.498	.480	-.479	.468	-.470									.200
.250	.490	-.254	.425	-.496	.434	-.483	.447	-.470	.453	-.463	.440	-.398	.463	-.212	.250		
.300	.437	-.242	.387	-.485	.387	-.487	.399	-.469									.300
.350									.361	-.463	.362	-.378	.411	-.195	.350		
.400	.384	-.199	.316	-.508	.306	-.500	.328	-.477	.291	-.444	.328	-.364	.372	-.182	.400		
.450									.291	-.444	.328	-.364	.372	-.182	.450		
.500	.293	-.209	.262	-.501	.255	-.497	.252	-.480	.259	-.430	.299	-.368	.350	-.181	.500		
.550																	.550
.600	.253	-.201	.203	-.464	.219	-.486	.229	-.471	.260	-.433	.294	-.393	.420	-.181	.600		
.650									.260	-.433	.294	-.393	.420	-.181	.650		
.700	.203	-.215	.196	-.363	.199	-.474	.223	-.462	.265	-.373	.310	-.410	.438	-.212	.700		
.750									.265	-.373	.310	-.410	.438	-.212	.750		
.800	.187	-.194	.184	-.262	.205	-.468	.241	-.451	.301	-.419	.354	-.407	.392	-.244	.800		
.850																	.850
.900	.216	-.219	.232	-.224	.249	-.472	.294										.900
.950																	.950
$\alpha = -16$																	
.0125	.751	-.474	.654	-.476	.606	-.466	.569	-.457									.0125
.025	.745	-.480	.665	-.476	.634	-.466	.573	-.457									.025
.050	.676	-.503	.644	-.471	.609	-.467		-.459	.568	-.446	.520	-.451	.499	-.295	.050		
.075	.629	-.507	.601	-.476	.563	-.467	.557		.569		.514	-.444	.523	-.295	.075		
.100	.579	-.507	.543	-.480	.528	-.468	.518	-.437									.100
.150	.496	-.431	.482	-.493	.488	-.472		-.461	.483	-.445	.440	-.432	.459	-.285	.150		
.200	.438	-.275	.397	-.492	.432	-.476	.415	-.463									.200
.250	.431	-.188	.373	-.487	.387	-.483	.392	-.463	.415	-.449	.399	-.402	.399	-.268	.250		
.300	.389	-.211	.333	-.472	.336	-.484	.341	-.459									.300
.350									.313	-.452	.309	-.381	.335	-.247	.350		
.400	.332	-.178	.264	-.494	.254	-.493	.270	-.467	.241	-.434	.259	-.367	.297	-.230	.400		
.450									.241	-.434	.259	-.367	.297	-.230	.450		
.500	.245	-.188	.211	-.486	.201	-.496	.200	-.473	.200	-.421	.223	-.367	.259	-.210	.500		
.550									.200	-.421	.223	-.367	.259	-.210	.550		
.600	.205	-.178	.160	-.391	.175	-.490	.170	-.468	.185	-.423	.221	-.388	.239	-.210	.600		
.650									.185	-.423	.221	-.388	.239	-.210	.650		
.700	.160	-.198	.150	-.231	.143	-.480	.154	-.462	.192	-.380	.229	-.397	.210	-.218	.700		
.750									.192	-.380	.229	-.397	.210	-.218	.750		
.800	.139	-.182	.135	-.192	.134	-.467	.166	-.439	.219	-.388	.257	-.414	.224	-.233	.800		
.850																	.850
.900	.168	-.202	.176	-.192	.171	-.477	.207										.900
.950																	.950
$\alpha = -14$																	
.0125	.721	-.470	.629	-.466	.589	-.450	.566	-.440									.0125
.025	.698	-.475	.632	-.467	.617	-.448	.554	-.435									.025
.050	.620	-.498	.597	-.460	.573	-.449		-.439	.556	-.435	.507	-.436	.493	-.288	.050		
.075	.575	-.507	.542	-.461	.536	-.452	.520		.531		.479	-.434	.502	-.287	.075		
.100	.521	-.507	.482	-.469	.476	-.455	.478	-.424									.100
.150	.433	-.374	.430	-.490	.441	-.458		-.444	.430	-.440	.392	-.424	.419	-.280	.150		
.200	.379	-.207	.344	-.490	.377	-.472	.361	-.446									.200
.250	.363	-.128	.320	-.482	.334	-.487	.340	-.446	.362	-.404	.344	-.402	.352	-.260	.250		
.300	.337	-.190	.281	-.459	.286	-.487	.290	-.456									.300
.350									.249	-.444	.255	-.374	.278	-.244	.350		
.400	.282	-.160	.215	-.476	.202	-.485	.214	-.470	.177	-.439	.202	-.359	.238	-.231	.400		
.450									.177	-.439	.202	-.359	.238	-.231	.450		
.500	.194	-.168	.163	-.438	.147	-.485	.141	-.480	.149	-.422	.162	-.353	.192	-.223	.500		
.550									.149	-.422	.162	-.353	.192	-.223	.550		
.600	.157	-.156	.118	-.218	.127	-.487	.124	-.475	.126	-.422	.152	-.369	.171	-.217	.600		
.650									.126	-.422	.152	-.369	.171	-.217	.650		
.700	.115	-.180	.103	-.158	.088	-.475	.100	-.470	.120	-.398	.155	-.368	.126	-.223	.700		
.750									.120	-.398	.155	-.368	.126	-.223	.750		
.800	.088	-.169	.090	-.164	.108	-.457	.109	-.459	.139	-.385	.171	-.365	.147	-.233	.800		
.850																	.850
.900	.118	-.183	.124	-.170	.110	-.461	.144										.900
.950																	.950





TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(a) M = 1.61 - Continued

x/e, nominal	Cp at y/b = 0.5 of:																x/e, nominal
	.05		.20		.35		.50		.70		.825		.95				
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower			
α = -0.4																	
.0125	.554	-.416	.516	-.361	.508	-.313	.511	-.297									.0125
.025	.454	-.417	.459	-.361	.449	-.311	.410	-.296									.025
.050	.357	-.447	.368	-.358	.359	-.312		-.300									.050
.075	.302	-.398	.292	-.374	.303	-.318	.304		.340			.328	-.302	.341	-.258		.075
.100	.256	-.165	.232	-.385	.217	-.319	.243	-.293									.100
.150	.172	-.057	.161	-.384	.176	-.344		.300	.204	-.288	.185	-.299	.231	-.250			.150
.200	.134	-.030	.077	-.371	.112	-.357	.111	-.314									.200
.250	.108	.009	.056	-.348	.083	-.355	.090	-.317	.122	-.290	.112	-.304	.117	-.233			.250
.300	.088	-.074	.032	-.301	.034	-.349	.035	-.326									.300
.350									.011	-.298	.023	-.297	.020	-.215			.350
.400	.051	-.022	-.022	.047	-.038	-.315	-.028	-.321									.400
.450									-.060	-.309	-.040	-.285	-.018	-.199			.450
.500	-.008	-.030	-.058	.010	-.097	-.281	-.102	-.309	-.113	-.315	-.097	-.278	-.058	-.189			.500
.550																	.550
.600	-.022	-.027	-.087	-.015	-.109	-.199	-.116	-.285	-.132	-.308	-.117	-.268	-.101	-.181			.600
.650																	.650
.700	-.064	-.053	-.100	-.044	-.126	.021	-.128	-.267	-.136	-.320	-.121	-.256	-.149	-.173			.700
.750																	.750
.800	-.084	-.070	-.098	-.049	-.112	-.008	-.129	-.248	-.132	-.270	-.113	-.253	-.149	-.173			.800
.850																	.850
.900	-.062	-.062	-.085	-.066	-.102	-.030	-.102										.900
.950																	.950
α = -0.2																	
.0125	.512	-.398	.478	-.331	.478	-.278	.482	-.255									.0125
.025	.399	-.397	.415	-.329	.427	-.274	.376	-.253									.025
.050	.309	-.413	.316	-.327	.320	-.274		-.254		.425	-.242	.408	-.258	.412	-.248		.050
.075	.253	-.354	.248	-.343	.254	-.278	.258		.300			.288	-.262	.298	-.246		.075
.100	.210	-.039	.175	-.354	.175	-.286	.195	-.251									.100
.150	.130	-.036	.112	-.337	.136	-.308		-.256	.160	-.248	.138	-.262	.181	-.245			.150
.200	.094	-.006	.035	-.319	.065	-.318	.061	-.270									.200
.250	.068	.032	.011	-.291	.041	-.312	.053	-.274	.077	-.248	.064	-.262					

TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

x/c, nominal	Cp at y/b <sub>2</sub> of:														x/c, nominal
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
α = .04															
.0125	.385	-.311	.377	-.202	.375	-.121	.379	-.080							.0125
.025	.284	-.302	.293	-.205	.298	-.117	.276	-.078							.025
.050	.166	-.233	.166	-.200	.179	-.113		-.077	.328	-.050	.317	-.070	.337	-.082	.050
.075	.115	-.004	.101	-.215	.110	-.119	.110		.157		.161	-.077	.163	-.086	.075
.100	.075	.043	.042	-.190	.042	-.129	.059	-.081							.100
.150	.021	.061	-.026	-.148	-.020	-.143		-.083	.012	-.056	.006	-.073	.052	-.088	.150
.200	-.015	.095	-.100	-.057	-.089	-.135	-.089	-.096							.200
.250	-.027	.105	-.115	.153	-.103	-.109	-.093	-.079	-.073	-.058	-.073	-.077	-.063	-.093	.250
.300	-.063	.110	-.134	.193	-.154	-.073	-.154	-.055							.300
.350									-.167	-.058	-.166	-.058	-.148	-.096	.350
.400	-.096	.131	-.176	.144	-.210	.104	-.214	-.012	-.230	-.049	-.209	-.043	-.196	-.089	.400
.450															.450
.500	-.131	.115	-.200	.134	-.234	.187	-.264	.018	-.278	-.039	-.252	-.044	-.233	-.078	.500
.550	-.157	.110	-.211	.109	-.269	.140	-.274	.052	-.289	-.009	-.274	-.041	-.267	-.057	.550
.600															.600
.650	-.193	.079	-.214	.091	-.271	.103	-.278	.088	-.293	-.044	-.280	-.038	-.288	-.072	.650
.700															.700
.750	-.173	.040	-.216	.073	-.249	.072	-.281	.084	-.292	-.001	-.281	-.049	-.261	-.091	.750
.800															.800
.850															.850
.900	-.167	.050	-.194	.062	-.232	.055	-.278								.900
.950															.950
α = .06															
.0125	.347	-.266	.344	-.149	.341	-.056	.341	-.006							.0125
.025	.244	-.253	.262	-.153	.251	-.054	.229	-.004	.290	.037	.283	.016	.298	.002	.025
.050	.124	-.133	.120	-.149	.132	-.049		-.003							.050
.075	.078	.043	.060	-.157	.061	-.053	.064		.126		.129	.009	.115	-.002	.075
.100	.041	.076	.003	-.122	.001	-.064	.008	-.011							.100
.150	-.012	.089	-.062	-.052	-.064	-.069		-.009	-.020	.035	-.024	.015	.008	-.005	.150
.200	-.040	.131	-.129	.116	-.121	-.054	-.129	-.009							.200
.250	-.040	.131	-.165	.217	-.145	-.002	-.143	.023	-.098	.033	-.108	.013	-.099	-.012	.250
.300	-.093	.170	-.168	.213	-.202	.067	-.196	.063							.300
.350									-.201	.059	-.196	.039	-.178	.004	.350
.400	-.123	.175	-.206	.180	-.248	.244	-.253	.107	-.255	.078	-.237	.058	-.227	.032	.400
.450															.450
.500	-.153	.162	-.234	.174	-.273	.198	-.303	.127	-.298	.082	-.278	.061	-.265	.042	.500
.550															.550
.600	-.177	.148	-.237	.146	-.291	.157	-.314	.147	-.313	.099	-.297	.067	-.298	.041	.600
.650															.650
.700	-.209	.119	-.240	.126	-.317	.132	-.323	.157	-.318	.062	-.305	.071	-.308	.026	.700
.750															.750
.800	-.191	.076	-.238	.109	-.288	.105	-.314	.130	-.318	.086	-.306	.066	-.281	-.011	.800
.850															.850
.900	-.187	.090	-.214	.095	-.262	.090	-.304								.900
.950															.950
α = .08															
.0125	.298	-.229	.296	-.091	.294	.020	.298	.035							.0125
.025	.190	-.217	.211	-.097	.206	.021	.180	.118	.239	.109	.220	.118	.241	.115	.025
.050	.072	.001	.065	-.088	.084	.030		.125							.050
.075	.032	.080	.009	-.079	.004	.026	.023	.110	.078		.071	.108	.048	.103	.075
.100	-.004	.111	-.047	-.023	-.048	.015	-.047	.120							.100
.150	-.048	.120	-.112	.110	-.038	.116	-.106	.195	-.090	.107	-.067	.124	-.048	.107	.150
.200	-.073	.167	-.169	.239	-.180	.088	-.173	.137							.200
.250	-.060	.171	-.204	.246	-.193	.175	-.186	.159	-.155	.190	-.171	.139	-.145	.165	.250
.300	-.140	.223	-.205	.250	-.239	.246	-.231	.181							.300
.350									-.244	.243	-.240	.180	-.237	.193	.350
.400	-.155	.225	-.237	.223	-.289	.270	-.288	.208	-.295	.208	-.280	.202	-.276	.190	.400
.450															.450
.500	-.184	.213	-.277	.219	-.322	.224	-.335	.211	-.334	.179	-.317	.181	-.305	.157	.500
.550															.550
.600	-.209	.192	-.269	.187	-.312	.195	-.347	.206	-.349	.187	-.337	.171	-.332	.128	.600
.650															.650
.700	-.234	.159	-.271	.167	-.343	.172	-.355	.200	-.354	.147	-.341	.160	-.336	.092	.700
.750															.750
.800	-.213	.110	-.266	.148	-.342	.151	-.351	.156	-.357	.139	-.341	.152	-.307	.053	.800
.850															.850
.900	-.211	.131	-.241	.133	-.295	.134	-.342								.900
.950															.950
α = .10															
.0125	.255	-.174	.255	-.022	.254	.101	.244	.084							.0125
.025	.140	-.159	.167	-.035	.155	.103	.122	.217	.179	.200	.143	.201	.172	.242	.025
.050	.029	.072	.022	-.004	.053	.113		.203							.050
.075	-.007	.111	-.033	.033	-.038	.114	-.015	.195	.022		-.008	.231	-.023	.276	.075
.100	-.031	.145	-.087	.110	-.086	.116	-.106	.258							.100
.150	-.076	.156	-.145	.231	-.149	.178		.258	-.137	.284	-.096	.273	-.117	.285	.150
.200	-.102	.213	-.205	.281	-.212	.250	-.215	.256							.200
.250	-.090	.225	-.232	.284	-.232	.302	-.227	.265	-.208	.326	-.210	.267	-.175	.295	.250
.300	-.165	.276	-.252	.287	-.271	.309	-.271	.280							.300
.350									-.278	.303	-.282	.294	-.266	.257	.350
.400	-.178	.271	-.232	.265	-.315	.288	-.321	.286	-.321	.247	-.317	.284	-.306	.248	.400
.450															.450
.500	-.204	.254	-.306	.264	-.353	.256	-.368	.259	-.359	.212	-.351	.234	-.338	.196	.500
.550															.550
.600	-.230	.233	-.293	.227	-.358	.225	-.381	.241	-.373	.221	-.364	.209	-.344	.166	.600
.650															.650
.700	-.253	.196	-.290	.211	-.353	.206	-.391	.225	-.378	.185	-.369	.196	-.345	.130	.700
.750															.750
.800	-.232	.150	-.284	.192	-.363	.178	-.387	.189	-.380	.174	-.370	.188	-.327	.088	.800
.850															.850
.900	-.231	.173	-.260	.176	-.289	.167	-.368								.900
.950															.950

TABLE II.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING  
(a)  $M = 1.61$  - Continued

$x/c$ , nominal	$C_p$ at $y/\frac{b}{2}$ of:														$x/c$ , nominal	
	.05		.20		.35		.50		.70		.825		.95			
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower		
$\alpha = 12$																
.0125	.213	-.085	.212	.061	.206	.161	.186	.190							.0125	
.025	.094	-.101	.120	.049	.107	.176	.064	.304	.097	.343	.046	.347	.056	.374	.025	
.050	-.006	.111	-.012	.108	.010	.239	.313								.050	
.075	-.039	.148	-.091	.167	-.076	.267	-.050		-.047		-.087	.358	-.123	.354	.075	
.100	-.065	.190	-.127	.247	-.149	.294	-.129	.330							.100	
.150	-.107	.203	-.189	.306	-.194	.325	.360	.360	-.175	.370	-.152	.333	-.204	.346	.150	
.200	-.131	.264	-.238	.325	-.250	.344	.349	.349							.200	
.250	-.114	.284	-.266	.327	-.267	.351	-.266	.356	-.245	.387	-.243	.297	-.260	.348	.250	
.300	-.184	.325	-.285	.332	-.307	.343	-.306	.354							.300	
.350									-.314	.365	-.317	.345	-.311	.303	.350	
.400	-.203	.320	-.274	.312	-.349	.326	-.351	.334	-.357	.294	-.349	.326	-.342	.299	.400	
.450															.450	
.500	-.224	.304	-.327	.308	-.381	.292	-.389	.301	-.384	.267	-.375	.272	-.366	.250	.500	
.550															.550	
.600	-.250	.278	-.318	.270	-.397	.268	-.408	.286	-.395	.262	-.392	.254	-.374	.206	.600	
.650															.650	
.700	-.269	.240	-.311	.253	-.392	.247	-.414	.256	-.402	.236	-.397	.239	-.376	.170	.700	
.750															.750	
.800	-.247	.197	-.304	.231	-.379	.223	-.412	.231	-.402	.219	-.397	.227	-.369	.122	.800	
.850															.850	
.900	-.247	.223	-.278	.223	-.261	.215	-.286								.900	
.950															.950	
$\alpha = 14$																
.0125	.169	-.012	.155	.141	.144	.287	.085	.393							.0125	
.025	.049	-.005	.065	.148	.042	.311	-.027	.390	-.027	.441	-.069	.430	-.083	.435	.025	
.050	-.039	.168	-.036	.280	-.029	.396	.396								.050	
.075	-.074	.201	-.129	.326	-.115	.369	-.080	.392	-.139		-.193	.411	-.210	.408	.075	
.100	-.096	.242	-.185	.342	-.184	.392	-.163	.415							.100	
.150	-.133	.267	-.232	.363	-.235	.397	.415	.430	-.223	.428	-.236	.403	-.278	.404	.150	
.200	-.155	.315	-.275	.376	-.289	.405	.417	.429							.200	
.250	-.146	.342	-.303	.373	-.306	.401	-.306	.418	-.282	.440	-.295	.360	-.423	.406	.250	
.300	-.198	.369	-.316	.379	-.338	.395	-.338	.403	-.352	.400	-.352	.398	-.367	.365	.300	
.350															.350	
.400	-.221	.367	-.316	.361	-.377	.372	-.378	.385	-.388	.345	-.380	.385	-.387	.364	.400	
.450															.450	
.500	-.245	.353	-.342	.356	-.405	.334	-.411	.348	-.411	.305	-.405	.330	-.407	.303	.500	
.550															.550	
.600	-.269	.333	-.344	.316	-.419	.314	-.423	.324	-.422	.308	-.418	.313	-.408	.267	.600	
.650															.650	
.700	-.282	.290	-.334	.298	-.427	.300	-.432	.301	-.428	.282	-.424	.298	-.411	.222	.700	
.750															.750	
.800	-.260	.249	-.324	.287	-.415	.273	-.429	.282	-.409	.273	-.424	.285	-.406	.177	.800	
.850															.850	
.900	-.262	.265	-.293	.269	-.297	.264	-.306								.900	
.950															.950	
$\alpha = 16$																
.0125	.114	.068	.094	.248	.058	.411	-.023	.510							.0125	
.025	-.008	.134	.002	.288	-.040	.439	-.137	.488	-.162	.540	-.218	.516	-.309	.540	.025	
.050	-.082	.207	-.078	.385	-.118	.447	.480								.050	
.075	-.120	.255	-.163	.404	-.159	.456	.477	.488	-.239		-.289	.478	-.332	.518	.075	
.100	-.133	.292	-.219	.409	-.211	.462	-.224	.490							.100	
.150	-.171	.329	-.278	.421	-.270	.458	.500	.500	-.300	.504	-.325	.478	-.359	.577	.150	
.200	-.187	.367	-.316	.433	-.321	.464	-.323	.486							.200	
.250	-.182	.399	-.338	.426	-.339	.456	-.339	.488	-.339	.504	-.365	.425	-.385	.626	.250	
.300	-.216	.423	-.350	.431	-.376	.448	-.374	.478							.300	
.350									-.392	.477	-.401	.478	-.412	.585	.350	
.400	-.244	.418	-.363	.414	-.407	.435	-.407	.461	-.418	.415	-.416	.465	-.423	.550	.400	
.450															.450	
.500	-.268	.404	-.359	.408	-.435	.407	-.434	.415	-.437	.384	-.435	.430	-.423	.453	.500	
.550															.550	
.600	-.290	.383	-.368	.383	-.446	.389	-.446	.397	-.444	.390	-.441	.451	-.426	.394	.600	
.650															.650	
.700	-.302	.347	-.355	.371	-.454	.369	-.454	.386	-.395	.357	-.425	.526	-.418	.322	.700	
.750															.750	
.800	-.276	.321	-.336	.363	-.437	.357	-.369	.358	-.351	.354	-.357	.465	-.389	.243	.800	
.850															.850	
.900	-.280	.340	-.312	.346	-.344	.338	-.339								.900	
.950															.950	
$\alpha = 18$																
.0125	.066	.152	.031	.387	-.028	.541	-.151	.641							.0125	
.025	-.049	.207	-.056	.410	-.116	.553	-.247	.601	-.328	.604	-.387	.623	-.451	.637	.025	
.050	-.115	.278	-.144	.433	-.190	.555	.585								.050	
.075	-.156	.316	-.211	.476	-.228	.561	-.275	.585	-.346		-.398	.601	-.448	.606	.075	
.100	-.167	.354	-.253	.485	-.267	.572	-.299	.573							.100	
.150	-.196	.396	-.313	.482	-.308	.554	.577	.577	-.379	.565	-.400	.611	-.451	.613	.150	
.200	-.208	.426	-.349	.510	-.352	.564	-.374	.555							.200	
.250	-.204	.460	-.373	.510	-.366	.558	-.388	.555	-.395	.583	-.428	.596	-.446	.601	.250	
.300	-.231	.484	-.386	.534	-.398	.543	-.410	.546							.300	
.350									-.437	.564	-.443	.644	-.451	.564	.350	
.400	-.261	.496	-.402	.521	-.427	.515	-.439	.525	-.452	.540	-.434	.621	-.451	.537	.400	
.450															.450	
.500	-.284	.510	-.384	.506	-.454	.482	-.460	.482	-.457	.539	-.428	.556	-.451	.459	.500	
.550															.550	
.600	-.303	.485	-.387	.469	-.467	.451	-.469	.472	-.399	.550	-.418	.515	-.445	.400	.600	
.650															.650	
.700	-.315	.439	-.377	.447	-.475	.441	-.475	.463	-.381	.522	-.411	.480	-.439	.331	.700	
.750															.750	
.800	-.287	.406	-.362	.439	-.429	.431	-.372	.453	-.381	.482	-.405	.423	-.429	.248	.800	
.850															.850	
.900	-.291	.411	-.335	.416	-.375	.407	-.374								.900	
.950															.950	



TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(b)  $M = 2.01$

x/c, nominal	Cp at $y/\frac{1}{2}$ of:														x/c, nominal
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
a = -20															
.0125	.828	-.303	.691	-.310	.649	-.312	.597	-.308							.0125
.025	.829	-.304	.741	-.311	.719	-.317	.664	-.311							.025
.050	.759	-.314	.718	-.307	.689	-.297	.655	-.308							.050
.075	.716	-.307	.685	-.310	.664	-.292			.645		.578	-.283	.606	-.252	.075
.100	.683	-.298	.631	-.307	.618	-.312	.619	-.284							.100
.150	.595	-.294	.575	-.302	.595	-.312	.584	-.312	.575	-.302	.539	-.281	.539	-.246	.150
.200	.505	-.270	.521	-.300	.533	-.304	.528	-.309							.200
.250	.464	-.239	.462	-.297	.481	-.306	.461	-.305	.513	-.299	.491	-.275	.499	-.241	.250
.300	.442	-.254		-.292	.433	-.298	.443	-.294							.300
.350									.414	-.294	.410	-.253	.439	-.231	.350
.400	.403	-.187	.345	-.302	.342	-.300	.365	-.299							.400
.450					.287	-.300	.288	-.291	.345	-.287		-.266	.396	-.221	.450
.500	.319	-.188	.295	-.293					.305	-.279	.323	-.266	.358	-.214	.500
.550					.253	-.297	.261	-.273							.550
.600	.288	-.185	.251	-.289					.286	-.285	.314	-.264	.342	-.212	.600
.650															.650
.700	.248	-.190	.217	-.293	.218	-.300	.239	-.266							.700
.750						-.296	.235	-.291	.279		.308	-.273	.321	-.209	.750
.800	.197	-.179	.203	-.290					.286	-.280	.317	-.270	.343	-.210	.800
.850					.221	-.292	.252	-.280							.850
.900	.220	-.190	.159	-.282											.900
.950															.950
a = -18															
.0125	.787	-.305	.680	-.318	.637	-.318	.597	-.317							.0125
.025	.769	-.310	.710	-.323	.690	-.324	.644	-.317							.025
.050	.705	-.321	.680	-.317	.649	-.305	.623	-.317	.610	-.309	.555	-.285	.537	-.252	.050
.075	.663	-.317	.638	-.320	.621	-.317			.620		.563	-.281	.584	-.251	.075
.100	.619	-.318	.580	-.317	.565	-.319	.580	-.288							.100
.150	.505	-.313	.521	-.318	.545	-.323	.530	-.317	.598	-.308	.504	-.280	.506	-.249	.150
.200	.448	-.275	.468	-.320	.477	-.317	.480	-.320							.200
.250	.406	-.243	.399	-.317	.430	-.320	.441	-.321	.475	-.309	.450	-.276	.459	-.243	.250
.300	.383	-.229		-.309	.383	-.315	.394	-.305							.300

TABLE II.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(b)  $M = 2.01$  - Continued

$x/c$ , nominal	$C_p$ at $y/b$ of :														$x/c$ , nominal	
	.05		.20		.35		.50		.70		.825		.95			
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower		
$\alpha = -12$																
.0125	.674	-.308	.618	-.316	.602	-.308	.570	-.312							.0125	
.025	.629	-.308	.599	-.313	.597	-.311	.565	-.312	.576	-.299	.530	-.278	.518	-.250	.025	
.050	.542	-.315	.535	-.311	.535	-.306	.519	-.312							.050	
.075	.486	-.315	.492	-.313	.492	-.310			.520		.495	-.270	.500	-.248	.075	
.100	.434	-.315	.431	-.314	.425	-.310	.440	-.281							.100	
.150	.346	-.293	.363	-.319	.387	-.313	.391	-.311	.412	-.301	.397	-.265	.416	-.245	.150	
.200	.302	-.158	.313	-.311	.324	-.318	.330	-.311							.200	
.250	.260	-.108	.242	-.305	.276	-.319	.290	-.312	.338	-.298	.326	-.260	.326	-.241	.250	
.300	.230	-.117		-.287	.238	-.315	.245	-.303							.300	
.350									.238	-.297	.244	-.257	.251	-.234	.350	
.400	.189	-.119	.155	-.267	.155	-.302	.170	-.305							.400	
.450									.162	-.299		-.256	.210	-.224	.450	
.500	.138	-.123	.102	-.247	.096	-.289	.094	-.297							.500	
.550									.110	-.294	.130	-.251	.162	-.214	.550	
.600	.117	-.118	.072	-.235	.071	-.279	.070	-.290	.090	-.283	.115	-.253	.142	-.206	.600	
.650									.083		.108	-.258	.108	-.201	.650	
.700	.072	-.131	.060	-.191	.046	-.270	.047	-.284	.087	-.264	.112	-.266	.120	-.205	.700	
.750															.750	
.800	.039	-.127	.050	-.125		-.263	.050	-.275							.800	
.850															.850	
.900	.061	-.135	.018	-.120	.052	-.258	.061	-.249							.900	
.950															.950	
$\alpha = -10$																
.0125	.639	-.310	.586	-.312	.576	-.304	.554	-.308							.0125	
.025	.577	-.310	.557	-.312	.561	-.304	.524	-.306	.557	-.298	.512	-.284	.509	-.254	.025	
.050	.488	-.318	.485	-.309	.493	-.307	.481	-.307							.050	
.075	.419	-.318	.435	-.312	.447	-.307			.480		.462	-.278	.468	-.252	.075	
.100	.375	-.312	.382	-.317	.371	-.307	.394	-.282							.100	
.150	.293	-.261	.314	-.317	.333	-.310	.345	-.309	.372	-.294	.352	-.271	.379	-.249	.150	
.200	.253	-.109	.259	-.305	.266	-.318	.279	-.312							.200	
.250	.211	-.099	.191	-.292	.228	-.318	.237	-.316	.288	-.295	.282	-.266	.285	-.243	.250	
.300	.188	-.099		-.270	.181	-.309	.192	-.302							.300	
.350									.190	-.294	.201	-.264	.203	-.237	.350	
.400	.145	-.104	.109	-.241	.107	-.288	.120	-.295							.400	
.450									.118	-.298		-.262	.164	-.230	.450	
.500	.095	-.109	.063	-.214	.042	-.267	.050	-.286	.066	-.292	.084	-.262	.119	-.224	.500	
.550									.066	-.292	.084	-.262	.119	-.224	.550	
.600	.075	-.104	.033	-.189	.024	-.255	.026	-.277	.048	-.282	.070	-.265	.090	-.218	.600	
.650									.048	-.282	.070	-.265	.090	-.218	.650	
.700	.033	-.119	.018	-.130	.006	-.242	.004	-.269	.038		.059	-.268	.061	-.217	.700	
.750									.038		.059	-.268	.061	-.217	.750	
.800	.006	-.116	.011	-.100		-.234	.002	-.253			.059	-.268	.061	-.217	.800	
.850											.059	-.268	.061	-.217	.850	
.900	.030	-.123	-.016	-.114	.009	-.229	.013	-.235	.038	-.251	.060	-.274	.068	-.221	.900	
.950															.950	
$\alpha = -08$																
.0125	.597	-.310	.560	-.310	.552	-.293	.534	-.303							.0125	
.025	.531	-.311	.527	-.310	.533	-.296	.492	-.303	.519	-.290	.499	-.289	.483	-.265	.025	
.050	.423	-.315	.442	-.308	.454	-.300	.453	-.303							.050	
.075	.364	-.309	.382	-.310	.402	-.295			.435		.425	-.286	.429	-.265	.075	
.100	.328	-.294	.336	-.314	.330	-.297	.347	-.278							.100	
.150	.258	-.216	.258	-.310	.282	-.303	.294	-.303	.322	-.287	.321	-.283	.341	-.264	.150	
.200	.198	-.077	.201	-.289	.220	-.307	.233	-.308							.200	
.250	.168	-.091	.144	-.268	.178	-.299	.202	-.311	.240	-.287	.235	-.279	.242	-.262	.250	
.300	.144	-.080		-.237	.129	-.286	.144	-.293							.300	
.350									.146	-.287	.153	-.272	.159	-.256	.350	
.400	.105	-.082	.064	-.204	.064	-.256	.078	-.274	.078	-.285		-.272	.120	-.251	.400	
.450									.078	-.285		-.272	.120	-.251	.450	
.500	.059	-.088	.018	-.176	.003	-.233	.011	-.261	.024	-.277	.045	-.267	.080	-.240	.500	
.550									.024	-.277	.045	-.267	.080	-.240	.550	
.600	.049	-.088	-.005	-.133	-.016	-.217	-.012	-.249	.006	-.267	.026	-.259	.054	-.230	.600	
.650									.006	-.267	.026	-.259	.054	-.230	.650	
.700	.003	-.099	-.020	-.088	-.035	-.205	-.035	-.240	.001		.020	-.251	.025	-.222	.700	
.750									.001		.020	-.251	.025	-.222	.750	
.800	-.024	-.104	-.025	-.086	-.199	-.041	-.220				.016	-.243	.026	-.220	.800	
.850											.016	-.243	.026	-.220	.850	
.900	-.008	-.104	-.046	-.101	-.025	-.194	-.031	-.213	-.006	-.224					.900	
.950															.950	
$\alpha = -06$																
.0125	.556	-.302	.534	-.298	.531	-.289	.504	-.296							.0125	
.025	.470	-.304	.490	-.295	.501	-.290	.448	-.294	.505	-.270	.488	-.278	.481	-.269	.025	
.050	.372	-.308	.396	-.296	.420	-.298	.415	-.298							.050	
.075	.319	-.300	.338	-.299	.360	-.291			.411		.402	-.276	.400	-.270	.075	
.100	.278	-.277	.289	-.309	.299	-.309	.294	-.309							.100	
.150	.198	-.139	.213	-.290	.226	-.303	.234	-.299	.289	-.269	.288	-.275	.312	-.268	.150	
.200	.153	-.056	.155	-.258	.173	-.298	.173	-.305							.200	
.250	.127	-.075	.099	-.226	.142	-.283	.146	-.305	.202	-.270	.196	-.273	.211	-.269	.250	
.300	.104	-.059		-.198	.088	-.268	.087	-.269							.300	
.350									.109	-.266	.118	-.269	.127	-.264	.350	
.400	.063	-.058	.026	-.155	.025	-.228	.026	-.238	.043	-.256		-.267	.086	-.262	.400	
.450									.043	-.256		-.267	.086	-.262	.450	
.500	.025	-.068	-.016	-.116	-.025	-.198	-.036	-.219	-.007	-.247	.014	-.255	.048	-.253	.500	
.550									-.007	-.247	.014	-.255	.048	-.253	.550	
.600	.005	-.064	-.040	-.083	-.050	-.183	-.059	-.208	-.024	-.230	-.005	-.242	.024	-.240	.600	
.650									-.024	-.230	-.005	-.242	.024	-.240	.650	
.700	-.030	-.076	-.051	-.066	-.074	-.171	-.080	-.194	-.028		-.012	-.222	-.010	-.224	.700	
.750									-.028		-.012	-.222	-.010	-.224	.750	
.800	-.051	-.090	-.051	-.069	-.162	-.083	-.177		-.037	-.197	-.012	-.205	-.010	-.210	.800	
.850											-.012	-.205	-.010	-.210	.850	
.900	-.033	-.082	-.070	-.083	-.067	-.154	-.074	-.172							.900	
.950															.950	

TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(b)  $M = 2.01$  - Continued[illegible]

TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(b)  $M = 2.01$  - Continued

x/c, nominal	Cp at $y/\frac{1}{2}$ of :														x/c, nominal
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
$\alpha = 04$															
.0125	.370	-.194	.390	-.217	.401	-.255	.407	-.244							.0125
.025	.258	-.187	.319	-.217	.329	-.250	.304	-.244							.025
.050	.157	-.183	.190	-.217	.217	-.249	.250	-.244							.050
.075	.107	-.103	.129	-.206	.148	-.244			.226		.234	-.263	.239	-.249	.075
.100	.074	-.006	.071	-.141	.096	-.215	.115	-.236							.100
.150	.014	.038	.011	-.013	.033	-.126	.039	-.163	.094	-.179	.092	-.202	.143	-.246	.150
.200	-.022	.068	-.038	.065	-.029	.019	-.015	-.100							.200
.250	-.023	.063	-.080	.082	-.053	.087	-.042	-.028	.009	-.096	.015	-.137	.023	-.190	.250
.300	-.045	.087		.086	-.100	.099	-.079	.072							.300
.350									-.068	-.013	-.063	-.084	-.037	-.111	.350
.400	-.081	.096	-.137	.091	-.145	.104	-.133	.108	-.119	.054		-.030	-.082	-.079	.400
.450															.450
.500	-.106	.079	-.160	.095	-.183	.086	-.176	.103	-.162	.058	-.138	.005	-.116	-.070	.500
.550															.550
.600	-.132	.091	-.175	.077	-.204	.074	-.193	.090	-.174	.065	-.156	.029	-.133	-.061	.600
.650															.650
.700	-.145	.061	-.175	.064	-.214	.061	-.205	.079							.700
.750									-.184		-.167	.039	-.157	-.032	.750
.800	-.137	.034	-.173	.053		.049	-.209	.061	-.184	.050	-.170	.044	-.163	-.010	.800
.850															.850
.900	-.133	.040	-.176	.036	-.202	.036	-.209	.039							.900
.950															.950
$\alpha = 06$															
.0125	.333	-.163	.359	-.184	.366	-.228	.372	-.232							.0125
.025	.219	-.156	.279	-.184	.283	-.223	.259	-.230	.351	-.223	.339	-.241	.354	-.240	.025
.050	.123	-.141	.159	-.183	.181	-.237	.212	-.235							.050
.075	.071	-.036	.087	-.179	.100	-.215			.196		.209	-.245	.200	-.235	.075
.100	.035	.033	.035	-.090	.052	-.170	.072	-.210							.100
.150	-.014	.068	-.026	.057	-.008	.003	.004	-.113	.065	-.136	.072	-.153	.111	-.230	.150
.200	-.040	.098	-.067	.115	-.064	.109	-.056	.013							.200
.250	-.042	.087	-.108	.118	-.089	.135	-.076	.121	-.011	.054	-.005	-.090	-.003	-.133	.250
.300	-.065	.121		.135	-.136	.141	-.114	.162							.300
.350									-.088	.144		-.080	.036	-.077	.350
.															



TABLE II.- Continued

PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING

(b) M = 2.01 - Continued

[illegible]

TABLE II.- Concluded  
 PRESSURE COEFFICIENTS FOR CAMBERED AND TWISTED WING  
 (b)  $M = 2.01$  - Concluded

x/c, nominal	Cp at y/b of :														x/c, nominal
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
α = 20															
.0125	.081	.177	.095	.347	.093	.457	.027	.575							.0125
.025	-.040	.209	.011	.385	.011	.488	-.079	.552	-.068	.595	-.113	.596	-.173		.025
.050	-.114	.279	-.080	.414	-.080	.472	-.100	.541							.050
.075	-.141	.327	-.124	.456	-.128	.515			-.140		-.168	.555	-.197		.075
.100	-.156	.347	-.172	.468	-.161	.526	-.159	.536							.100
.150	-.172	.380	-.219	.468	-.200	.522	-.198	.552	-.194	.567	-.198	.541	-.221		.150
.200	-.182	.420	-.248	.491	-.236	.530	-.238	.530							.200
.250	-.190	.420	-.261	.499	-.254	.522	-.250	.529	-.232	.563	-.236	.493	-.241		.250
.300	-.185	.489		.498	-.272	.509	-.273	.529							.300
.350									-.259	.540	-.262	.525	-.265		.350
.400	-.221	.495	-.298	.477	-.296	.491	-.293	.511							.400
.450									-.276	.477		.525	-.272		.450
.500	-.231	.470	-.311	.465	-.312	.446	-.305	.471							.500
.550									-.287	.449	-.281	.467	-.273		.550
.600	-.233	.421	-.298	.429	-.317	.417	-.315	.447							.600
.650									-.284	.437	-.280	.444	-.269		.650
.700	-.250	.372	-.296	.404	-.317	.395	-.319	.421							.700
.750									-.275		-.274	.426	-.267		.750
.800	-.234	.359	-.298	.385		.368	-.285	.390							.800
.850									-.260	.388	-.264	.408	-.264		.850
.900	-.229	.380	-.295	.364	-.271	.358	-.266	.366							.900
.950															.950

TABLE III  
PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(a)  $M = 1.61$

[illegible]



TABLE III.- Continued

PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(a) M = 1.61 - Continued

[illegible]

TABLE III.- Continued  
 PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING  
 (a)  $M = 1.61$  - Continued

$x/c$	$C_p$ at $y/\frac{b}{2}$ of :														$x/c$	
	.05		.20		.35		.50		.70		.825		.95			
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower		
$\alpha = 0^\circ$																
.0125	.147	.020	.139	-.057	.178	-.047	.144	.052			.031	.288		.306	.0125	
.025	.136	.020	.112	-.004	.129	-.003	.107	.066							.025	
.050	.092	.052	.080	.034	.088	.042	.087	.094							.050	
.075	.069	.049	.039	.039	.021	.053		.197	.042	.270	-.000	.288	-.049	.310	.075	
.100	.041	.058		.061	-.008	.113	-.011	.213							.100	
.150	-.014	.066	-.049	.116	-.065	.201	-.068	.259	-.093	.295	-.065	.295	-.114	.296	.150	
.200	-.039	.113	-.120	.179	-.126	.249	-.139	.274							.200	
.250	-.053	.154	-.149	.215	-.192	.266	-.182	.285	-.142	.337	-.223	.275	-.202	.301	.250	
.300	-.133		-.182	.238	-.231	.272	-.245	.315							.300	
.350									-.291	.295	-.298	.249	-.299	.286	.350	
.400	-.173	.192	-.257	.225	-.294	.251	-.306	.269							.400	
.450									-.336	.214	-.343	.227	-.360	.233	.450	
.500	-.172	.179	-.281	.187	-.315	.197	-.349	.197	-.349	.127	-.355	.147	-.371	.139	.500	
.550															.550	
.600	-.175	.122	-.213	.092	-.304	.088	-.332	.093	-.318	.024	-.333	.015	-.342	-.008	.600	
.650															.650	
.700	-.145	.032		-.023	-.201	.021	-.277	-.049	-.268	-.097	-.262	-.100	-.276	-.157	.700	
.750															.750	
.800	-.084	-.077	-.053	-.120	-.084	-.140	-.132	-.140	-.197	-.183	-.209	-.186	-.189	-.224	.800	
.850															.850	
.900	-.027	-.098	.022	-.182	-.015	-.188	-.054	-.199							.900	
.950															.950	
$\alpha = 0^\circ$																
.0125	.078	.080	.037	.075	.072	.132	.000	.244							.0125	
.025	.078	.075	.049	.100	.042	.133	-.006	.229	-.128		-.201	.394		.387	.025	
.050	.056	.101	.037	.110	.027	.160	.015	.265							.050	
.075	.032	.092	-.014	.104	-.020	.181		.271	-.065	.352	-.155	.352	-.251	.363	.075	
.100	.012	.101		.120	-.074	.203	-.065	.288							.100	
.150	-.044	.096	-.084	.162	-.107	.269	-.129	.312	-.128	.347	-.151	.345	-.254	.340	.150	
.200	-.068	.157	-.151	.242	-.167	.303	-.182	.324							.200	
.250	-.071	.209	-.192	.266	-.226	.320	-.219	.326	-.191	.376	-.248	.311	-.297	.339	.250	
.300	-.150		-.220	.284	-.265	.317	-.274	.353							.300	
.350									-.330	.328	-.326	.276	-.343	.313	.350	
.400	-.199	.243	-.274	.270	-.321	.288	-.333	.301	-.368	.242	-.374	.259	-.375	.260	.400	
.450															.450	
.500	-.198	.225	-.321	.228	-.353	.234	-.373	.239	-.372	.149	-.387	.175	-.382	.162	.500	
.550															.550	
.600	-.203	.163	-.242	.126	-.319	.123	-.359	.121	-.356	.052	-.376	.036	-.349	.021	.600	
.650															.650	
.700	-.169	.069		.013	-.252	.007	-.317	-.016	-.311	-.075	-.332	-.074	-.306	-.129	.700	
.750															.750	
.800	-.099	-.046	-.082	-.096	-.119	-.110	-.193	-.115	-.252	-.163	-.268	-.160	-.242	-.197	.800	
.850															.850	
.900	-.042	-.072	-.007	-.154	-.061	-.160	-.103	-.180							.900	
.950															.950	
$\alpha = 0^\circ$																
.0125	.005	.149	-.099	.183	-.073	.281	-.186	.378							.0125	
.025	.031	.130	-.048	.186	-.066	.258	-.172	.344	-.287		-.314	.459		.434	.025	
.050	.022	.145	-.016	.182	-.050	.275	-.123	.359							.050	
.075	.005	.137	-.060	.168	-.063	.273		.352	-.233	.413	-.282	.406	-.337	.406	.075	
.100	-.017	.145		.185	-.115	.285	-.095	.358							.100	
.150	-.068	.135	-.124	.229	-.154	.329	-.177	.363	-.239	.400	-.279	.389	-.330	.382	.150	
.200	-.094	.199	-.183	.298	-.200	.352	-.233	.374							.200	
.250	-.094	.261	-.219	.319	-.259	.363	-.262	.375	-.262	.415	-.329	.341	-.364	.375	.250	
.300	-.164		-.253	.333	-.292	.358	-.316	.395							.300	
.350									-.338	.363	-.375	.316	-.400	.340	.350	
.400	-.215	.291	-.286	.316	-.345	.330	-.361	.338	-.379	.271	-.402	.288	-.423	.289	.400	
.450															.450	
.500	-.216	.269	-.341	.269	-.380	.273	-.396	.268	-.393	.187	-.412	.201	-.414	.189	.500	
.550															.550	
.600	-.221	.204	-.272	.166	-.355	.157	-.387	.157	-.376	.081	-.398	.072	-.398	.052	.600	
.650															.650	
.700	-.183	.104		.053	-.281	.039	-.355	.014	-.333	-.040	-.355	-.040	-.370	-.100	.700	
.750															.750	
.800	-.117	-.016	-.112	-.066	-.155	-.078	-.235	-.077	-.285	-.130	-.296	-.128	-.321	-.167	.800	
.850															.850	
.900	-.064	-.044	-.034	-.116	-.100	-.134	-.142	-.139							.900	
.950															.950	
$\alpha = 10^\circ$																
.0125	-.083	.219	-.205	.274	-.215	.399	-.304	.474			-.400	.507		.461	.0125	
.025	-.031	.192	-.184	.265	-.192	.365	-.306	.421	-.376						.025	
.050	-.023	.185	-.140	.245	-.190	.363	-.270	.423							.050	
.075	-.031	.175	-.126	.235	-.188	.345		.417	-.329	.458	-.371	.440	-.408	.438	.075	
.100	-.051	.187		.254	-.195	.351	-.254	.419							.100	
.150	-.101	.173	-.167	.307	-.266	.384	-.262	.413	-.328	.436	-.360	.418	-.399	.409	.150	
.200	-.125	.244	-.222	.349	-.241	.401	-.292	.422							.200	
.250	-.125	.314	-.253	.369	-.294	.409	-.315	.417	-.345	.446	-.398	.367	-.421	.396	.250	
.300	-.192		-.293	.379	-.321	.401	-.355	.432							.300	
.350									-.413	.392	-.426	.337	-.445	.352	.350	
.400	-.241	.332	-.305	.361	-.374	.374	-.393	.375	-.438	.303	-.445	.309	-.452	.307	.400	
.450															.450	
.500	-.247	.312	-.358	.315	-.400	.307	-.421	.301	-.443	.208	-.450	.226	-.445	.209	.500	
.550															.550	
.600	-.247	.246	-.301	.203	-.394	.187	-.412	.185	-.428	.114	-.440	.090	-.441	.076	.600	
.650															.650	
.700	-.207	.139		.090	-.306	.078	-.382	.051	-.385	-.017	-.411	-.019	-.431	-.076	.700	
.750															.750	
.800	-.140	.015	-.150		-.207	-.051	-.299	-.058	-.345	-.102	-.369	-.109	-.398	-.145	.800	
.850															.850	
.900	-.087	-.014	-.069	-.084	-.153	-.102	-.176	-.117							.900	
.950															.950	

TABLE III.- Continued

PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(a)  $M = 1.61$  - Continued

[illegible]





TABLE III.- Continued

PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(b)  $M = 2.01$

x/c	Cp at $y/b_2$ of:														x/c
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
a = -20															
.0125	.762	-.294	.727	-.300	.683	-.302	.635	-.298							.0125
.025	.733	-.284	.690	-.300	.678	-.302	.645	-.297							.025
.050	.668	-.292	.660	-.303	.631	-.302	.621	-.296							.050
.075	.638	-.296	.623	-.296	.612	-.299	-.299		.567	-.290	.562	-.293	.539	-.295	.075
.100	.602	-.287	-.296		.568	-.274	.576	-.268							.100
.150	.500	-.284	.520	-.296	.503	-.299	.480	-.293	.491	-.285	.459	-.289		-.290	.150
.200	.453	-.233	.442	-.298	.448	-.302	.427	-.295							.200
.250	.387	-.133	.385	-.292	.358	-.299	.362	-.301	.428	-.285	.353	-.286	.345	-.282	.250
.300	.367	-.121	.327	-.284	.302	-.295	.296	-.278							.300
.350									.247	-.283	.251	-.286	.249	-.282	.350
.400	.272	-.175	.218	-.260	.205	-.286	.206	-.285							.400
.450									.177	-.280	.195	-.281	.442	-.282	.450
.500	.244	-.167	.198	-.260	.181	-.280	.159	-.275	.171	-.279	.378	-.283	.530	-.284	.500
.550									.171	-.279	.378	-.283	.530	-.284	.550
.600	.256	-.181	.243	-.266	.238	-.280	.241	-.275	.457	-.270	.519	-.278	.565	-.275	.600
.650															.650
.700	.295	-.207		-.280	.364	-.288	.409	-.278	.554	-.243	.576	-.261	.565	-.267	.700
.750									.554	-.243	.576	-.261	.565	-.267	.750
.800	.326	-.225	.413	-.287	.467	-.296	.523	-.293	.593	-.243	.577	-.245	.544	-.262	.800
.850															.850
.900	.422	-.246	.494	-.290	.528	-.302	.557	-.269							.900
.950															.950
a = -18															
.0125	.705	-.303	.693	-.305	.657	-.307	.610	-.297							.0125
.025	.679	-.288	.645	-.307	.644	-.304	.618	-.298							.025
.050	.608	-.293	.608	-.310	.592	-.304	.589	-.299	.576		.560	-.298		-.289	.050
.075	.579	-.296	.572	-.302	.564	-.304	-.297		.533	-.293	.531	-.298	.512	-.297	.075
.100	.542	-.284	-.304		.521	-.286	.528	-.267							.100
.150	.444	-.283	.464	-.302	.451	-.305	.443	-.295	.448	-.292	.418	-.294		-.294	.150
.200	.392	-.191	.388	-.304	.396	-.307	.384	-.298							.200
.250	.335	-.180	.338	-.294	.304	-.307	.315	-.300	.389	-.290	.309	-.290	.307	-.290	.250
.300	.318	-.085	.270	-.282	.257	-.302	.254	-.288							.300
.350									.203	-.291	.212	-.291	.211	-.287	.350
.400	.225	-.137	.169	-.259	.156	-.287									

TABLE III.- Continued  
PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(b) M = 2.01 - Continued

x/c	Cp at $y/\frac{b}{2}$ of:														x/c	
	.05		.20		.35		.50		.70		.825		.95			
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower		
$\alpha = -12$																
.0125	.560	-.259	.586	-.299	.585	-.312	.557	-.312			.491	-.289		-.288	.0125	
.025	.528	-.236	.525	-.287	.531	-.312	.520	-.306	.509					-.288	.025	
.050	.439	-.246	.464	-.286	.474	-.307	.474	-.306							.050	
.075	.404	-.233	.427	-.283	.424	-.305		-.303	.429	-.290	.429	-.287	.423	-.299	.075	
.100	.363	-.218		-.289	.382	-.295	.396	-.274							.100	
.150	.292	-.212	.320	-.280	.306	-.300	.307	-.302	.338	-.288	.308	-.283		-.292	.150	
.200	.251	-.101	.243	-.236	.253	-.293	.243	-.308							.200	
.250	.199	-.087	.183	-.204	.165	-.267	.193	-.298							.250	
.300	.195	-.052	.137	-.187	.125	-.232	.119	-.270	.266	-.288	.193	-.283	.184	-.288	.300	
.350									.079	-.278	.090	-.283	.095	-.258	.350	
.400	.099	-.072	.042	-.165	.045	-.194	.047	-.221							.400	
.450									.021	-.230	.023	-.256	.035	-.227	.450	
.500	.080	-.098	.033	-.151	-.004	-.154	-.005	-.210	.001	-.223	.006	-.228	.022	-.227	.500	
.550															.550	
.600	.095	-.113	.063	-.152	.044	-.171	.014	-.226	.048	-.245	.061	-.251	.067	-.246	.600	
.650															.650	
.700	.114	-.153		-.181	.130	-.211	.096	-.259	.170	-.270	.173	-.275	.197	-.272	.700	
.750															.750	
.800	.164	-.181	.220	-.221	.235	-.251	.245	-.286	.303	-.271	.291	-.273	.286	-.269	.800	
.850															.850	
.900	.227	-.212	.284	-.246	.311	-.270	.325	-.259							.900	
.950															.950	
$\alpha = -10$																
.0125	.514	-.224	.555	-.289	.558	-.316	.526	-.316			.463	-.285		-.280	.0125	
.025	.479	-.208	.486	-.270	.496	-.320	.487	-.307	.481						.025	
.050	.380	-.212	.427	-.266	.432	-.308	.431	-.303							.050	
.075	.354	-.199	.386	-.266	.386	-.298		-.297	.393	-.280	.394	-.278	.394	-.289	.075	
.100	.319	-.180		-.272	.334	-.291	.355	-.266							.100	
.150	.256	-.146	.276	-.256	.267	-.280	.271	-.290	.275	-.278	.275	-.275		-.285	.150	
.200	.208	-.084	.202	-.227	.209	-.266	.194	-.287							.200	
.250	.161	-.060	.147	-.152	.126	-.245	.136	-.259	.228	-.277	.155	-.274	.142	-.264	.250	
.300	.159	-.039	.097	-.096	.084	-.216	.076	-.231							.300	
.350									.041	-.224	.052	-.250	.062	-.213	.350	
.400	.066	-.062	.012	-.066	.017	-.162	.001	-.180	-.017	-.183	-.015	-.200	-.002	-.192	.400	
.450															.450	
.500	.050	-.076	-.004	-.087	-.039	-.115	-.036	-.171	-.037	-.187	-.031	-.190	-.020	-.197	.500	
.550															.550	
.600	.058	-.098	.027	-.106	-.005	-.126	-.011	-.192	.008	-.215	.017	-.218	.018	-.223	.600	
.650															.650	
.700	.081	-.130		-.155	.079	-.174	.044	-.234	.099	-.253	.101	-.258	.114	-.257	.700	
.750															.750	
.800	.123	-.167	.176	-.206	.188	-.225	.177	-.268	.220	-.274	.221	-.278	.199	-.274	.800	
.850															.850	
.900	.193	-.193	.246	-.233	.263	-.254	.263	-.271							.900	
.950															.950	
$\alpha = -08$																
.0125	.453	-.203	.498	-.292	.511	-.311	.492	-.315			.429	-.291		-.263	.0125	
.025	.409	-.190	.433	-.266	.456	-.291	.449	-.308	.446						.025	
.050	.322	-.196	.369	-.253	.384	-.291	.392	-.296							.050	
.075	.303	-.186	.332	-.254	.350	-.281		-.287	.356	-.276	.354	-.273	.354	-.271	.075	
.100	.268	-.158		-.259	.293	-.278	.307	-.250							.100	
.150	.201	-.126	.217	-.246	.207	-.256	.222	-.267	.240	-.276	.235	-.265		-.265	.150	
.200	.152	-.076	.142	-.211	.157	-.238	.153	-.258							.200	
.250	.109	-.051	.092	-.121	.074	-.219	.108	-.231	.184	-.255	.112	-.242	.094	-.225	.250	
.300	.111	-.037	.044	-.062	.029	-.185	.034	-.195							.300	
.350									-.004	-.172	.011	-.184	.019	-.180	.350	
.400	.021	-.039	-.027	-.035	-.046	-.113	-.039	-.147	-.057	-.142	-.057	-.146	-.036	-.159	.400	
.450															.450	
.500	.003	-.055	-.056	-.047	-.075	-.058	-.075	-.136	-.075	-.153	-.073	-.147	-.058	-.165	.500	
.550															.550	
.600	.018	-.081	-.018	-.086	-.071	-.091	-.062	-.163	-.039	-.186	-.031	-.187	-.028	-.198	.600	
.650															.650	
.700	.037	-.127		-.144	.009	-.146	-.004	-.208	.035	-.237	.042	-.234	.050	-.240	.700	
.750															.750	
.800	.076	-.162	.117	-.206	.118	-.213	.095	-.227	.133	-.269	.152	-.269	.124	-.262	.800	
.850															.850	
.900	.146	-.190	.188	-.231	.198	-.242	.200	-.252							.900	
.950															.950	
$\alpha = -06$																
.0125	.409	-.151	.453	-.258	.474	-.251	.455	-.292			.410	-.275		-.226	.0125	
.025	.363	-.153	.391	-.221	.416	-.227	.402	-.282	.424						.025	
.050	.282	-.154	.334	-.225	.340	-.252	.349	-.268							.050	
.075	.264	-.148	.292	-.212	.287	-.239		-.254	.332	-.233	.327	-.238	.324	-.234	.075	
.100	.220	-.110		-.211	.253	-.242	.264	-.219							.100	
.150	.164	-.085	.182	-.202	.162	-.213	.188	-.222	.204	-.216	.204	-.223		-.217	.150	
.200	.115	-.048	.107	-.175	.110	-.189	.114	-.208							.200	
.250	.078	-.017	.095	-.115	.037	-.163	.076	-.180	.150	-.190	.086	-.185	.061	-.164	.250	
.300	.078	-.017	.012	.010	-.003	-.122	-.004	-.135							.300	
.350									-.022	-.109	-.007	-.124	-.004	-.130	.350	
.400	-.002	-.004	-.050	.013	-.076	-.022	-.071	-.078	-.077	-.077	-.071	-.084	-.062	-.107	.400	
.450															.450	
.500	-.023	-.026	-.086	-.009	-.114	.001	-.108	-.066	-.093	-.089	-.090	-.091	-.085	-.113	.500	
.550															.550	
.600	-.006	-.048	-.052	-.052	-.098	-.049	-.087	-.069	-.062	-.131	-.060	-.139	-.055	-.153	.600	
.650															.650	
.700	.010	-.096		-.123	-.036	-.119	-.038	-.116	.003	-.189	.006	-.196	.006	-.200	.700	
.750															.750	
.800	.049	-.139	.084	-.185	.072	-.183	.037	-.171	.080	-.231	.116	-.235	.071	-.230	.800	
.850															.850	
.900	.112	-.156	.152	-.184	.150	-.217	.133	-.207							.900	
.950															.950	

TABLE III.- Continued

PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(b) M = 2.01 - Continued

x/c	Cp at $y/\frac{b}{2}$ of:														x/c
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
a = .04															
.0125	.353	-.116	.407	-.227	.438	-.376	.416	-.270							.0125
.025	.316	-.132	.341	-.221	.376	-.249	.366	-.261	.386		.372	-.259		-.209	.025
.050	.243	-.122	.287	-.201	.304	-.218	.310	-.242							.050
.075	.225	-.115	.247	-.186	.247	-.210		-.226	.291	-.212	.288	-.213	.288	-.214	.075
.100	.188	-.078		-.182	.214	-.215	.220	-.201							.100
.150	.129	-.052	.140	-.159	.134	-.176	.152	-.195	.156	-.185	.158	-.194		-.191	.150
.200	.083	-.021	.073	-.141	.080	-.156	.080	-.178							.200
.250	.053	.014	.015	-.018	.006	-.125	.042	-.147	.104	-.158	.049	-.151	.028	-.138	.250
.300	.053	.001	-.018	.041	-.036	-.080	-.027	-.100							.300
.350									-.056	-.064	-.040	-.086	-.041	-.092	.350
.400	-.028	.016	-.075	.049	-.098	.053	-.098	-.027							.400
.450									-.112	-.028	-.104	-.031	-.095	-.062	.450
.500	-.043	.001	-.106	.030	-.134	.033	-.137	.024	-.125	-.049	-.119	-.047	-.116	-.068	.500
.550															.550
.600	-.033	-.025	-.075	-.026	-.110	-.025	-.126	-.019	-.101	-.103	-.093	-.107	-.093	-.117	.600
.650															.650
.700	-.016	-.082		-.106	-.060	-.105	-.084	-.093							.700
.750									-.042	-.163	-.037	-.172	-.039	-.177	.750
.800	.021	-.129	.053	-.170	.034	-.172	-.005	-.159	.026	-.213	.075	-.218	.021	-.212	.800
.850															.850
.900	.077	-.148	.109	-.205	.108	-.210	.077	-.196							.900
.950															.950
a = .02															
.0125	.299	-.071	.350	-.178	.388	-.367	.367	-.225							.0125
.025	.248	-.108	.295	-.174	.324	-.205	.315	-.208	.345		.329	-.223		-.175	.025
.050	.199	-.089	.244	-.155	.261	-.177	.267	-.190							.050
.075	.172	-.085	.198	-.143	.206	-.167		-.178	.255	-.174	.250	-.175	.250	-.174	.075
.100	.143	-.047		-.132	.167	-.179	.179	-.154							.100
.150	.089	-.031	.097	-.108	.094	-.138	.105	-.143	.117	-.138	.122	-.144		-.146	.150
.200	.046	.003	.035	-.070	.047	-.111	.041	-.120							.200
.250	.015	.044	-.015	.049	-.031	-.073	.008	-.092	.071	-.108	.017	-.099	.001	-.097	.250
.300	.015	.025	-.057	.083	-.068	.067	-.058	-.028							.300
.350									-.086	.015	-.072	-.021	-.		



TABLE III.- Continued

PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(b)  $M = 2.01$  - Continued

x/c	Cp at $y/\frac{1}{2}$ of :												x/c		
	.05		.20		.35		.50		.70		.825			.95	
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower		Upper	Lower
$\alpha = 12$															
.0125	-.045	.271	-.087	.341	-.081	.359	-.123	.443	-.175	.549	-.206	.542	.524	.0125	
.025	-.069	.225	-.097	.305	-.084	.328	-.134	.405	-.186	.478	-.189	.493	.505	.025	
.050	-.062	.198	-.106	.252	-.109	.301	-.123	.417	-.155	.494	-.189	.493	.505	.050	
.075	-.052	.198	-.106	.252	-.109	.301	-.123	.417	-.155	.494	-.189	.493	.505	.075	
.100	-.059	.200	-.105	.259	-.123	.312	-.129	.419	-.167	.478	-.186	.475	.479	.100	
.150	-.085	.208	-.150	.257	-.157	.356	-.155	.427	-.167	.478	-.186	.475	.479	.150	
.200	-.112	.246	-.177	.323	-.193	.398	-.189	.432	-.185	.494	-.221	.446	.459	.200	
.250	-.124	.285	-.196	.360	-.215	.412	-.216	.435	-.185	.494	-.221	.446	.459	.250	
.300	-.112	.343	-.217	.378	-.232	.424	-.253	.469	-.249	.454	-.248	.417	.429	.300	
.350	-.216	.346	-.249	.396	-.261	.395	-.265	.420	-.268	.383	-.267	.387	.400	.350	
.400	-.216	.346	-.249	.396	-.261	.395	-.265	.420	-.268	.383	-.267	.387	.400	.400	
.450	-.200	.306	-.245	.336	-.280	.351	-.282	.371	-.272	.301	-.276	.308	.314	.450	
.500	-.200	.306	-.245	.336	-.280	.351	-.282	.371	-.272	.301	-.276	.308	.314	.500	
.550	-.192	.268	-.223	.251	-.280	.249	-.283	.260	-.264	.201	-.266	.185	.198	.550	
.600	-.192	.268	-.223	.251	-.280	.249	-.283	.260	-.264	.201	-.266	.185	.198	.600	
.650	-.177	.180		.133	-.258	.134	-.252	.140	-.252	.089	-.253	.085	.068	.650	
.700	-.177	.180		.133	-.258	.134	-.252	.140	-.252	.089	-.253	.085	.068	.700	
.750	-.130	.042	-.158	.032	-.222	.020	-.238	.036	-.225	.005	-.136	.004	-.245	.750	
.800	-.130	.042	-.158	.032	-.222	.020	-.238	.036	-.225	.005	-.136	.004	-.245	.800	
.850	-.089	.018	-.111	-.035	-.155	-.038	-.214	-.022						.850	
.900	-.089	.018	-.111	-.035	-.155	-.038	-.214	-.022						.900	
.950	-.089	.018	-.111	-.035	-.155	-.038	-.214	-.022						.950	
$\alpha = 14$															
.0125	-.084	.325	-.126	.412	-.138	.425	-.178	.534	-.227	.549	-.242	.585	.557	.0125	
.025	-.105	.285	-.140	.375	-.138	.402	-.183	.490	-.227	.549	-.242	.585	.557	.025	
.050	-.092	.263	-.137	.334	-.144	.402	-.171	.490	-.201	.549	-.227	.542	.543	.050	
.075	-.087	.243	-.142	.313	-.152	.378	-.170	.476	-.201	.549	-.227	.542	.543	.075	
.100	-.097	.244		.312	-.163	.386	-.170	.481	-.208	.527	-.220	.515	.515	.100	
.150	-.122	.253	-.188	.330	-.189	.427	-.187	.481	-.208	.527	-.220	.515	.515	.150	
.200	-.133	.301	-.206	.382	-.221	.458	-.214	.481	-.217	.539	-.246	.479	.492	.200	
.250	-.137	.369	-.224	.440											

TABLE III.- Concluded  
PRESSURE COEFFICIENTS FOR REFLEX CAMBERED WING

(b)  $M = 2.01$  - Concluded

x/c	Cp at y/D of :														x/c
	.05		.20		.35		.50		.70		.825		.95		
	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	Upper	Lower	
α = 20															
.0125	-.167	.517	-.220	.619	-.249	.649	-.294	.702							.0125
.025	-.186	.474	-.228	.577	-.238	.649	-.289	.662	-.299		-.288	.645		.602	.025
.050	-.169	.421	-.223	.521	-.245	.626	-.279	.662						.617	.050
.075	-.176	.413	-.224	.511	-.245	.608		.650	-.285	.663	-.285	.635	-.291		.075
.100	-.184	.414		.509	-.245	.605	-.272	.658							.100
.150	-.164	.412	-.248	.528	-.256	.621	-.278	.639	-.287	.651	-.267	.608		.585	.150
.200	-.172	.470	-.263	.589	-.269	.631	-.284	.645							.200
.250	-.188	.511	-.274	.608	-.282	.622	-.298	.630	-.288	.637	-.281	.572	-.288	.577	.250
.300	-.175	.583	-.283	.614	-.291	.616	-.302	.638							.300
.350									-.298	.588	-.280	.541	-.292	.546	.350
.400	-.260	.585	-.297	.591	-.303	.575	-.309	.583							.400
.450									-.299	.515	-.284	.503	-.290	.507	.450
.500	-.238	.540	-.297	.536	-.309	.513	-.309	.516	-.299	.432	-.284	.435	-.296	.428	.500
.550															.550
.600	-.238	.459	-.275	.418	-.309	.402	-.312	.400	-.297	.336	-.281	.314	-.289	.319	.600
.650															.650
.700	-.219	.349		.307	-.301	.283	-.305	.275	-.294	.216	-.282	.213	-.289	.189	.700
.750															.750
.800	-.179	.200	-.268	.173	-.290	.149	-.300	.163	-.292	.128	-.151	.125	-.287	.121	.800
.850															.850
.900	-.142	.167	-.235	.103	-.260	.083	-.294	.102							.900
.950															.950

TABLE IV  
PRESSURE COEFFICIENTS FOR FLAT WING

[From reference 1]

(a)  $M = 1.61$

X/C		FRACTION OF SEMISPAN												X/C
		0.050		0.200		0.350		0.500		0.700		0.900		
		U	L	U	L	U	L	U	L	U	L	U	L	
		a = 0.0												
.0125	.133			.125		.128		.167		.043				.0125
.0250	.082			.055		.066		.036						.0250
.0500	.069			.051		.053		.044				.036		.0500
.1000	.059			.034		.017		.038		.053		.017		.1000
.1500	.049			.036		.023		.018		.006		-.006		.1500
.2000	.037			.013		.017		.006		-.023		-.020		.2000
.2500	.041			.013		.009		-.013		-.019		-.041		.2500
.3000	.022			.015		-.011		-.028		-.043		-.077		.3000
.3500	.006			-.011		-.030		-.035		-.064		-.086		.3500
.4000	.005			-.023		-.049		-.045		-.065		-.082		.4000
.4500	.028			-.029		-.057		-.071		-.087				.4500
.5000	.017			-.029		-.064		-.089		-.102		-.114		.5000
.5500	.037			-.049		-.075		-.092		-.112		-.123		.5500
.6000	.036			-.076		-.086		-.100		-.122		-.128		.6000
.6500	.047			-.072		-.092		-.105		-.130		-.135		.6500
.7000	.055			-.067		-.105		-.099		-.123		-.139		.7000
.7500	.067			-.073		-.090		-.106		-.132		-.145		.7500
.8000	.055			-.074		-.084		-.102		-.134		-.153		.8000
.8500	.053			-.073		-.082		-.102		-.130		-.157		.8500
.9000	.055			-.071		-.082		-.106		-.131				.9000
.9500	.057													.9500
a = 0.2														
.0125	.064	.199	.034	.212	.003	.246	.023	.297		-.150	.169			.0125
.0250	.023	.140	-.041	.139	-.059	.167	-.118	.165		-.116	.150			.0250
.0500	.026	.112	-.021	.123	-.047	.131	-.083	.137		-.116	.150	-.142	.143	.0500
.1000	.023	.100	-.030	.088	-.040	.080	-.044	.104	-.071	.124	-.128	.097		.1000
.1500	.015	.093	-.004	.078	-.031	.079	-.061	.089	-.061	.071	-.116	.063	.1500	
.2000	-.002	.071	-.032	.067	-.031	.082	-.070	.083	.059	.059	-.106	.043	.2000	
.2500	.008	.074	-.029	.055	-.041	.063	-.067	.045	-.092	.042	-.104	.051	.2500	
.3000	-.007	.055	-.017	.047	-.057	.031	-.082	.027	-.102	.008	-.114	-.013	.3000	
.3500	-.027	.039	-.047	.028	-.065	.010	-.088	.010	-.121	-.020	-.144	-.020	.3500	
.4000	-.027	.039	-.059	.011	-.087	-.001	-.094	-.001	-.125	-.018	-.159	-.024	.4000	
.4500	-.005	.057	-.064	.019	-.102	-.013	-.104	-.018	-.138	-.036			.4500	
.5000	-.048	.011	-.064	.005	-.106	-.022	-.132	-.036	-.150	-.053	-.190	-.065	.5000	
.5500	-.065	-.009	-.081	-.018	-.116	-.035	-.133	-.044	-.159	-.066				

TABLE IV.- Continued  
PRESSURE COEFFICIENTS FOR FLAT WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U		U		U		U		U		U			
	$\alpha = 08$													
.0125	-.124	.386	-.225	.429	-.268	.460	-.237	.469	-.364	.398			.0125	
.0250	-.162	.325	-.272	.368	-.318	.394	-.353	.395	-.368	.349			.0250	
.0500	-.128	.253	-.281	.289	-.313	.323	-.355	.346	-.345	.293	-.378	.348	.0500	
.1000	-.072	.234	-.262	.246	-.301	.274	-.332	.279	-.334	.244	-.366	.286	.1000	
.1500	-.075	.224	-.225	.228	-.286	.255	-.314	.251	-.334	.244	-.363	.246	.1500	
.2000	-.087	.198	-.159	.221	-.278	.240	-.313	.234	-.330	.232	-.358	.214	.2000	
.2500	-.083	.202	-.142	.182	-.272	.201	-.309	.178	-.328	.206	-.355	.181	.2500	
.3000	-.090	.170	-.133	.178	-.264	.172	-.311	.178	-.322	.182	-.356	.169	.3000	
.3500	-.107	.165	-.132	.157	-.214	.154	-.313	.163	-.337	.148	-.358	.152	.3500	
.4000	-.111	.153	-.155	.141	-.200	.137	-.311	.144	-.337	.148	-.358	.141	.4000	
.4500	-.096	.173	-.165	.130	-.196	.124	-.315	.125	-.346	.108			.4500	
.5000	-.119	.128	-.165	.115	-.200	.104	-.321	.103	-.350	.099	-.368	.095	.5000	
.5500	-.145	.105	-.168	.086	-.212	.076	-.321	.083	-.352	.078	-.370	.095	.5500	
.6000	-.140	.104	-.182	.067	-.221	.059	-.301	.062	-.355	.059	-.369	.077	.6000	
.6500	-.151	.092	-.184	.069	-.223	.065	-.270	.058	-.357	.048	-.371	.058	.6500	
.7000	-.151	.076	-.176	.072	-.225	.047	-.255	.059	-.354	.060	-.374	.054	.7000	
.7500	-.141	.054	-.182	.062	-.210	.073	-.255	.054	-.358	.045	-.372	.044	.7500	
.8000	-.139	.066	-.181	.059	-.207	.065	-.243	.055	-.358	.048	-.375	.040	.8000	
.8500	-.143	.070	-.174	.067	-.204	.064	-.239	.062	-.358	.046	-.376	.033	.8500	
.9000	-.147	.071	-.168	.070	-.205	.064	-.237	.055	-.357	.044			.9000	
.9500	-.155	.067											.9500	
$\alpha = 10$														
.0125	-.168	.450	-.283	.491	-.324	.515	-.307	.505	-.408	.453			.0125	
.0250	-.207	.392	-.321	.437	-.374	.455	-.392	.444	-.414	.402			.0250	
.0500	-.181	.318	-.325	.346	-.368	.390	-.396	.404	-.387	.348	-.416	.392	.0500	
.1000	-.092	.301	-.310	.313	-.350	.334	-.378	.333	-.378	.301	-.404	.293	.1000	
.1500	-.101	.275	-.284	.296	-.340	.314	-.359	.316	-.378	.281	-.399	.261	.1500	
.2000	-.111	.250	-.264	.276	-.333	.302	-.359	.292	-.374	.264	-.396	.234	.2000	
.2500	-.111	.254	-.254	.276	-.333	.302	-.359	.292	-.374	.264	-.396	.234	.2500	
.3000	-.107	.218	-.173	.232	-.333	.222	-.358	.239	-.378	.235	-.394	.220	.3000	
.3500	-.128	.214	-.161	.210	-.329	.211	-.357	.220	-.380	.199	-.395	.210	.3500	
.4000	-.132	.206	-.175	.196	-.292	.192	-.357	.195	-.378	.189	-.394	.199	.4000	
.4500	-.120	.223	-.183	.180	-.259	.172	-.361	.170	-.380	.150			.4500	
.5000	-.136	.167	-.185	.160	-.247	.149	-.365	.145	-.384	.141	-.403	.145	.5000	
.5500	-.161	.154	-.186	.135	-.247	.119	-.368	.128	-.387	.115	-.404	.141	.5500	
.6000	-.159	.148	-.201	.115	-.253	.109	-.368	.119	-.390	.105	-.404	.123	.6000	
.6500	-.168	.135	-.199	.114	-.251	.111	-.365	.110	-.391	.107	-.406	.109	.6500	
.7000	-.168	.113	-.193	.115	-.252	.099	-.359	.109	-.388	.119	-.406	.113	.7000	
.7500	-.157	.093	-.199	.105	-.240	.113	-.330	.108	-.391	.095	-.407	.106	.7500	
.8000	-.156	.109	-.195	.101	-.233	.109	-.303	.103	-.392	.099	-.408	.101	.8000	
.8500	-.156	.110	-.192	.113	-.228	.107	-.290	.103	-.391	.103	-.407	.090	.8500	
.9000	-.168	.115	-.186	.112	-.224	.108	-.281	.097	-.388	.092			.9000	
.9500	-.172	.110											.9500	
$\alpha = 12$														
.0125	-.215	.508	-.333	.538	-.375	.545	-.367	.526	-.448	.488			.0125	
.0250	-.249	.442	-.371	.482	-.406	.498	-.433	.493	-.451	.442			.0250	
.0500	-.240	.368	-.373	.400	-.445	.445	-.452	.452	-.451	.393	-.457	.443	.0500	
.1000	-.116	.351	-.357	.359	-.390	.384	-.413	.384	-.419	.346	-.443	.349	.1000	
.1500	-.125	.317	-.331	.343	-.378	.375	-.400	.347	-.416	.332	-.440	.320	.1500	
.2000	-.134	.296	-.308	.316	-.371	.352	-.398	.347	-.416	.332	-.440	.320	.2000	
.2500	-.135	.301	-.269	.275	-.367	.306	-.396	.311	-.414	.311	-.436	.295	.2500	
.3000	-.126	.262	-.220	.278	-.370	.273	-.395	.286	-.414	.278	-.434	.284	.3000	
.3500	-.151	.251	-.224	.257	-.371	.259	-.394	.264	-.416	.243	-.434	.261	.3500	
.4000	-.155	.244	-.219	.239	-.372	.239	-.394	.241	-.416	.233	-.434	.248	.4000	
.4500	-.145	.264	-.212	.219	-.341	.213	-.397	.210	-.417	.197			.4500	
.5000	-.155	.204	-.208	.192	-.307	.188	-.400	.186	-.420	.193	-.436	.202	.5000	
.5500	-.185	.197	-.211	.180	-.300	.168	-.402	.174	-.420	.172	-.437	.194	.5500	
.6000	-.180	.185	-.222	.155	-.293	.154	-.405	.159	-.421	.158	-.437	.174	.6000	
.6500	-.190	.172	-.219	.151	-.292	.160	-.399	.153	-.424	.157	-.430	.164	.6500	
.7000	-.188	.148	-.211	.156	-.290	.147	-.402	.156	-.422	.165	-.428	.169	.7000	
.7500	-.175	.127	-.220	.142	-.275	.156	-.398	.152	-.425	.147	-.425	.160	.7500	
.8000	-.178	.146	-.215	.141	-.266	.154	-.361	.148	-.423	.143	-.425	.162	.8000	
.8500	-.181	.146	-.207	.158	-.259	.154	-.293	.144	-.420	.142	-.422	.156	.8500	
.9000	-.185	.152	-.201	.156	-.253	.152	-.256	.139	-.429	.140			.9000	
.9500	-.191	.146											.9500	
$\alpha = 14$														
.0125	-.269	.565	-.375	.575	-.420	.564	-.424	.533	-.474	.515			.0125	
.0250	-.285	.499	-.404	.535	-.444	.545	-.465	.532	-.480	.435			.0250	
.0500	-.292	.427	-.402	.466	-.442	.495	-.463	.499	-.463	.435	-.481	.497	.0500	
.1000	-.173	.401	-.389	.417	-.429	.438	-.448	.426	-.456	.400	-.478	.444	.1000	
.1500	-.149	.369	-.361	.408	-.416	.429	-.436	.417	-.447	.400	-.468	.397	.1500	
.2000	-.153	.348	-.361	.369	-.410	.407	-.431	.400	-.446	.387	-.463	.378	.2000	
.2500	-.156	.356	-.330	.329	-.407	.353	-.429	.358	-.443	.354	-.458	.366	.2500	
.3000	-.146	.314	-.268	.330	-.408	.326	-.426	.333	-.440	.325	-.459	.358	.3000	
.3500	-.169	.301	-.227	.308	-.410	.308	-.424	.304	-.443	.292	-.456	.339	.3500	
.4000	-.177	.294	-.258	.286	-.411	.281	-.424	.282	-.441	.285	-.456	.331	.4000	
.4500	-.167	.315	-.265	.262	-.407	.253	-.426	.261	-.443	.253			.4500	
.5000	-.177	.246	-.251	.237	-.391	.237	-.429	.235	-.447	.233	-.458	.283	.5000	
.5500	-.200	.245	-.241	.225	-.368	.211	-.429	.220	-.446	.239	-.458	.277	.5500	
.6000	-.198	.234	-.244	.201	-.341	.200	-.430	.205	-.446	.236	-.454	.262	.6000	
.6500	-.208	.219	-.240	.197	-.340	.205	-.424	.203	-.446	.233	-.454	.261	.6500	
.7000	-.203	.195	-.227	.195	-.334	.182	-.432	.207	-.445	.244	-.453	.274	.7000	
.7500	-.188	.169	-.236	.181	-.313	.201	-.434	.206	-.449	.229	-.451	.284	.7500	
.8000	-.194	.193	-.228	.188	-.301	.197	-.424	.213	-.444	.225	-.453	.304	.8000	
.8500	-.196	.190	-.228	.198	-.295	.201	-.346	.220	-.368	.231	-.448	.321	.8500	
.9000	-.201	.195	-.222	.199	-.291	.211	-.301	.220	-.330	.221			.9000	
.9500	-.208	.189											.9500	







TABLE IV.- Continued  
PRESSURE COEFFICIENTS FOR FLAT WING

[From reference 1]

(b) M = 2.01 - Continued

x/c	FRACTION OF SEMISPAN												x/c	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U		U		U		U		U		U			
	L		L		L		L		L		L			
$\alpha = 08$														
.0125	-.018	.384	-.059	.422	-.080	.453	-.045	.477	-.174	.392	-.185	.354	.0125	
.0250	-.088	.297	-.119	.346	-.135	.374	-.159	.384	-.184	.350	-.182	.300	.0250	
.0500	-.101	.264	-.140	.298	-.144	.322	-.165	.337	-.184	.301	-.185	.259	.0500	
.1000	-.077	.196	-.145	.227	-.152	.248	-.159	.279	-.170	.242	-.183	.233	.1000	
.1500	-.063	.180	-.154	.200	-.148	.237	-.154	.249	-.166	.242	-.185	.259	.1500	
.2000	-.076	.161	-.155	.185	-.153	.213	-.159	.233	-.170	.230	-.183	.233	.2000	
.2500	-.082	.150	-.146	.161	-.161	.188	-.164	.202	-.174	.204	-.184	.201	.2500	
.3000	-.073	.150	-.124	.160	-.169	.183	-.169	.173	-.181	.180	-.187	.177	.3000	
.3500	-.106	.144	-.124	.141	-.178	.142	-.175	.159	-.189	.158	-.193	.169	.3500	
.4000	-.102	.131	-.129	.121	-.184	.128	-.179	.142	-.193	.146	-.193	.158	.4000	
.4500	-.100	.122	-.133	.106	-.187	.111	-.185	.118	-.200	.119	-.208	.119	.4500	
.5000	-.116	.094	-.137	.095	-.193	.096	-.194	.099	-.205	.105	-.212	.107	.5000	
.5500	-.119	.094	-.148	.079	-.192	.077	-.199	.085	-.208	.091	-.212	.107	.5500	
.6000	-.122	.085	-.155	.059	-.181	.059	-.204	.069	-.213	.076	-.216	.095	.6000	
.6500	-.124	.078	-.155	.061	-.178	.057	-.209	.058	-.216	.066	-.217	.086	.6500	
.7000	-.123	.073	-.150	.066	-.182	.039	-.209	.058	-.215	.073	-.217	.079	.7000	
.7500	-.127	.047	-.152	.054	-.174	.054	-.214	.047	-.218	.054	-.220	.066	.7500	
.8000	-.112	.042	-.149	.055	-.173	.053	-.220	.042	-.219	.049	-.223	.058	.8000	
.8500	-.116	.053	-.144	.052	-.170	.054	-.218	.039	-.219	.046	-.225	.052	.8500	
.9000	-.120	.056	-.142	.046	-.170	.050	-.221	.038	-.222	.039			.9000	
.9500	-.124	.054											.9500	
$\alpha = 10$														
.0125	-.050	.438	-.115	.475	-.131	.502	-.105	.514	-.218	.446	-.227	.407	.0125	
.0250	-.125	.354	-.162	.402	-.182	.433	-.201	.440	-.225	.405	-.226	.350	.0250	
.0500	-.143	.302	-.178	.344	-.189	.378	-.211	.396	-.212	.346	-.227	.309	.0500	
.1000	-.115	.242	-.182	.281	-.192	.308	-.201	.334	-.212	.301	-.227	.278	.1000	
.1500	-.085	.218	-.189	.250	-.187	.296	-.195	.301	-.207	.291	-.227	.278	.1500	
.2000	-.098	.207	-.193	.233	-.191	.266	-.196	.282	-.208	.284	-.221	.278	.2000	
.2500	-.102	.195	-.191	.207	-.195	.241	-.201	.255	-.212	.262	-.222	.247	.2500	
.3000	-.093	.194	-.174	.207	-.201	.211	-.208	.226	-.217	.232	-.224	.231	.3000	
.3500	-.127	.184	-.158	.186	-.207	.193	-.211	.208	-.222	.206	-.226	.218	.3500	
.4000	-.119	.172	-.156	.163	-.212	.172	-.212	.189	-.226	.193	-.227	.207	.4000	
.4500	-.122	.163	-.157	.145	-.219	.156	-.219	.165	-.231	.161	-.237	.162	.4500	
.5000	-.135	.135	-.159	.136	-.222	.138	-.223	.144	-.235	.150	-.241	.152	.5000	
.5500	-.136	.135	-.167	.118	-.227	.117	-.228	.127	-.237	.134	-.241	.141	.5500	
.6000	-.139	.122	-.178	.100	-.235	.100	-.235	.112	-.240	.120	-.242	.141	.6000	
.6500	-.142	.114	-.175	.100	-.227	.095	-.238	.102	-.244	.109	-.245	.127	.6500	
.7000	-.143	.111	-.168	.102	-.219	.079	-.238	.099	-.242	.115	-.246	.123	.7000	
.7500	-.143	.078	-.171	.092	-.204	.096	-.241	.089	-.242	.095	-.248	.112	.7500	
.8000	-.127	.078	-.168	.089	-.198	.094	-.243	.083	-.244	.090	-.246	.104	.8000	
.8500	-.132	.087	-.163	.089	-.195	.093	-.242	.082	-.243	.088	-.236	.097	.8500	
.9000	-.138	.090	-.159	.084	-.193	.089	-.246	.082	-.246				.9000	
.9500	-.141	.086											.9500	
$\alpha = 12$														
.0125	-.083	.492	-.156	.520	-.169	.540	-.156	.541	-.248	.492	-.254	.453	.0125	
.0250	-.153	.417	-.196	.459	-.213	.489	-.229	.490	-.253	.455	-.251	.396	.0250	
.0500	-.175	.351	-.207	.388	-.218	.430	-.234	.447	-.253	.396	-.251	.350	.0500	
.1000	-.142	.294	-.209	.332	-.216	.358	-.227	.381	-.241	.345	-.249	.323	.1000	
.1500	-.104	.263	-.213	.305	-.213	.342	-.213	.352	-.237	.335	-.245	.323	.1500	
.2000	-.113	.252	-.216	.284	-.215	.313	-.222	.328	-.237	.313	-.245	.296	.2000	
.2500	-.120	.236	-.216	.254	-.219	.287	-.226	.299	-.237	.283	-.246	.282	.2500	
.3000	-.111	.239	-.207	.256	-.224	.257	-.230	.272	-.239	.254	-.249	.267	.3000	
.3500	-.142	.229	-.196	.231	-.228	.237	-.231	.255	-.244	.254	-.249	.267	.3500	
.4000	-.135	.215	-.186	.207	-.232	.216	-.234	.232	-.245	.241	-.249	.257	.4000	
.4500	-.135	.205	-.181	.187	-.237	.196	-.237	.203	-.249	.207	-.256	.211	.4500	
.5000	-.150	.176	-.178	.178	-.241	.179	-.241	.182	-.254	.199	-.257	.204	.5000	
.5500	-.150	.177	-.184	.158	-.245	.155	-.245	.166	-.255	.180	-.257	.204	.5500	
.6000	-.154	.160	-.192	.139	-.249	.138	-.249	.152	-.258	.164	-.260	.188	.6000	
.6500	-.155	.154	-.189	.138	-.250	.137	-.251	.140	-.260	.154	-.262	.177	.6500	
.7000	-.155	.150	-.184	.142	-.246	.119	-.252	.138	-.258	.160	-.260	.169	.7000	
.7500	-.154	.112	-.188	.130	-.234	.134	-.256	.131	-.260	.130	-.256	.156	.7500	
.8000	-.141	.116	-.183	.128	-.225	.131	-.261	.129	-.260	.136	-.251	.152	.8000	
.8500	-.146	.127	-.177	.124	-.216	.131	-.260	.126	-.260	.134	-.246	.143	.8500	
.9000	-.150	.129	-.173	.117	-.214	.125	-.261	.123	-.262	.126			.9000	
.9500	-.153	.122											.9500	
$\alpha = 14$														
.0125	-.122	.546	-.193	.563	-.212	.576	-.206	.567	-.276	.535	-.282	.489	.0125	
.0250	-.183	.478	-.229	.511	-.246	.535	-.260	.539	-.281	.492	-.278	.441	.0250	
.0500	-.204	.394	-.239	.430	-.249	.475	-.263	.494	-.281	.443	-.276	.401	.0500	
.1000	-.184	.344	-.239	.384	-.245	.410	-.258	.428	-.280	.394	-.276	.364	.1000	
.1500	-.121	.310	-.239	.356	-.241	.394	-.249	.404	-.260	.381	-.272	.364	.1500	
.2000	-.131	.296	-.242	.332	-.243	.368	-.248	.378	-.262	.354	-.271	.341	.2000	
.2500	-.134	.278	-.240	.302	-.244	.331	-.252	.349	-.263	.325	-.271	.325	.2500	
.3000	-.130	.283	-.233	.302	-.248	.302	-.253	.319	-.263	.325	-.271	.325	.3000	
.3500	-.153	.273	-.225	.275	-.253	.283	-.257	.297	-.267	.297	-.272	.313	.3500	
.4000	-.152	.256	-.222	.245	-.256	.260	-.258	.277	-.267	.281	-.271	.301	.4000	
.4500	-.149	.245	-.213	.230	-.261	.238	-.263	.249	-.273	.284			.4500	
.5000	-.164	.217	-.206	.219	-.260	.221	-.264	.226	-.275				.5000	
.5500	-.164	.213	-.207	.199	-.267	.197	-.266	.209	-.274	.221	-.271	.249	.5500	
.6000	-.168	.200	-.213	.178	-.271	.180	-.271	.192	-.277	.206	-.271	.227	.6000	
.6500	-.170	.194	-.210	.177	-.270	.177	-.272	.181	-.279	.194	-.271	.215	.6500	
.7000	-.169	.189	-.206	.178	-.267	.156	-.272	.181	-.278	.202	-.270	.210	.7000	
.7500	-.168	.147	-.207	.167	-.263	.174	-.275	.173	-.278	.180	-.267	.200	.7500	
.8000	-.155	.154	-.205	.165	-.259	.170	-.278	.171	-.275	.176	-.264	.196	.8000	
.8500	-.160	.155	-.201	.160	-.255	.169	-.278	.166	-.271	.175	-.261	.186	.8500	
.9000	-.164	.166	-.196	.154	-.252	.162	-.278	.162	-.271	.168			.9000	
.9500	-.169	.162											.9500	



TABLE V  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a)  $M = 1.61$

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = -20$													
.0125	.905	-.465			.593	-.474	.522	-.475					.0125	
.0250	.838	-.481			.646	-.476	.621	-.474					.0250	
.0500	.834	-.498	.702		.651	-.479	.636	-.473	.599	-.464	.639	-.313	.0500	
.1000	.722	-.469	.647		.620	-.481	.609	-.472	.631	-.464	.639	-.309	.1000	
.1500	.623	-.423	.580		.472	.572	-.482	.591	-.472	.610	-.461	.641	-.301	.1500
.2000	.535	-.317	.529		.484	.539	-.482	.541	-.470	.553	-.462	.590	-.297	.2000
.2500	.518		.452		.492	.466	-.483	.472	-.474	.491	-.461	.563	-.294	.2500
.3000	.509	-.310	.385		.497	.432	-.483	.427	-.475	.470	-.461	.551	-.290	.3000
.3500	.454		.382		.501	.420	-.487	.412	-.473	.441	-.456	.535	-.288	.3500
.4000	.387		.337		.503	.407	-.491	.390	-.478	.438	-.438	.491	-.283	.4000
.4500	.369		.344		.509	.353	-.497	.363		.433	-.419	.458	-.289	.4500
.5000	.323	-.213	.348		.508	.345	-.493	.331	-.485	.424	-.410	.445	-.318	.5000
.6000	.284	-.210	.320		.492	.313	-.490	.303	-.487	.422	-.422	.437	-.355	.6000
.7000	.303	-.214	.294		.434	.294	-.481	.360	-.483	.451	-.425	.403	-.322	.7000
.8000	.333	-.202	.329		.362		-.456	.438	-.445	.433	-.402	.398	-.293	.8000
.9000	.358	-.220	.367		-.275	.405	-.425	.450	-.387	.410	-.379		-.9000	
$\alpha = -18$														
.0125	.879	-.485			.584	-.471	.503	-.461					.0125	
.0250	.794	-.493			.633	-.472	.597	-.459	.585	-.441			.0250	
.0500	.779	-.506	.664		.624	-.473	.582	-.458	.575	-.437	.599	-.287	.0500	
.1000	.663	-.432	.592		.574	-.472	.527	-.457	.566	-.443	.588	-.283	.1000	
.1500	.557	-.393	.525		.470	.515	.471	.495	-.457	.525	-.440	.578	-.276	.1500
.2000	.474	-.281	.478		.481	.456	.442	-.454	.457	-.441	.501	-.273	.2000	
.2500	.451		.403		.480	.384	-.473	.388	-.456	.379	-.441	.459	-.270	.2500
.3000	.457	-.299	.328		.479	.341	-.474	.349	-.456	.352	-.439	.440	-.267	.3000
.3500	.397		.317		.487	.310	-.479	.340	-.456	.311	-.438	.433	-.264	.3500
.4000	.332		.284		.499	.295	-.481	.306	-.461	.305	-.429	.397	-.260	.4000
.4500	.318		.259		.507	.251	-.489	.274		.311	-.407	.377	-.259	.4500
.5000	.272	-.196	.245		.500	.254	-.487	.242	-.468	.300	-.395	.371	-.262	.5000
.6000	.206	-.196	.197		.466	.223	-.489	.218	-.473	.256	-.403	.393	-.243	.6000
.7000	.198	-.204	.214		.366	.224	-.478	.214	-.475	.330	-.415	.378	-.227	.7000
.8000	.226	-.179	.244		.259		-.451	.254	-.441	.372	-.387	.382	-.264	.8000
.9000	.278	-.203	.283		-.194	.276	-.431	.325	-.383	.386	-.357		-.9000	
$\alpha = -16$														
.0125	.835	-.489			.577	-.460	.494	-.444					.0125	
.0250	.754	-.494			.613	-.461	.577	-.443	.560	-.425			.0250	
.0500	.723	-.509	.624		.590	-.461	.549	-.443	.522	-.422	.518	-.283	.0500	
.1000	.609	-.421	.541		.529	-.461	.484	-.443	.501	-.425	.495	-.278	.1000	
.1500	.484	-.335	.467		.463	.466	.460	.448	.452	-.424	.471	-.272	.1500	
.2000	.413	-.246	.427		.474	.402	-.459	.379	-.440	.377	-.425	.384	-.267	.2000
.2500	.393		.357		.469	.327	-.466	.309	-.441	.299	-.425	.346	-.263	.2500
.3000	.401	-.254	.277		.461	.289	-.470	.260	-.446	.276	-.423	.327	-.258	.3000
.3500	.343		.258		.470	.262	-.469	.241	-.448	.236	-.422	.314	-.256	.3500
.4000	.283		.237		.487	.236	-.466	.210	-.456	.223	-.420	.266	-.253	.4000
.4500	.264		.210		.491	.178	-.464	.189		.199	-.407	.231	-.252	.4500
.5000	.222	-.179	.198		.476	.173	-.468	.163	-.463	.172	-.391	.223	-.250	.5000
.6000	.161	-.179	.134		.394	.107	-.477	.113	-.464	.113	-.390	.244	-.240	.6000
.7000	.145	-.188	.124		.224	.127	-.474	.129	-.463	.156	-.383	.259	-.237	.7000
.8000	.143	-.163	.133		.163		-.453	.153	-.442	.179	-.377	.282	-.244	.8000
.9000	.174	-.190	.201		-.159	.187	-.433	.188	-.390	.240	-.345		-.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN												X/C
	0.050		0.200		0.350		0.500		0.700		0.900		
	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha = -12$												
.0125	.721	-.476			.551	-.402	.485	-.394					.0125
.0250	.658	-.481		-.427	.567	-.404	.538	-.392	.531	-.382			.0250
.0500	.603	-.498	.545	-.428	.527	-.407	.497	-.392	.473	-.378	.452	-.304	.0500
.1000	.491	-.436	.439	-.432	.448	-.417	.397	-.393	.426	-.381	.412	-.296	.1000
.1500	.370	-.204	.344	-.464	.369	-.416	.342	-.396	.354	-.378	.366	-.286	.1500
.2000	.303	-.145	.302	-.461	.297	-.419	.273	-.394	.274	-.380	.244	-.281	.2000
.2500	.287		.268	-.440	.228	-.448	.214	-.398	.187	-.380	.191	-.275	.2500
.3000	.293	-.168	.186	-.418	.176	-.452		-.402	.145	-.381	.158	-.269	.3000
.3500	.246		.157	-.417	.162	-.452	.153	-.407	.115	-.377	.150	-.266	.3500
.4000	.192		.148	-.436	.151	-.452	.114	-.417	.103	-.377	.107	-.259	.4000
.4500	.173		.113	-.394	.063	-.445	.079		.069	-.379	.043	-.258	.4500
.5000	.130	-.129	.107	-.254	.086	-.433	.067	-.421	.061	-.374	.030	-.253	.5000
.6000	.080	-.136	.047	-.089	.020	-.408	.003	-.415	-.016	-.364	-.008	-.244	.6000
.7000	.061	-.143	.040	-.099	.012	-.391	-.015	-.412	-.020	-.349	-.028	-.239	.7000
.8000	.057	-.127	.039	-.118		-.371	.003	-.409	-.017	-.343	-.026	-.235	.8000
.9000	.074	-.149	.071	-.122	.035	-.356	.024	-.306	.003	-.334			.9000
$\alpha = -10$													
.0125		-.465			.536	-.374	.474	-.369					.0125
.0250		-.466		-.406	.538	-.377	.509	-.368	.506	-.350			.0250
.0500		-.478	.507	-.412	.480	-.379	.462	-.367	.440	-.347	.422	-.298	.0500
.1000		-.400	.385	-.419	.395	-.393	.350	-.368	.382	-.351	.385	-.293	.1000
.1500		-.120	.282	-.439	.314	-.389	.288	-.367	.304	-.348	.337	-.283	.1500
.2000	.252	-.113	.239	-.447	.244	-.400	.215	-.366	.230	-.350	.199	-.277	.2000
.2500	.235		.218	-.427	.182	-.432	.154	-.370	.141	-.347	.143	-.270	.2500
.3000	.236	-.122	.142	-.401	.123	-.430	.102	-.383	.099	-.345	.108	-.264	.3000
.3500	.197		.106	-.396	.098	-.425	.102	-.389	.068	-.344	.096	-.260	.3500
.4000	.147		.100	-.386	.110	-.414	.081	-.390	.066	-.347	.061	-.255	.4000
.4500	.129		.058	-.192	.005	-.408	.006		-.008	-.349	-.033	-.254	.4500
.5000	.087	-.099	.063	-.045	.040	-.390	.000	-.388	-.009	-.344	-.038	-.252	.5000
.6000	.038	-.107	.000	-.063	-.026	-.360	-.043	-.372	-.068	-.338	-.068	-.242	.6000
.7000	.024	-.113	-.004	-.080	-.036	-.342	-.060	-.365	-.068	-.323	-.076	-.237	.7000
.8000	.019	-.099	-.004	-.096		-.313	-.054	-.357	-.062	-.303	-.066	-.236	.8000
.9000	.035	-.121	.024	-.098	-.011	-.224	-.025	-.270	-.057	-.287			.9000
$\alpha = -08$													
.0125	.657	-.449			.518	-.346	.460	-.339					.0125
.0250	.580	-.447		-.384	.509	-.348	.482	-.336	.474	-.315			.0250
.0500	.498	-.459	.476	-.387	.445	-.350	.430	-.337	.407	-.315	.381	-.296	.0500
.1000	.393	-.330	.336	-.405	.339	-.361	.308	-.336	.336	-.314	.344	-.292	.1000
.1500	.267	-.081	.217	-.422	.260	-.359	.242	-.333	.252	-.313	.297	-.287	.1500
.2000	.201	-.098	.184	-.413	.190	-.379	.168	-.335	.173	-.314	.145	-.282	.2000
.2500	.187		.166	-.406	.139	-.403	.105	-.344	.085	-.313	.084	-.276	.2500
.3000	.187	-.097	.098	-.371	.077	-.394	.043	-.358	.040	-.310	.050	-.271	.3000
.3500	.152		.060	-.355	.043	-.382	.032	-.360	.010	-.309	.047	-.266	.3500
.4000	.103		.056	-.211	.063	-.364	.043	-.358	.028	-.313	.029	-.262	.4000
.4500	.085		.010	.010	-.037	-.350	-.028		-.037	-.310	-.064	-.263	.4500
.5000	.045	-.072	.022	-.000	-.016	-.336	-.050	-.344	-.045	-.309	-.077	-.264	.5000
.6000	-.000	-.079	-.038	-.049	-.064	-.311	-.084	-.325	-.114	-.310	-.116	-.263	.6000
.7000	-.015	-.088	-.043	-.056	-.078	-.284	-.099	-.318	-.125	-.298	-.128	-.259	.7000
.8000	-.019	-.076	-.039	-.074		-.169	-.096	-.308	-.117	-.274	-.126	-.261	.8000
.9000	-.004	-.093	-.016	-.079	-.052	-.057	-.074	-.282	-.094	-.256			.9000
$\alpha = -06$													
.0125	.611	-.430			.495	-.316	.441	-.302					.0125
.0250	.544	-.428		-.356	.469	-.315	.445	-.300	.438	-.273			.0250
.0500	.447	-.443	.427	-.359	.403	-.318	.380	-.300	.369	-.272	.331	-.266	.0500
.1000	.347	-.211	.285	-.384	.285	-.322	.251	-.298	.292	-.272	.293	-.265	.1000
.1500	.218	-.053	.165	-.403	.200	-.326	.194	-.294	.204	-.270	.259	-.265	.1500
.2000	.153	-.071	.134	-.389	.135	-.344	.127	-.296	.131	-.269	.101	-.264	.2000
.2500	.142		.113	-.366	.097	-.355	.064	-.307	.042	-.267	.023	-.260	.2500
.3000	.143	-.080	.060	-.331	.040	-.347	-.002	-.319	-.008	-.265	-.013	-.256	.3000
.3500	.108		.018	-.219	-.002	-.334	-.025	-.313	-.047	-.265	-.030	-.253	.3500
.4000	.063		.019	.032	.014	-.316	-.017	-.302	-.047	-.265	-.040	-.248	.4000
.4500	.044		-.035	.046	-.071	-.298	-.067		-.093	-.265	-.114	-.247	.4500
.5000	.008	-.045	-.016	.017	-.063	-.286	-.089	-.283	-.092	-.266	-.127	-.248	.5000
.6000	-.038	-.049	-.076	-.026	-.099	-.252	-.142	-.269	-.141	-.269	-.162	-.248	.6000
.7000	-.052	-.061	-.077	-.027	-.115	-.164	-.136	-.258	-.154	-.258	-.170	-.241	.7000
.8000	-.052	-.059	-.071	-.048		-.001	-.135	-.239	-.162	-.246	-.165	-.231	.8000
.9000	-.038	-.067	-.051	-.054	-.091	-.012	-.127	-.226	-.157	-.215			.9000

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN												X/C
	0.050		0.200		0.350		0.500		0.700		0.900		
	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha = -0.4$												
.0125	.507	-.409			.474	-.279	.424	-.260					.0125
.0250	.399	-.406			.433	-.280	.408	-.260					.0250
.0500	.299	-.426	.389	-.330	.357	-.283	.334	-.260	.404	-.228	.291	-.228	.0500
.1000	.299	-.075	.236	-.351	.232	-.283	.199	-.256	.244	-.227	.253	-.228	.1000
.1500	.173	-.022	.112	-.361	.145	-.293	.128	-.250	.150	-.226	.215	-.232	.1500
.2000	.111	-.039	.084	-.330	.079	-.304	.074	-.256	.086	-.225	.067	-.231	.2000
.2500	.100		.065	-.300	.047	-.309	.022	-.266	.005	-.221	-.023	-.229	.2500
.3000	.101	-.056	.019	-.250	.001	-.304	-.039	-.272	-.046	-.220	-.059	-.225	.3000
.3500	.068		-.022	.029	-.042	-.284	-.066	-.259	-.086	-.220	-.081	-.221	.3500
.4000	.023		-.021	.094	-.038	-.260	-.064	-.241	-.092	-.220	-.090	-.217	.4000
.4500	.008		-.075	.067	-.111	-.242	-.116		-.138	-.220	-.154	-.215	.4500
.5000	-.027	-.018	-.055	.031	-.103	-.235	-.134	-.230	-.143	-.218	-.170	-.214	.5000
.6000	-.070	-.020	-.109	.002	-.136	-.154	-.176	-.205	-.191	-.220	-.204	-.217	.6000
.7000	-.083	-.035	-.111	-.004	-.150	.003	-.188	-.193	-.199	-.208	-.213	-.211	.7000
.8000	-.073	-.038	-.105	.023		.034	-.170	-.171	-.199	-.205	-.215	-.200	.8000
.9000	-.063	-.045	-.081	-.034	-.124	-.002	-.162	-.154	-.193	-.175			.9000
$\alpha = -0.2$													
.0125	.537	-.392			.452	-.238	.400	-.212					.0125
.0250	.459	-.387		-.300	.405	-.240	.380	-.210	.365	-.179			.0250
.0500	.347	-.390	.353	-.303	.321	-.241	.295	-.211	.283	-.176	.252	-.180	.0500
.1000	.255	-.006	.199	-.322	.184	-.240	.152	-.208	.200	-.176	.206	-.179	.1000
.1500	.132	.001	.065	-.301	.087	-.251	.075	-.200	.094	-.176	.170	-.183	.1500
.2000	.074	-.010	.030	-.254	.028	-.258	.020	-.205	.026	-.174	.028	-.185	.2000
.2500	.060		.015	-.238	-.015	-.264	-.042	-.219	-.043	-.172	-.057	-.183	.2500
.3000	.065	-.034	-.024	-.096	-.044	-.243	-.073	-.218	-.089	-.170	-.099	-.179	.3000
.3500	.029		-.059	.128	-.075	-.216	-.101	-.201	-.122	-.171	-.117	-.174	.3500
.4000	-.014		-.062	.107	-.077	-.196	-.105	-.184	-.128	-.168	-.137	-.170	.4000
.4500	-.029		-.111	.083	-.149	-.177	-.148		-.179	-.168	-.193	-.168	.4500
.5000	-.060	.010	-.094	.053	-.143	-.147	-.169	-.160	-.184	-.164	-.206	-.164	.5000
.6000	-.102	.012	-.140	.026	-.181	-.013	-.216	-.136	-.232	-.163	-.241	-.166	.6000
.7000	-.114	-.006	-.143	.023	-.181	.085	-.220	-.113	-.239	-.149	-.247	-.164	.7000
.8000	-.099	-.017	-.136	.004		.049	-.218	-.092	-.238	-.151	-.248	-.161	.8000
.9000	-.094	-.016	-.113	-.011	-.157	.016	-.190	-.075	-.234	-.136			.9000
$\alpha = 0.0$													
.0125	.501	-.366			.426	-.186	.369	-.152					.0125
.0250	.414	-.359		-.262	.368	-.189	.343	-.152	.332	-.111			.0250
.0500	.297	-.316	.309	-.264	.282	-.188	.256	-.152	.242	-.111	.208	-.123	.0500
.1000	.212	.033	.154	-.278	.142	-.188	.101	-.147	.163	-.110	.157	-.124	.1000
.1500	.095	.035	.013	-.230	.033	-.197	.014	-.138	.048	-.108	.113	-.131	.1500
.2000	.033	.017	-.024	-.199	-.032	-.194	-.037	-.147	-.034	-.108	-.023	-.132	.2000
.2500	.025		-.037	-.130	-.071	-.194	-.089	-.157	-.087	-.105	-.100	-.126	.2500
.3000	.031	-.009	-.067	.138	-.100	-.163	-.135	-.147	-.130	-.101	-.149	-.120	.3000
.3500	-.009		-.095	.138	-.124	-.140	-.156	-.130	-.172	-.099	-.167	-.111	.3500
.4000	-.048		-.097	.134	-.118	-.115	-.153	-.104	-.179	-.094	-.181	-.102	.4000
.4500	-.063		-.146	.113	-.183	-.073	-.188		-.224	-.093	-.237	-.099	.4500
.5000	-.089	.046	-.132	.087	-.177	-.016	-.203	-.073	-.232	-.087	-.248	-.094	.5000
.6000	-.129	.051	-.170	.058	-.218	.107	-.251	-.037	-.273	-.083	-.285	-.092	.6000
.7000	-.140	.031	-.174	.056	-.216	.104	-.258	-.019	-.277	-.066	-.291	-.092	.7000
.8000	-.125	.015	-.164	.038		.061	-.255	.006	-.272	-.063	-.293	-.094	.8000
.9000	-.122	.022	-.140	.023	-.194	.039	-.240	.005	-.268	-.060			.9000
$\alpha = 0.2$													
.0125	.464	-.343			.402	-.129	.342	-.091					.0125
.0250	.377	-.334		-.222	.331	-.133	.306	-.091	.292	-.043			.0250
.0500	.262	-.241	.271	-.222	.249	-.133	.223	-.089	.202	-.042	.169	-.063	.0500
.1000	.167	.055	.120	-.224	.102	-.132	.065	-.084	.129	-.042	.120	-.065	.1000
.1500	.065	.062	-.029	-.173	-.011	-.140	-.029	-.076	-.003	-.041	.077	-.070	.1500
.2000	.001	.048	-.066	-.127	-.080	-.132	-.082	-.084	-.080	-.039	-.068	-.069	.2000
.2500	-.005		-.083	.093	-.122	-.124	-.132	-.091	-.130	-.034	-.130	-.058	.2500
.3000	.003	.023	-.108	.176	-.146	-.087	-.184	-.076	-.174	-.025	-.186	-.048	.3000
.3500	-.047		-.126	.148	-.169	-.061	-.204	-.048	-.209	-.022	-.210	-.035	.3500
.4000	-.081		-.132	.155	-.161	-.015	-.203	-.018	-.218	-.010	-.236	-.016	.4000
.4500	-.090		-.184	.142	-.208	.050	-.235		-.258	-.006	-.272	-.010	.4500
.5000	-.117	.075	-.170	.114	-.215	.118	-.247	.019	-.267	.001	-.280	.001	.5000
.6000	-.156	.083	-.200	.092	-.251	.147	-.276	.062	-.306	.007	-.313	.008	.6000
.7000	-.167	.062	-.204	.084	-.255	.118	-.286	.067	-.316	.026	-.317	.022	.7000
.8000	-.148	.046	-.193	.065		.082	-.286	.076	-.314	.027	-.322	.019	.8000
.9000	-.148	.053	-.167	.055	-.224	.062	-.277	.060	-.306	.028			.9000

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN												X/C
	0.050		0.200		0.350		0.500		0.700		0.900		
	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha=04$												
.0125	.424	-.311			.372	-.062	.306	-.016		.248	.043		.0125
.0250	.341	-.295			.295	-.064	.266	-.016					.0250
.0500	.226	-.124	.230		.213	-.062	.179	-.015	.161	.043	.117	.017	.0500
.1000	.126	.081	.074		.065	-.060	.027	-.008	.082	.045	.066	.027	.1000
.1500	.031	.092	-.065	-.086	-.052	-.060	-.072	.001	-.032	.048	.024	.028	.1500
.2000	-.034	.084	-.118	.046	-.125	-.049	-.127	.001	-.131	.053	-.108	.030	.2000
.2500	-.030		-.135	.232	-.171	-.028	-.175	.009	-.175	.066	-.175	.050	.2500
.3000	-.024	.057	-.146	.193	-.199	.016	-.223	.032	-.208	.079	-.218	.074	.3000
.3500	-.089		-.153	.177	-.222	.060	-.248	.064	-.244	.090	-.243	.098	.3500
.4000	-.110		-.167	.186	-.210	.117	-.253	.105	-.251	.101	-.269	.139	.4000
.4500	-.120		-.214	.176	-.245	.188	-.282		-.288	.108	-.310	.145	.4500
.5000	-.144	.115	-.210	.149	-.234	.205	-.293	.128	-.297	.117	-.325	.160	.5000
.6000	-.180	.123	-.225	.133	-.281	.168	-.328	.149	-.336	.111	-.351	.144	.6000
.7000	-.191	.101	-.229	.123	-.296	.140	-.318	.135	-.346	.118	-.356	.143	.7000
.8000	-.171	.083	-.217	.102		.117	-.309	.124	-.344	.104	-.358	.128	.8000
.9000	-.173	.091	-.194	.093	-.255	.091	-.310	.097	-.344	.100			.9000
$\alpha=06$													
.0125	.385	-.279			.336	.002	.269	.062					.0125
.0250	.308	-.252			.260	.000	.226	.061	.197	.118			.0250
.0500	.192	.000	.194		.114	.170	.004	.141	.064	.114	.118	.067	.175
.1000	.092	.099	.039		.070	.032	.013	-.009	.069	.043	.127	.019	.192
.1500	-.008	.113	-.099	.043	-.088	.025	-.103	.085	-.064	.145	-.024	.198	.1500
.2000	-.064	.121	-.153	.210	-.157	.047	-.165	.105	-.166	.169	-.142	.201	.2000
.2500	.038		-.173	.243	-.208	.081	-.208	.119	-.220	.187	-.203	.225	.2500
.3000	-.062	.095	-.179	.212	-.236	.135	-.254	.145	-.247	.203	-.246	.238	.3000
.3500	-.125		-.177	.208	-.262	.182	-.278	.175	-.277	.209	-.266	.253	.3500
.4000	-.138		-.193	.217	-.262	.216	-.285	.204	-.285	.216	-.291	.259	.4000
.4500	-.147		-.243	.210	-.295	.248	-.315		-.318	.210	-.333	.243	.4500
.5000	-.169	.156	-.237	.182	-.254	.232	-.321	.201	-.326	.206	-.343	.235	.5000
.6000	-.202	.163	-.245	.172	-.301	.192	-.357	.201	-.363	.178	-.369	.190	.6000
.7000	-.213	.138	-.249	.160	-.322	.168	-.364	.170	-.373	.173	-.375	.187	.7000
.8000	-.191	.119	-.236	.140		.149	-.342	.153	-.372	.145	-.380	.164	.8000
.9000	-.193	.130	-.213	.131	-.280	.127	-.326	.122	-.372	.135			.9000
$\alpha=08$													
.0125	.341	-.242			.290	.084	.214	.146					.0125
.0250	.258	-.161			.212	.081	.173	.140	.128	.224			.0250
.0500	.154	.090	.149		.130	.090	.093	.153	.058	.249	-.027	.324	.0500
.1000	.055	.136	.000	.047	.103	.105	-.048	.175	-.004	.280	-.063	.385	.1000
.1500	-.043	.154	-.138	.181	-.127	.134	-.147	.221	-.113	.309	-.097	.352	.1500
.2000	-.094	.160	-.191	.280	-.191	.181	-.201	.254	-.193	.307	-.209	.309	.2000
.2500	-.064		-.215	.269	-.242	.226	-.244	.258	-.253	.315	-.262	.318	.2500
.3000	-.102	.167	-.224	.250	-.275	.268	-.289	.267	-.289	.311	-.295	.317	.3000
.3500	-.157		-.210	.250	-.301	.273	-.311	.273	-.322	.304	-.312	.325	.3500
.4000	-.169		-.222	.258	-.307	.267	-.321	.283	-.324	.299	-.328	.313	.4000
.4500	-.176		-.270	.252	-.340	.273	-.347		-.351	.282	-.368	.297	.4500
.5000	-.195	.207	-.273	.225	-.333	.256	-.357	.255	-.358	.277	-.384	.277	.5000
.6000	-.224	.210	-.277	.215	-.319	.228	-.390	.241	-.389	.226	-.406	.239	.6000
.7000	-.235	.181	-.276	.199	-.340	.203	-.398	.214	-.399	.220	-.410	.223	.7000
.8000	-.214	.162	-.262	.179		.189	-.397	.194	-.401	.182	-.412	.204	.8000
.9000	-.216	.171	-.240	.175	-.311	.162	-.368	.157	-.400	.177			.9000
$\alpha=10$													
.0125	.299	-.189			.242	.168	.149	.210					.0125
.0250	.216	-.087			.157	.170	.105	.236	.012	.379			.0250
.0500	.111	.153	.114		.070	.087	.194	.040	.297	-.028	.381	-.179	.0500
.1000	.018	.171	-.041	.175	-.060	.226	-.092	.335	-.069	.378	-.175	.427	.1000
.1500	-.070	.191	-.166	.287	-.159	.274	-.188	.358	-.148	.387	-.184	.396	.1500
.2000	-.116	.198	-.224	.317	-.225	.310	-.242	.355	-.234	.364	-.257	.364	.2000
.2500	-.092		-.253	.304	-.270	.311	-.275	.330	-.289	.372	-.303	.361	.2500
.3000	-.134	.210	-.271	.291	-.302	.327	-.318	.330	-.320	.361	-.330	.363	.3000
.3500	-.174		-.253	.294	-.331	.307	-.340	.325	-.350	.352	-.345	.375	.3500
.4000	-.191		-.245	.299	-.335	.297	-.348	.325	-.354	.349	-.366	.358	.4000
.4500	-.201		-.285	.293	-.369	.305	-.372		-.384	.322	-.393	.346	.4500
.5000	-.217	.251	-.292	.263	-.368	.296	-.380	.295	-.388	.319	-.404	.329	.5000
.6000	-.245	.251	-.302	.259	-.373	.267	-.410	.278	-.416	.268	-.421	.293	.6000
.7000	-.253	.222	-.295	.242	-.356	.238	-.417	.248	-.421	.255	-.426	.277	.7000
.8000	-.235	.203	-.282	.224		.230	-.417	.228	-.421	.222	-.426	.261	.8000
.9000	-.235	.215	-.259	.216	-.333	.201	-.293	.197	-.418	.219			.9000



TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha=12$													
.0125	.254	-.115			.176	.259	.045	.372	-.171	.467			.0125	
.0250	.174	.015		.152	.097	.295	.005	.384	-.160	.445			.0250	
.0500	.077	.193	.077	.201	.036	.356	-.033	.409	-.167	.428	-.321	.458	.0500	
.1000	-.019	.213	-.080	.297	-.096	.375	-.127	.409	-.216	.435	-.320	.467	.1000	
.1500	-.103	.233	-.203	.362	-.201	.372	-.220	.418	-.275	.418	-.326	.431	.1500	
.2000	-.145	.251	-.256	.356	-.255	.377	-.278	.415	-.318	.423	-.361	.406	.2000	
.2500	-.135		-.286	.350	-.297	.365	-.312	.384	-.350	.386	-.379	.405	.2500	
.3000	-.149	.255	-.310	.342	-.330	.372	-.350	.368	-.404	.419	-.417	.422	.3000	
.3500	-.207		-.303	.342	-.356	.345	-.368	.376	-.384	.400	-.443	.393	.3500	
.4000	-.219		-.270	.346	-.367	.340	-.375	.373	-.411	.377	-.447	.383	.4000	
.4500	-.223		-.305	.338	-.395	.348	-.398		-.416	.366	-.460	.342	.4500	
.5000	-.234	.297	-.312	.311	-.395	.342	-.408	.341	-.443	.308	-.464	.330	.5000	
.6000	-.262	.300	-.330	.300	-.415	.310	-.436	.326	-.450	.305	-.462	.339	.6000	
.7000	-.271	.271	-.317	.281	-.410	.283	-.441	.295	-.449	.271	-.462		.7000	
.8000	-.250	.251	-.302	.267		.273	-.441	.268					.8000	
.9000	-.253	.257	-.280	.256	-.312	.241	-.298	.240					.9000	
$\alpha=14$														
.0125	.205	-.026		.300	.075	.376	-.076	.472	-.304	.528			.0125	
.0250	.130	.096		.300	.015	.403	-.115	.470	-.292	.497	-.393	.527	.0250	
.0500	.043	.251	.038	.339	-.026	.433	-.129	.478	-.290	.471	-.392	.531	.0500	
.1000	-.048	.284	-.130	.375	-.135	.441	-.182	.455	-.313	.488	-.393	.495	.1000	
.1500	-.132	.288	-.232	.408	-.236	.430	-.257	.468	-.344	.468	-.413	.473	.1500	
.2000	-.175	.304	-.284	.402	-.293	.435	-.307	.466	-.374	.467	-.432	.482	.2000	
.2500	-.167		-.313	.395	-.331	.416	-.340	.439	-.394	.461	-.445	.486	.2500	
.3000	-.157	.304	-.340	.389	-.355	.421	-.374	.431	-.411	.454	-.452	.494	.3000	
.3500	-.241		-.349	.395	-.380	.395	-.394	.426	-.414	.442	-.458	.490	.3500	
.4000	-.239		-.307	.391	-.389	.389	-.406	.426	-.435	.426	-.471	.484	.4000	
.4500	-.240		-.323	.377	-.416	.405	-.422		-.439	.426	-.477	.484	.4500	
.5000	-.253	.347	-.329	.355	-.422	.391	-.427	.388	-.458	.379	-.467	.507	.5000	
.6000	-.281	.351	-.356	.356	-.433	.359	-.452	.368	-.466	.378	-.466	.538	.6000	
.7000	-.285	.315	-.337	.327	-.442	.328	-.457	.346	-.413	.350	-.462	.488	.7000	
.8000	-.265	.296	-.322	.312		.320	-.453	.344					.8000	
.9000	-.269	.301	-.299	.300	-.326	.313	-.320	.309					.9000	
$\alpha=16$														
.0125		.073		.388	-.050	.493	-.200	.545	-.389	.593			.0125	
.0250		.178		.402	-.090	.497	-.227	.543	-.380	.575	-.469	.674	.0250	
.0500	.008	.300	-.018	.466	-.110	.494	-.239	.535	-.374	.548	-.463	.668	.0500	
.1000	-.077	.322	-.168	.466	-.174	.482	-.268	.503	-.367	.567	-.465	.641	.1000	
.1500	-.159	.345	-.272	.453	-.268	.495	-.320	.522	-.348	.561	-.464	.632	.1500	
.2000	-.196	.359	-.313	.462	-.325	.499	-.351	.527	-.374	.541	-.464	.632	.2000	
.2500	-.190		-.341	.454	-.357	.470	-.374	.488	-.425	.550	-.470	.636	.2500	
.3000	-.166	.364	-.366	.446	-.381	.472	-.399	.479	-.439	.539	-.472	.637	.3000	
.3500	-.266		-.380	.455	-.407	.444	-.416	.479	-.452	.537	-.474	.640	.3500	
.4000	-.253		-.353	.444	-.414	.444	-.424	.497	-.455	.524	-.474	.625	.4000	
.4500	-.256		-.349	.430	-.434	.456	-.442		-.467	.507	-.478	.611	.4500	
.5000	-.268	.398	-.348	.412	-.436	.438	-.447	.479	-.468	.510	-.475	.583	.5000	
.6000	-.295	.404	-.372	.409	-.450	.424	-.468	.460	-.473	.490	-.468	.540	.6000	
.7000	-.300	.366	-.355	.381	-.462	.420	-.474	.441	-.419	.546	-.466	.531	.7000	
.8000	-.278	.353	-.338	.407		.418	-.393	.415	-.394	.546	-.461	.475	.8000	
.9000	-.283	.384	-.318	.394	-.355	.389	-.346	.401	-.385	.495			.9000	
$\alpha=18$														
.0125	.088	.147		.466	-.160	.579	-.308	.606	-.448	.657			.0125	
.0250	.038	.253		.472	-.193	.562	-.320	.601	-.438	.650	-.466	.682	.0250	
.0500	-.023	.354	-.077	.533	-.203	.553	-.317	.579	-.433	.638	-.462	.678	.0500	
.1000	-.106	.380	-.193	.497	-.235	.541	-.336	.567	-.436	.644	-.461	.652	.1000	
.1500	-.183	.400	-.304	.497	-.306	.553	-.368	.628	-.447	.638	-.459	.642	.1500	
.2000	-.220	.414	-.345	.511	-.351	.547	-.393	.624	-.460	.638	-.457	.645	.2000	
.2500	-.216		-.367	.516	-.377	.517	-.412	.587	-.460	.679	-.455	.647	.2500	
.3000	-.178	.432	-.389	.498	-.401	.521	-.430	.582	-.467	.686	-.451	.648	.3000	
.3500	-.274		-.405	.510	-.422	.499	-.441	.587	-.465	.692	-.451	.633	.3500	
.4000	-.270		-.395	.491	-.434	.527	-.448	.593	-.465	.689	-.450	.617	.4000	
.4500	-.270		-.389	.477	-.449	.557	-.459		-.466	.672	-.448	.592	.4500	
.5000	-.280	.446	-.376	.470	-.454	.545	-.465	.557	-.453	.660	-.444	.553	.5000	
.6000	-.305	.461	-.393	.492	-.464	.531	-.480	.548	-.432	.599	-.440	.538	.6000	
.7000	-.312	.428	-.378	.493	-.472	.503	-.483	.565	-.436	.534	-.441	.480	.7000	
.8000	-.286	.460	-.360	.504		.507	-.393	.590					.8000	
.9000	-.292	.485	-.341	.479	-.375	.492	-.363	.530					.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(a)  $M = 1.61$  - Concluded

X/C	FRACTION OF SEMISPAN												X/C		
	0.050		0.200		0.350		0.500		0.700		0.900				
	U	L	U	L	U	L	U	L	U	L	U	L			
	$\alpha = 20$														
.0125	.028	.231			-.263	.651	-.417	.723					.0125		
.0250	-.015	.331			-.275	.628	-.418	.715	-.454	.759			.0250		
.0500	-.056	.400			-.284	.618	-.403	.707	-.453	.760	-.398	.683	.0500		
.1000	-.138	.455	-.134		.561	.592	-.312	.613	-.404	.717	-.451	.763	-.400	.685	.1000
.1500	-.205	.474	-.320		.554	-.362	.634	-.421	.733	-.453	.758	-.401	.663	.1500	
.2000	-.241	.480	-.367		.582	-.391	.653	-.435	.702	-.449	.752	-.405	.653	.2000	
.2500	-.231		-.395		.580	-.408	.657	-.447	.680	-.445	.746	-.405	.655	.2500	
.3000	-.206	.505	-.412		.560	-.423	.668	-.459	.703	-.436	.736	-.401	.656	.3000	
.3500	-.274		-.424		.575	-.438	.646	-.463	.719	-.432	.727	-.401	.660	.3500	
.4000	-.294		-.421		.574	-.443	.657	-.466	.731	-.430	.714	-.400	.649	.4000	
.4500	-.284		-.421		.612	-.460	.661	-.476		-.430	.693	-.401	.632	.4500	
.5000	-.290	.516	-.403		.629	-.464	.647	-.477	.718	-.424	.681	-.400	.609	.5000	
.6000	-.314	.577	-.410		.621	-.469	.656	-.477	.683	-.414	.624	-.399	.578	.6000	
.7000	-.320	.573	-.401		.612	-.477	.669	-.459	.654	-.410	.614	-.398	.548	.7000	
.8000	-.291	.591	-.387		.624		.651	-.433	.604	-.409	.552	-.398	.487	.8000	
.9000	-.301	.592	-.362		.628	-.410	.575	-.403	.536	-.408	.500		.9000		

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b) M = 2.01

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = -20$													
.0125	.853	-.325	.683	-.305	.630	-.323	.574	-.324	.604	-.313	.583	-.267	.0125	
.0250	.812	-.327	.713	-.303	.676	-.323	.630	-.323	.605	-.298			.0250	
.0500	.782	-.331	.689	-.303	.662	-.324	.628	-.323	.605	-.313			.0500	
.1000	.700	-.287	.617	-.306	.626	-.324	.580	-.322	.576	-.313	.564	-.266	.1000	
.1500	.564	-.185	.538	-.314	.576	-.324	.538	-.320	.544	-.313	.535	-.266	.1500	
.2000	.462	-.283	.487	-.320	.499	-.323	.483	-.320	.480	-.310	.451	-.267	.2000	
.2500	.446		.450	-.319	.425	-.322	.417	-.317	.391	-.313	.413	-.264	.2500	
.3000	.437	-.248	.379	-.314	.377	-.324	.366	-.316	.357	-.309	.381	-.263	.3000	
.3500	.416		.341	-.309	.356	-.324	.347	-.316	.334	-.303	.364	-.263	.3500	
.4000	.359		.320	-.306	.335	-.325	.302	-.316	.320	-.295	.319	-.257	.4000	
.4500	.342		.278	-.300	.256	-.324	.285		.288	-.288	.269	-.257	.4500	
.5000	.288	-.185	.275	-.297	.269	-.324	.255	-.316	.263	-.282	.246	-.254	.5000	
.6000	.239	-.174	.207	-.296	.197	-.323	.186	-.314	.193	-.278	.219	-.253	.6000	
.7000	.221	-.175	.195	-.304	.179	-.320	.167	-.311	.183	-.292	.198	-.258	.7000	
.8000	.203	-.165	.188	-.305		-.315	.172	-.303	.186	-.295	.198	-.249	.8000	
.9000	.221	-.174	.202	-.284	.180	-.308	.180	-.252	.189	-.292			.9000	
$\alpha = -18$														
.0125	.818	-.326	.666	-.303	.621	-.321	.564	-.321	.594	-.310			.0125	
.0250	.781	-.328	.690	-.303	.660	-.321	.614	-.321					.0250	
.0500	.726	-.335	.658	-.308	.630	-.322	.604	-.319	.583	-.297	.554	-.265	.0500	
.1000	.643	-.300	.569	-.307	.578	-.321	.536	-.318	.544	-.311	.532	-.265	.1000	
.1500	.504	-.164	.481	-.313	.523	-.322	.493	-.318	.502	-.310	.496	-.265	.1500	
.2000	.415	-.224	.435	-.318	.447	-.321	.430	-.317	.437	-.308	.498	-.266	.2000	
.2500	.394		.394	-.318	.380	-.323	.365	-.312	.348	-.310	.395	-.266	.2500	
.3000	.381	-.247	.333	-.314	.325	-.323	.319	-.312	.313	-.308	.321	-.265	.3000	
.3500	.363		.290	-.310	.302	-.325	.307	-.312	.282	-.305	.313	-.262	.3500	
.4000	.310		.278	-.303	.291	-.323	.268	-.314	.266	-.300	.278	-.259	.4000	
.4500	.287		.223	-.300	.194	-.323	.211		.222	-.298	.225	-.257	.4500	
.5000	.244	-.188	.231	-.296	.221	-.320	.207	-.312	.223	-.294	.204	-.257	.5000	
.6000	.193	-.169	.164	-.298	.154	-.313	.146	-.310	.152	-.285	.171	-.250	.6000	
.7000	.177	-.167	.156	-.299	.134	-.306	.124	-.305	.138	-.283	.150	-.247	.7000	
.8000	.155	-.154	.150	-.279		-.298	.127	-.300	.138	-.279	.148	-.248	.8000	
.9000	.177	-.165	.160	-.218	.135	-.294	.136	-.242	.142	-.275			.9000	
$\alpha = -16$														
.0125	.775	-.328	.661	-.304	.610	-.314	.553	-.312	.584	-.300			.0125	
.0250	.741	-.329	.669	-.302	.642	-.314	.605	-.313					.0250	
.0500	.670	-.336	.628	-.306	.604	-.315	.586	-.312	.562	-.290	.538		.0500	
.1000	.587	-.309	.525	-.307	.532	-.317	.501	-.310	.517	-.300	.512		.1000	
.1500	.452	-.160	.427	-.314	.475	-.318	.453	-.310	.467	-.299	.477		.1500	
.2000	.371	-.164	.382	-.318	.400	-.318	.387	-.310	.398	-.298	.366		.2000	
.2500	.347		.343	-.315	.339	-.321	.321	-.304	.314	-.298	.322		.2500	
.3000	.333	-.202	.287	-.310	.282	-.322	.267	-.304	.278	-.298	.286		.3000	
.3500	.319		.243	-.304	.252	-.319	.264	-.305	.250	-.296	.274		.3500	
.4000	.268		.235	-.299	.258	-.316	.236	-.307	.239	-.295	.231		.4000	
.4500	.246		.180	-.292	.157	-.313	.169		.173	-.293			.4500	
.5000	.207	-.163	.188	-.287	.169	-.308	.151	-.304	.172	-.292			.5000	
.6000	.158	-.151	.124	-.280	.119	-.300	.109	-.300	.110	-.282			.6000	
.7000	.139	-.155	.117	-.266	.093	-.290	.087	-.295	.103	-.269			.7000	
.8000	.114	-.142	.113	-.211		-.280	.085	-.291	.105	-.258			.8000	
.9000	.139	-.153	.119	-.144	.100	-.276	.096	-.228	.105	-.255			.9000	
$\alpha = -14$														
.0125	.734	-.326	.641	-.302	.592	-.304	.537	-.303	.564	-.289			.0125	
.0250	.701	-.326	.642	-.301	.611	-.305	.577	-.303					.0250	
.0500	.609	-.333	.591	-.305	.568	-.307	.549	-.301	.531	-.280	.507	-.255	.0500	
.1000	.525	-.317	.475	-.309	.479	-.309	.450	-.299	.477	-.289	.479	-.256	.1000	
.1500	.393	-.178	.370	-.315	.417	-.308	.398	-.298	.417	-.289	.445	-.259	.1500	
.2000	.320	-.129	.325	-.309	.340	-.311	.336	-.298	.344	-.288	.325	-.261	.2000	
.2500	.293		.288	-.305	.286	-.318	.271	-.292	.260	-.288	.279	-.258	.2500	
.3000	.279	-.153	.233	-.299	.235	-.317	.210	-.297	.223	-.288	.242	-.258	.3000	
.3500	.267		.196	-.289	.194	-.309	.196	-.295	.200	-.287	.233	-.256	.3500	
.4000	.217		.187	-.283	.204	-.302	.190	-.293	.200	-.287	.200	-.252	.4000	
.4500	.195		.131	-.273	.114	-.297	.128		.134	-.286	.121	-.250	.4500	
.5000	.162	-.131	.140	-.266	.117	-.288	.108	-.286	.127	-.284	.113	-.247	.5000	
.6000	.115	-.133	.082	-.255	.073	-.279	.054	-.281	.066	-.279	.083	-.243	.6000	
.7000	.095	-.138	.079	-.184	.049	-.268	.045	-.273	.050	-.261	.086	-.235	.7000	
.8000	.069	-.125	.073	-.115		-.259	.041	-.268	.046	-.247	.060	-.227	.8000	
.9000	.095	-.141	.077	-.119	.056	-.258	.046	-.220	.055	-.237			.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b) M = 2.01 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = -12$													
.0125	.695	-.319	.616	-.298	.576	-.294	.522	-.290	.545	-.276			.0125	
.0250	.651	-.321	.612	-.295	.582	-.295	.535	-.291					.0250	
.0500	.549	-.321	.551	-.300	.529	-.295	.513	-.289	.498	-.271	.475	-.272	.0500	
.1000	.467	-.318	.426	-.304	.430	-.298	.401	-.288	.441	-.276	.440	-.272	.1000	
.1500	.344	-.177	.313	-.305	.362	-.298	.349	-.285	.371	-.276	.409	-.272	.1500	
.2000	.271	-.115	.271	-.295	.286	-.303	.290	-.285	.299	-.275	.274	-.269	.2000	
.2500	.245		.238	-.284	.237	-.307	.226	-.281	.214	-.275	.222	-.269	.2500	
.3000	.230	-.112	.184	-.274	.188	-.297	.163	-.286	.173	-.276	.186	-.268	.3000	
.3500	.222		.154	-.261	.151	-.289	.144	-.283	.142	-.275	.179	-.264	.3500	
.4000	.173		.144	-.252	.155	-.277	.146	-.276	.145	-.273	.162	-.265	.4000	
.4500	.151		.088	-.242	.077	-.270	.088		.092	-.272	.088	-.265	.4500	
.5000	.121	-.121	.098	-.231	.075	-.264	.069	-.265	.085	-.270	.074	-.264	.5000	
.6000	.079	-.110	.043	-.201	.031	-.250	.012	-.256	.028	-.265	.041	-.258	.6000	
.7000	.060	-.118	.041	-.095	.014	-.235	-.004	-.248	.011	-.251	.025	-.250	.7000	
.8000	.036	-.109	.037	-.090		-.227	.003	-.243	.006	-.228	.020	-.236	.8000	
.9000	.058	-.123	.041	-.103	.018	-.231	.008	-.220	.008	-.211			.9000	
$\alpha = -10$														
.0125	.661	-.308	.585	-.288	.559	-.281	.510	-.275					.0125	
.0250	.611	-.312	.562	-.284	.547	-.282	.516	-.276	.513	-.264			.0250	
.0500	.502	-.304	.516	-.288	.489	-.282	.464	-.274	.458	-.258	.452	-.279	.0500	
.1000	.417	-.301	.389	-.293	.381	-.284	.348	-.274	.400	-.264	.405	-.280	.1000	
.1500	.295	-.134	.269	-.294	.309	-.287	.295	-.272	.325	-.264	.376	-.279	.1500	
.2000	.217	-.099	.225	-.267	.239	-.293	.246	-.274	.263	-.263	.235	-.281	.2000	
.2500	.204		.192	-.248	.198	-.286	.185	-.273	.179	-.264	.178	-.281	.2500	
.3000	.187	-.046	.147	-.236	.148	-.275	.122	-.273	.132	-.264	.141	-.283	.3000	
.3500	.180		.112	-.226	.106	-.262	.099	-.263	.102	-.262	.129	-.281	.3500	
.4000	.133		.108	-.216	.112	-.250	.097	-.253	.098	-.258	.118	-.272	.4000	
.4500	.112		.056	-.201	.040	-.238	.050		.053	-.253	.048	-.267	.4500	
.5000	.087	-.101	.061	-.185	.044	-.229	.033	-.239	.046	-.248	.035	-.259	.5000	
.6000	.042	-.087	.013	-.136	-.006	-.213	-.021	-.224	-.006	-.239	.003	-.252	.6000	
.7000	.026	-.097	.006	-.056	-.017	-.200	-.036	-.215	-.017	-.229	-.013	-.239	.7000	
.8000	.002	-.086	.008	-.074		-.194	-.037	-.208	-.024	-.215	-.016	-.235	.8000	
.9000	.025	-.100	.013	-.088	-.015	-.197	-.027	-.200	-.025	-.184			.9000	
$\alpha = -08$														
.0125	.629	-.303	.564	-.280	.546	-.269	.496	-.259					.0125	
.0250	.570	-.301	.551	-.277	.526	-.270	.490	-.259	.496	-.247			.0250	
.0500	.459	-.300	.479	-.282	.462	-.270	.431	-.259	.431	-.244	.412	-.276	.0500	
.1000	.369	-.273	.343	-.286	.348	-.271	.309	-.258	.365	-.249	.374	-.277	.1000	
.1500	.244	-.085	.217	-.273	.270	-.276	.252	-.258	.283	-.248	.339	-.278	.1500	
.2000	.173	-.077	.178	-.237	.199	-.276	.199	-.262	.228	-.250	.204	-.280	.2000	
.2500	.164		.139	-.220	.150	-.266	.138	-.255	.145	-.250	.137	-.281	.2500	
.3000	.149	-.020	.105	-.201	.115	-.248	.089	-.250	.101	-.248	.103	-.281	.3000	
.3500	.142		.069	-.185	.075	-.230	.063	-.233	.067	-.240	.088	-.278	.3500	
.4000	.099		.073	-.169	.072	-.215	.055	-.219	.060	-.233	.078	-.259	.4000	
.4500	.077		.019	-.149	.012	-.203	.010		.015	-.225	.013	-.240	.4500	
.5000	.054	-.081	.024	-.130	.018	-.190	-.005	-.204	.010	-.217	.000	-.224	.5000	
.6000	.008	-.066	-.015	-.069	-.029	-.172	-.051	-.189	-.039	-.207	-.033	-.218	.6000	
.7000	-.005	-.079	-.024	-.033	-.044	-.161	-.064	-.178	-.050	-.197	-.047	-.208	.7000	
.8000	-.024	-.071	-.024	-.052	-.156	-.156	-.070	-.172	-.055	-.193	-.052	-.207	.8000	
.9000	-.004	-.080	-.017	-.065	-.047	-.156	-.062	-.170	-.055	-.177			.9000	
$\alpha = -06$														
.0125	.592	-.288	.543	-.266	.520	-.252	.475	-.243					.0125	
.0250	.530	-.285	.519	-.264	.487	-.253	.464	-.242	.463	-.233			.0250	
.0500	.415	-.285	.444	-.268	.420	-.253	.396	-.242	.401	-.229	.367	-.272	.0500	
.1000	.322	-.232	.297	-.272	.298	-.253	.267	-.241	.322	-.235	.335	-.271	.1000	
.1500	.202	-.042	.173	-.243	.211	-.263	.199	-.242	.232	-.236	.300	-.272	.1500	
.2000	.128	-.057	.124	-.208	.144	-.253	.154	-.246	.178	-.238	.170	-.272	.2000	
.2500	.124		.093	-.181	.097	-.233	.092	-.231	.104	-.238	.095	-.275	.2500	
.3000	.110	-.009	.064	-.163	.064	-.211	.043	-.218	.060	-.226	.062	-.275	.3000	
.3500	.106		.033	-.144	.030	-.192	.024	-.199	.029	-.210	.045	-.264	.3500	
.4000	.065		.037	-.130	.026	-.175	.018	-.182	.022	-.199	.033	-.215	.4000	
.4500	.037		-.012	-.100	-.030	-.158	-.023		-.022	-.187	-.024	-.188	.4500	
.5000	.022	-.059	-.011	-.078	-.027	-.147	-.038	-.161	-.027	-.177	-.037	-.181	.5000	
.6000	-.019	-.046	-.045	-.031	-.065	-.129	-.086	-.144	-.072	-.171	-.068	-.175	.6000	
.7000	-.034	-.055	-.057	-.025	-.084	-.121	-.096	-.137	-.082	-.160	-.080	-.172	.7000	
.8000	-.042	-.060	-.051	-.042	-.111	-.095	-.132	-.086	-.159	-.086	-.172	-.172	.8000	
.9000	-.031	-.057	-.048	-.054	-.080	-.112	-.095	-.130	-.088	-.155			.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b) M = 2.01 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = -0.4$													
.0125	.554	-.277	.515	-.251	.499	-.237	.453	-.224	.429	-.216	.335	-.261	.0125	
.0250	.487	-.267	.488	-.249	.457	-.240	.432	-.226	.367	-.216	.296	-.261	.0250	
.0500	.370	-.264	.405	-.252	.384	-.237	.363	-.225	.289	-.220	.296	-.261	.0500	
.1000	.277	-.187	.258	-.250	.261	-.239	.226	-.226	.189	-.222	.265	-.261	.1000	
.1500	.158	-.017	.129	-.208	.169	-.246	.154	-.229	.189	-.222	.265	-.261	.1500	
.2000	.092	-.037	.077	-.159	.097	-.220	.107	-.219	.134	-.223	.134	-.261	.2000	
.2500	.084		.051	-.127	.048	-.193	.049	-.194	.069	-.204	.064	-.260	.2500	
.3000	.074	.011	.020	-.113	.020	-.165	-.002	-.173	.026	-.180	.030	-.243	.3000	
.3500	.069		-.003	-.083	-.010	-.143	-.018	-.151	-.007	-.159	.013	-.202	.3500	
.4000	.028		-.003	-.063	-.015	-.125	-.025	-.131	-.016	-.146	-.009	-.153	.4000	
.4500	.009		-.038	-.037	-.067	-.106	-.063		-.055	-.137	-.057	-.135	.4500	
.5000	-.010	-.029	-.045	-.017	-.063	-.093	-.079	-.111	-.063	-.127	-.069	-.130	.5000	
.6000	-.052	-.022	-.075	-.000	-.097	-.079	-.120	-.092	-.104	-.124	-.102	-.129	.6000	
.7000	-.060	-.029	-.085	-.007	-.115	-.067	-.130	-.088	-.114	-.116	-.111	-.127	.7000	
.8000	-.063	-.039	-.079	-.024		-.060	-.132	-.083	-.116	-.115	-.117	-.131	.8000	
.9000	-.051	-.039	-.071	-.033	-.110	-.062	-.123	-.084	-.117	-.112			.9000	
$\alpha = -0.2$														
.0125	.441	-.214	.434	-.206	.428	-.208	.433	-.206	.348	-.204	.239	-.196	.0125	
.0250	.357	-.195	.392	-.205	.365	-.207	.340		.270	-.199	.239	-.196	.0250	
.0500	.252	-.179	.294	-.209	.283	-.208	.271	-.217	.203	-.205	.195	-.199	.0500	
.1000	.151	.008	.149	-.098	.155	-.208	.123	-.216	.085	-.175	.159	-.199	.1000	
.1500	.052	.044	.026	-.018	.051	-.085	.042	-.106	.009	-.075	.038	-.182	.1500	
.2000	-.002	.045	-.030	.024	-.009	-.019	-.011	-.030	.009	-.075	.038	-.182	.2000	
.2500	-.008		-.058	.057	-.061	.018	-.055	.015	-.033	-.030	-.023	-.094	.2500	
.3000	-.006	.079	-.084	.070	-.089	.049	-.097	.044	-.067	.011	-.066	-.037	.3000	
.3500	-.019		-.103	.081	-.115	.066	-.117	.071	-.097	.050	-.090	-.012	.3500	
.4000	-.055		-.099	.094	-.122	.075	-.127	.081	-.106	.076	-.110	.011	.4000	
.4500	-.075		-.121	.093	-.159	.084	-.154		-.139	.079	-.144	.029	.4500	
.5000	-.086	.059	-.120	.083	-.159	.091	-.166	.093	-.145	.091	-.156	.037	.5000	
.6000	-.116	.063	-.151	.077	-.186	.071	-.199	.085	-.179	.082	-.180	.042	.6000	
.7000	-.122	.056	-.151	.064	-.186	.059	-.208	.067	-.191	.074	-.185	.054	.7000	
.8000	-.116	.037	-.146	.050		.049	-.210	.060	-.192	.059	-.189	.049	.8000	
.9000	-.112	.039	-.135	.037	-.182	.033	-.205	.043	-.194	.052			.9000	
$\alpha = 0.0$														
.0125	.479	-.237	.465	-.216	.458	-.204	.407	-.195	.380	-.184	.272	-.222	.0125	
.0250	.402	-.223	.426	-.215	.397	-.204	.378	-.195	.305	-.183	.228	-.222	.0250	
.0500	.291	-.214	.336	-.217	.323	-.203	.302	-.195	.234	-.188	.194	-.223	.0500	
.1000	.194	-.064	.187	-.177	.190	-.211	.161	-.203	.116	-.186	.068	-.219	.1000	
.1500	.084	.020	.060	-.097	.084	-.164	.076	-.176	.054	-.128	.054	-.219	.1500	
.2000	.030	.013	.007	-.046	.023	-.111	.028	-.125	.054	-.128	.068	-.219	.2000	
.2500	.020		-.022	-.014	-.025	-.075	-.024	-.083	-.001	-.090	.002	-.160	.2500	
.3000	.018	.049	-.054	.007	-.058	-.042	-.069	-.058	-.037	-.066	-.048	-.095	.3000	
.3500	.007		-.072	.033	-.084	-.020	-.090	-.030	-.064	-.043	-.069	-.063	.3500	
.4000	-.030		-.069	.051	-.091	-.001	-.097	-.007	-.079	-.027	-.083	-.043	.4000	
.4500	-.049		-.093	.060	-.129	.019	-.128		-.112	-.021	-.117	-.031	.4500	
.5000	-.062	.028	-.097	.054	-.133	.031	-.140	.011	-.121	-.010	-.130	-.027	.5000	
.6000	-.095	.034	-.127	.049	-.156	.033	-.178	.028	-.156	-.013	-.157	-.028	.6000	
.7000	-.104	.024	-.131	.038	-.161	.030	-.187	.018	-.169	-.002	-.164	-.027	.7000	
.8000	-.099	.009	-.125	.022		.021	-.189	.018	-.171	-.005	-.170	-.035	.8000	
.9000	-.095	.010	-.115	.012	-.160	.006	-.183	.009	-.171	-.003			.9000	
$\alpha = 0.2$														
.0500	.250	-.183	.293	-.189	.283	-.172	.271		.803	-.157	.191	-.162	.0500	
.1000	.149	.019	.146	-.164	.154	-.172	.129	-.165	.803	-.155	.155	-.159	.1000	
.1500	.054	.047	.016	-.051	.046	-.143	.048	-.161	.803	-.155	.155	-.159	.1500	
.2000	.006	.043	-.032	.014	-.023	-.043	-.014	-.058	.803	-.103	.031	-.159	.2000	
.2500	-.008		-.064	.060	-.064	.007	-.054	.803	.803	-.038	-.029	-.144	.2500	
.3000	.001	.078	-.088	.072	-.098	.038	-.095	.803	.803	-.005	-.068	-.090	.3000	
.3500	-.020		-.103	.087	-.120	.065	-.115	.803	.803	-.034	-.093	-.020	.3500	
.4000	-.054		-.095	.096	-.132	.078	-.125	.803	.803	-.056	-.111	.024	.4000	
.4500	-.069		-.118	.090	-.157	.093	-.152		.803	.063	-.147	.044	.4500	
.5000	-.083	.058	-.122	.083	-.165	.100	-.162	.803	-.144	.072	-.156	.048	.5000	
.6000	-.115	.060	-.153	.063	-.186	.071	-.195	.803	-.176	.070	-.182	.041	.6000	
.7000	-.123	.054	-.156	.062	-.190	.065	-.206	.803	-.190	.067	-.188	.047	.7000	
.8000	-.115	.031	-.149	.047		.054	-.205	.803	-.190	.056	-.194	.041	.8000	
.9000	-.110	.035	-.138	.034	-.180	.037	-.201	.803	-.190	.056			.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b) M = 2.01 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = 0.4$													
.0125	.403	-.184	.403	-.194	.398	-.210	.349	-.211	.312	-.181			.0125	
.0250	.320	-.162	.362	-.192	.329	-.208	.312	-.211	.236	-.179	.203	-.152	.0250	
.0500	.214	-.124	.255	-.194	.253	-.212	.236	-.213	.236	-.179	.203	-.152	.0500	
.1000	.112	.050	.110	.010	.121	-.150	.093	-.185	.173	-.182	.157	-.155	.1000	
.1500	.022	.072	-.007	.062	.018	.015	.010	-.049	.065	-.084	.122	-.153	.1500	
.2000	-.030	.071	-.062	.099	-.042	.093	-.042	.087	-.022	.027	.008	-.086	.2000	
.2500	-.035		-.093	.128	-.089	.117	-.082	.132	-.068	.104	-.054	.034	.2500	
.3000	-.027	.110	-.120	.121	-.120	.137	-.123	.150	-.094	.148	-.089	.087	.3000	
.3500	-.051		-.137	.116	-.144	.137	-.144	.162	-.123	.174	-.115	.128	.3500	
.4000	-.087		-.131	.126	-.155	.138	-.155	.147	-.133	.175	-.138	.158	.4000	
.4500	-.098		-.145	.128	-.185	.134	-.179		-.163	.171	-.172	.165	.4500	
.5000	-.109	.091	-.144	.118	-.186	.140	-.189	.159	-.170	.172	-.179	.161	.5000	
.6000	-.137	.096	-.174	.109	-.211	.102	-.222	.133	-.201	.139	-.206	.143	.6000	
.7000	-.146	.088	-.173	.095	-.218	.093	-.231	.108	-.212	.125	-.212	.133	.7000	
.8000	-.135	.065	-.166	.083		.084	-.234	.099	-.214	.108	-.218	.113	.8000	
.9000	-.135	.067	-.155	.068	-.204	.065	-.230	.079	-.213	.099			.9000	
$\alpha = 0.6$														
.0125	.364	-.157	.375	-.164	.367	-.173	.315	-.166	.277	-.065			.0125	
.0250	.279	-.132	.334	-.163	.294	-.172	.274	-.169	.201	-.062	.168	.051	.0250	
.0500	.178	-.054	.225	-.160	.222	-.175	.200	-.171	.144	-.051	.124	.055	.0500	
.1000	.077	.074	.044	.079	.079	-.034	.071	-.077	.107	.033	.039	.087	.1000	
.1500	-.004	.101	-.037	.122	-.008	.102	-.018	.107	.033	.039	.087	.069	.1500	
.2000	-.048	.096	-.087	.150	-.069	.159	-.070	.171	-.050	.098	.091	.083	.2000	
.2500	-.053		-.116	.175	-.111	.173	-.109	.187	-.100	.144	-.076	.107	.2500	
.3000	-.043	.147	-.144	.165	-.140	.185	-.148	.201	-.131	.189	-.114	.123	.3000	
.3500	-.077		-.159	.158	-.164	.181	-.166	.211	-.151	.229	-.135	.141	.3500	
.4000	-.106		-.154	.169	-.174	.181	-.175	.200	-.157	.238	-.158	.175	.4000	
.4500	-.118		-.169	.167	-.201	.180	-.197		-.184	.238	-.187	.194	.4500	
.5000	-.129	.127	-.160	.154	-.204	.182	-.208	.199	-.190	.228	-.197	.200	.5000	
.6000	-.152	.130	-.189	.145	-.225	.140	-.238	.179	-.219	.228	-.223	.206	.6000	
.7000	-.157	.120	-.189	.131	-.235	.128	-.243	.148	-.229	.169	-.227	.197	.7000	
.8000	-.150	.097	-.180	.115		.117	-.247	.135	-.232	.148	-.232	.172	.8000	
.9000	-.151	.099	-.169	.094	-.220	.098	-.245	.114	-.232	.140			.9000	
$\alpha = 0.8$														
.0125	.326	-.112	.339	-.115	.333	-.087	.281	.003	.234	.114			.0125	
.0250	.240	-.089	.297	-.113	.258	-.086	.238	.005	.161	.118	.128	.207	.0250	
.0500	.148	.018	.199	-.083	.186	-.088	.168	.001	.107	.135	.085	.223	.0500	
.1000	.049	.103	.032	.118	.046	.057	.037	.058	.107	.135	.085	.223	.1000	
.1500	-.030	.134	-.070	.188	-.040	.160	-.056	.125	.009	.173	.050	.217	.1500	
.2000	-.073	.129	-.116	.192	-.095	.216	-.098	.188	-.068	.199	-.054	.216	.2000	
.2500	-.067		-.145	.216	-.137	.230	-.135	.225	-.123	.231	-.101	.238	.2500	
.3000	-.069	.194	-.172	.204	-.165	.240	-.172	.251	-.156	.252	-.135	.258	.3000	
.3500	-.104		-.184	.200	-.191	.231	-.188	.271	-.176	.271	-.154	.280	.3500	
.4000	-.131		-.183	.210	-.199	.231	-.199	.260	-.187	.275	-.177	.300	.4000	
.4500	-.138		-.192	.206	-.222	.230	-.220		-.206	.269	-.207	.304	.4500	
.5000	-.147	.167	-.181	.197	-.226	.229	-.226	.248	-.213	.267	-.216	.282	.5000	
.6000	-.170	.167	-.208	.186	-.245	.183	-.256	.227	-.237	.226	-.239	.245	.6000	
.7000	-.174	.156	-.211	.171	-.295	.170	-.261	.194	-.246	.216	-.243	.232	.7000	
.8000	-.165	.132	-.199	.155		.155	-.264	.175	-.248	.191	-.248	.207	.8000	
.9000	-.168	.134	-.186	.139	-.244	.136	-.262	.156	-.248	.182			.9000	
$\alpha = 1.0$														
.0125	.286	-.085	.297	-.056	.292	.011	.238	.114	.180	.237			.0125	
.0250	.203	-.039	.258	-.049	.216	.011	.194	.111	.144	.249	.063	.355	.0250	
.0500	.117	.055	.166	-.018	.149	.020	.129	.126	.114	.278	.031	.405	.0500	
.1000	.021	.148	.001	.202	.020	.141	.012	.184	.067	.278	.004	.385	.1000	
.1500	-.052	.170	-.096	.237	-.071	.239	-.074	.242	-.016	.314	-.089	.348	.1500	
.2000	-.092	.167	-.141	.239	-.124	.284	-.126	.286	-.092	.325	-.133	.350	.2000	
.2500	-.087		-.170	.267	-.160	.280	-.163	.298	-.142	.345	-.161	.354	.2500	
.3000	-.083	.239	-.192	.243	-.187	.288	-.191	.309	-.172	.342	-.180	.360	.3000	
.3500	-.127		-.205	.239	-.207	.281	-.208	.318	-.195	.344	-.195	.350	.3500	
.4000	-.147		-.207	.254	-.218	.274	-.218	.301	-.205	.333	-.221	.341	.4000	
.4500	-.155		-.216	.253	-.237	.280	-.236		-.223	.317	-.229	.322	.4500	
.5000	-.162	.209	-.203	.239	-.242	.273	-.242	.284	-.229	.309	-.251	.285	.5000	
.6000	-.181	.212	-.221	.225	-.260	.228	-.267	.267	-.251	.265	-.255	.272	.6000	
.7000	-.185	.203	-.226	.209	-.268	.213	-.272	.234	-.260	.257	-.255	.272	.7000	
.8000	-.177	.165	-.215	.192		.197	-.274	.216	-.262	.232	-.259	.251	.8000	
.9000	-.177	.172	-.200	.173	-.264	.176	-.272	.191	-.262	.219			.9000	

TABLE V.- Continued  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b) M = 2.01 - Continued

X/C	FRACTION OF SEMISPAN												X/C	
	0.050		0.200		0.350		0.500		0.700		0.900			
	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = 12$													
.0125	.235	-.039	.251	.016	.246	.127	.185	.226	.099	.387			.0125	
.0250	.158	.023	.208	.025	.167	.129	.140	.231					.0250	
.0500	.080	.110	.128	.059	.109	.158	.079	.283	.053	.436	-.060	.448	.0500	
.1000	-.015	.185	-.031	.278	-.007	.261	-.015	.327	.011	.429	-.067	.459	.1000	
.1500	-.079	.211	-.127	.290	-.097	.333	-.102	.366	-.049	.428	-.081	.442	.1500	
.2000	-.117	.213	-.167	.290	-.150	.347	-.152	.374	-.111	.401	-.142	.407	.2000	
.2500	-.120		-.196	.319	-.186	.326	-.184	.365	-.160	.405	-.175	.401	.2500	
.3000	-.104	.283	-.217	.286	-.209	.340	-.214	.372	-.189	.397	-.198	.404	.3000	
.3500	-.152		-.227	.291	-.233	.331	-.232	.370	-.210	.399	-.210	.415	.3500	
.4000	-.172		-.230	.304	-.240	.321	-.239	.351	-.218	.380	-.223	.400	.4000	
.4500	-.176		-.247	.302	-.258	.333	-.255		-.239	.363	-.246	.393	.4500	
.5000	-.182	.258	-.235	.283	-.260	.319	-.263	.334	-.244	.352	-.253	.369	.5000	
.6000	-.202	.263	-.238	.269	-.276	.276	-.283	.315	-.262	.309	-.268	.334	.6000	
.7000	-.205	.246	-.244	.253	-.285	.260	-.289	.281	-.271	.301	-.271	.324	.7000	
.8000	-.194	.206	-.234	.234	-.241	.241	-.289	.264	-.272	.277	-.275	.299	.8000	
.9000	-.198	.215	-.221	.213	-.256	.215	-.249	.238	-.272	.260			.9000	
$\alpha = 14$														
.0125	.197	.001	.213	.118	.200	.235	.127	.363	-.018	.497			.0125	
.0250	.124	.091	.175	.134	.122	.242	.078	.374					.0250	
.0500	.053	.161	.085	.224	.063	.315	.026	.437	-.033	.486	-.145	.502	.0500	
.1000	-.037	.231	-.050	.330	-.028	.375	-.048	.415	-.057	.472	-.151	.504	.1000	
.1500	-.098	.261	-.139	.341	-.119	.394	-.127	.434	-.103	.468	-.159	.483	.1500	
.2000	-.132	.263	-.185	.345	-.170	.394	-.166	.428	-.146	.451	-.194	.450	.2000	
.2500	-.135		-.207	.371	-.200	.374	-.199	.422	-.187	.458	-.217	.444	.2500	
.3000	-.117	.330	-.230	.334	-.222	.390	-.230	.423	-.210	.447	-.234	.454	.3000	
.3500	-.159		-.242	.342	-.242	.379	-.245	.412	-.229	.449	-.243	.459	.3500	
.4000	-.185		-.248	.355	-.251	.368	-.252	.396	-.235	.430	-.251	.450	.4000	
.4500	-.188		-.259	.355	-.266	.380	-.268		-.251	.406	-.265	.427	.4500	
.5000	-.192	.310	-.252	.339	-.269	.362	-.273	.373	-.255	.405	-.272	.414	.5000	
.6000	-.210	.311	-.251	.318	-.284	.318	-.292	.358	-.273	.353	-.283	.383	.6000	
.7000	-.213	.294	-.254	.300	-.290	.303	-.298	.322	-.280	.347	-.280	.370	.7000	
.8000	-.203	.255	-.244	.278	-.285	.285	-.298	.304	-.280	.321	-.279	.347	.8000	
.9000	-.204	.261	-.233	.259	-.243	.258	-.229	.276	-.279	.306			.9000	
$\alpha = 16$														
.0125	.158	.059	.167	.225	.143	.354	.048	.463	-.113	.562			.0125	
.0250	.088	.139	.130	.252	.069	.376	.002	.471					.0250	
.0500	.024	.184	.044	.346	.017	.437	-.037	.481	-.119	.536	-.207	.540	.0500	
.1000	-.062	.281	-.075	.380	-.061	.450	-.099	.470	-.130	.522	-.210	.551	.1000	
.1500	-.114	.301	-.169	.391	-.145	.450	-.163	.494	-.161	.522	-.216	.520	.1500	
.2000	-.154	.303	-.207	.402	-.193	.444	-.198	.482	-.191	.504	-.239	.490	.2000	
.2500	-.155		-.233	.418	-.219	.422	-.220	.478	-.220	.505	-.255	.495	.2500	
.3000	-.140	.371	-.253	.384	-.243	.440	-.244	.472	-.237	.493	-.269	.496	.3000	
.3500	-.165		-.259	.394	-.261	.424	-.261	.457	-.252	.496	-.276	.503	.3500	
.4000	-.208		-.264	.403	-.266	.417	-.269	.447	-.256	.472	-.280	.492	.4000	
.4500	-.197		-.276	.404	-.283	.427	-.281		-.269	.452	-.290	.482	.4500	
.5000	-.202	.356	-.274	.384	-.284	.406	-.288	.418	-.273	.452	-.293	.461	.5000	
.6000	-.220	.359	-.269	.364	-.295	.366	-.303	.407	-.286	.399	-.294	.428	.6000	
.7000	-.224	.341	-.267	.346	-.304	.349	-.307	.369	-.291	.393	-.291	.413	.7000	
.8000	-.211	.298	-.258	.324	-.332	.332	-.301	.349	-.290	.365	-.290	.391	.8000	
.9000	-.213	.304	-.249	.301	-.244	.304	-.232	.319	-.253	.346			.9000	
$\alpha = 18$														
.0125	.109	.129	.108	.339	.059	.472	-.038	.556	-.186	.623			.0125	
.0250	.042	.189	.082	.382	.002	.483	-.075	.555					.0250	
.0500	-.015	.244	.002	.404	.040	.500	-.106	.555	-.187	.590	-.250	.592	.0500	
.1000	-.090	.338	-.108	.440	-.104	.510	-.148	.539	-.192	.572	-.250	.607	.1000	
.1500	-.140	.355	-.191	.453	-.174	.511	-.194	.556	-.210	.576	-.254	.566	.1500	
.2000	-.173	.358	-.226	.472	-.211	.503	-.225	.551	-.229	.555	-.266	.538	.2000	
.2500	-.175		-.248	.468	-.236	.485	-.246	.534	-.250	.557	-.278	.545	.2500	
.3000	-.164	.420	-.264	.437	-.253	.502	-.264	.521	-.262	.548	-.286	.547	.3000	
.3500	-.177		-.273	.452	-.271	.482	-.273	.510	-.272	.544	-.286	.551	.3500	
.4000	-.221		-.278	.458	-.279	.481	-.281	.499	-.274	.526	-.287	.547	.4000	
.4500	-.212		-.290	.461	-.292	.480	-.290		-.285	.511	-.293	.529	.4500	
.5000	-.217	.409	-.290	.436	-.293	.455	-.294	.473	-.287	.505	-.294	.512	.5000	
.6000	-.233	.416	-.284	.415	-.304	.426	-.309	.458	-.291	.452	-.293	.481	.6000	
.7000	-.234	.396	-.280	.397	-.311	.397	-.313	.421	-.289	.447	-.291	.463	.7000	
.8000	-.222	.351	-.274	.374	-.382	.382	-.286	.403	-.281	.413	-.291	.448	.8000	
.9000	-.225	.358	-.266	.354	-.256	.352	-.246	.368	-.259	.398			.9000	

TABLE V.- Concluded  
PRESSURE COEFFICIENTS FOR CAMBERED WING

[From reference 1]

(b)  $M = 2.01$  - Concluded

X/C	FRACTION OF SEMISPAN												X/C
	0.050		0.200		0.350		0.500		0.700		0.900		
	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha = 20$												
.0125	.077	.189	.054	.432	-.008	.562	-.116	.617					.0125
.0250	.007	.239	.034	.456	-.057	.557	-.135	.613	-.229	.661			.0250
.0500	-.045	.303	-.031	.474	-.090	.562	-.156	.608	-.229	.627	-.305	.605	.0500
.1000	-.107	.377	-.129	.496	-.143	.559	-.189	.586	-.230	.606	-.303	.618	.1000
.1500	-.158	.410	-.207	.508	-.198	.557	-.224	.605	-.242	.623	-.305	.577	.1500
.2000	-.188	.414	-.241	.523	-.231	.553	-.251	.598	-.255	.594	-.310	.554	.2000
.2500	-.187		-.261	.514	-.253	.538	-.269	.575	-.272	.601	-.312	.560	.2500
.3000	-.177	.462	-.275	.486	-.268	.553	-.281	.563	-.279	.591	-.317	.560	.3000
.3500	-.192		-.281	.501	-.283	.532	-.287	.556	-.288	.592	-.316	.566	.3500
.4000	-.227		-.286	.507	-.288	.524	-.294	.547	-.286	.573	-.316	.561	.4000
.4500	-.222		-.295	.509	-.298	.526	-.304		-.287	.554	-.317	.545	.4500
.5000	-.226	.458	-.298	.485	-.300	.504	-.308	.519	-.286	.549	-.317	.529	.5000
.6000	-.242	.469	-.294	.462	-.311	.477	-.319	.501	-.278	.498	-.317	.499	.6000
.7000	-.244	.444	-.289	.445	-.317	.443	-.315	.465	-.275	.489	-.317	.482	.7000
.8000	-.230	.404	-.283	.418		.426	-.294	.444	-.275	.454	-.315	.475	.8000
.9000	-.230	.409	-.279	.401	-.275	.398	-.267	.407	-.267	.447			.9000



TABLE VI  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a)  $M = 1.61$

X/C		FRACTION OF SEMISPAN														X/C	
		0.050		0.200		0.350		0.500		0.700		0.825		0.950			
		U	L	U	L	U	L	U	L	U	L	U	L	U	L		
		α=-20															
.0125	.686	-.435	.698	-.484	.638	-.484	.647	-.440	.638	-.424	.612	-.320	.459	-.289	.0125		
.0250	.685	-.434	.688	-.484	.671	-.485	.672	-.440							.0250		
.0500	.619	-.423	.670	-.479	.667	-.485	.690	-.442							.0500		
.0750	.580	-.429	.651	-.470	.666	-.482	.696	-.440	.708	-.422	.685	-.320	.595	-.293	.0750		
.1000	.580	-.423	.629	-.465	.654	-.481	.694	-.441							.1000		
.1500	.549	-.316	.582	-.460		-.476	.681	-.441	.724	-.416	.724	-.318	.656	-.298	.1500		
.2000	.531	-.183	.551	-.469	.613	-.462	.647	-.441							.2000		
.2500	.565	-.214	.553	-.474	.605	-.457	.625	-.441	.675	-.407	.702	-.316	.666	-.299	.2500		
.3000	.499	-.200	.536	-.473	.583	-.452	.611	-.441							.3000		
.3500									.660	-.401	.657	-.318	.672	-.300	.3500		
.4000	.512	-.256	.500	-.460	.564	-.449	.576	-.440	.649	-.397	.633	-.321	.630	-.300	.4000		
.4500															.4500		
.5000	.453	-.250	.509	-.462	.529	-.448	.616	-.439	.621	-.392	.580	-.325	.578	-.299	.5000		
.5500															.5500		
.6000	.466	-.284	.503	-.458	.501	-.447	.646	-.437	.595	-.379	.552	-.326	.501	-.298	.6000		
.6500															.6500		
.7000	.452	-.307	.416	-.399	.533	-.444	.609	-.432	.555	-.369	.531	-.325	.412	-.296	.7000		
.7500															.7500		
.8000	.480	-.265	.493	-.338	.634	-.439	.579	-.414	.493	-.364	.485	-.321	.354	-.296	.8000		
.8500															.8500		
.9000	.517	-.287	.552	-.316	.570	-.386	.531	-.392							.9000		
α=-18																	
.0125	.628	-.408	.657	-.476	.621	-.486	.603	-.458	.582	-.423	.622	-.355	.485	-.307	.0125		
.0250	.612	-.414	.638	-.476	.635	-.488	.605	-.457							.0250		
.0500	.549	-.396	.588	-.472	.614	-.488	.598	-.459							.0500		
.0750	.513	-.415	.597	-.470	.604	-.488	.601	-.458	.583	-.418	.664	-.351	.565	-.306	.0750		
.1000	.504	-.383	.577	-.465	.599	-.486	.604	-.458							.1000		
.1500	.476	-.195	.529	-.452		-.488	.604	-.460	.585	-.412	.699	-.348	.650	-.307	.1500		
.2000	.465	-.178	.482	-.422	.517	-.481	.558	-.460							.2000		
.2500	.502	-.194	.482	-.394	.491	-.472	.540	-.455	.532	-.405	.679	-.343	.647	-.304	.2500		
.3000	.437	-.184	.471	-.376	.458	-.465	.525	-.454							.3000		
.3500									.550	-.393	.640	-.337	.651	-.301	.3500		
.4000	.444	-.238	.405	-.387	.443	-.464	.480	-.450	.606	-.385	.616	-.335	.617	-.299	.4000		
.4500															.4500		
.5000	.381	-.238	.381	-.391	.436	-.465	.450	-.450	.610	-.379	.566	-.332	.563	-.298	.5000		
.5500															.5500		
.6000	.340	-.265	.382	-.378	.424	-.458	.428	-.443	.583	-.373	.536	-.332	.485	-.296	.6000		
.6500															.6500		
.7000	.324	-.290	.334	-.345	.393	-.454	.451	-.426	.546	-.366	.511	-.328	.395	-.295	.7000		
.7500															.7500		
.8000	.369	-.254	.385	-.311	.420	-.446	.537	-.406	.487	-.358	.480	-.322	.334	-.298	.8000		
.8500															.8500		
.9000	.411	-.262	.408	-.297	.460	-.328	.540	-.382							.9000		
α=-16																	
.0125	.571	-.368	.615	-.476	.596	-.512	.582	-.493	.558	-.446	.553	-.366	.509	-.310	.0125		
.0250	.549	-.374	.595	-.466	.594	-.513	.581	-.492							.0250		
.0500	.485	-.358	.528	-.466	.552	-.509	.558	-.493							.0500		
.0750	.447	-.358	.523	-.472	.539	-.503	.546	-.493	.505	-.445	.535	-.362	.546	-.313	.0750		
.1000	.432	-.362	.510	-.475	.528	-.499	.544	-.491							.1000		
.1500	.419	-.184	.483	-.457		-.493	.527	-.491	.506	-.441	.548	-.355	.629	-.317	.1500		
.2000	.407	-.169	.423	-.443	.459	-.487	.471	-.492							.2000		
.2500	.438	-.184	.419	-.401	.432	-.479									.2500		
.3000	.381	-.172	.410	-.336	.396	-.476	.429	-.484	.419	-.426	.504	-.345	.628	-.328	.3000		
.3500															.3500		
.4000	.381	-.222	.347	-.299	.361	-.468	.371	-.471	.412	-.420	.520	-.347	.391	-.333	.4000		
.4500															.4500		
.5000	.323	-.222	.323	-.299	.318	-.447	.359	-.464	.394	-.416	.531	-.361	.540	-.239	.5000		
.5500															.5500		
.6000	.284	-.244	.284	-.296	.307	-.412	.335	-.456	.401	-.407	.526	-.367	.465	-.341	.6000		
.6500															.6500		
.7000	.257	-.275	.218	-.297	.304	-.406	.328	-.449	.467	-.402	.502	-.364	.375	-.342	.7000		
.7500															.7500		
.8000	.253	-.243	.270	-.307	.306	-.402	.345	-.432	.488	-.393	.477	-.359	.315	-.341	.8000		
.8500															.8500		
.9000	.298	-.245	.301	-.306	.336	-.391	.367	-.444							.9000		
α=-14																	
.0125	.523	-.329	.572	-.456	.570	-.503	.550	-.508							.0125		
.0250	.486	-.334	.546	-.457	.549	-.506	.543	-.508	.534	-.482	.519	-.465	.469	-.443	.0250		
.0500	.415	-.313	.476	-.439	.505	-.496	.519	-.506							.0500		
.0750	.388	-.315	.457	-.432	.487	-.489	.497	-.503	.445	-.488	.477	-.469	.477	-.441	.0750		
.1000	.397	-.211	.441	-.432	.472	-.482	.492	-.500							.1000		
.1500	.365	-.155	.429	-.420		-.473	.476	-.495	.428	-.485	.479	-.466	.486	-.437	.1500		
.2000	.350	-.153	.372	-.403	.415	-.466	.419	-.491							.2000		
.2500	.378	-.164	.362	-.369	.386	-.455	.394	-.483	.368	-.486	.436	-.465	.461	-.431	.2500		
.3000	.330	-.160	.346	-.312	.348	-.457	.376	-.482							.3000		
.3500									.339	-.475	.392	-.466	.474	-.430	.3500		
.4000	.323	-.205	.297	-.248	.311	-.450	.311	-.470	.323	-.474	.373	-.465	.471	-.428	.4000		
.4500															.4500		
.5000	.273	-.205	.273	-.255	.275	-.400	.274	-.471	.306	-.471	.346	-.465	.469	-.426	.5000		
.5500															.5500		
.6000	.233	-.227	.234	-.256	.244	-.382	.241	-.466	.299	-.463	.328	-.454	.447	-.424	.6000		
.6500															.6500		
.7000	.206	-.256	.171	-.256	.222	-.374	.228	-.463	.289	-.403	.329	-.440	.373	-.420	.7000		
.7500															.7500		
.8000	.197	-.230	.189	-.268	.212	-.368	.244	-.403	.296	-.373	.396	-.426	.311	-.420	.8000		
.8500															.8500		
.9000	.219	-.228	.207	-.268	.249	-.352	.262	-.350							.9000		

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha = -12$														
.0125	.477	-.273	.520	-.413	.532	-.466	.522	-.513	.509	-.498	.488	-.504	.495	-.504	.0125
.0250	.429	-.260	.498	-.426	.534	-.472	.508	-.510							.0250
.0500	.362	-.248	.423	-.413	.457	-.465	.494	-.498							.0500
.0750	.332	-.241	.403	-.397	.434	-.454	.456	-.489	.408	-.506	.436	-.505	.437	-.503	.0750
.1000	.346	-.194	.384	-.390	.417	-.444	.449	-.480							.1000
.1500	.319	-.134	.370	-.376		-.435	.427	-.469	.388	-.495	.428	-.500	.424	-.504	.1500
.2000	.299	-.127	.326	-.346	.360	-.428	.372	-.460							.2000
.2500	.321	-.143	.311	-.248	.337	-.421	.346	-.452	.325	-.487	.380	-.495	.396	-.498	.2500
.3000	.285	-.138	.293	-.240	.298	-.423	.330	-.450							.3000
.3500									.285	-.479	.325	-.493	.398	-.498	.3500
.4000	.270	-.184	.252	-.229	.262	-.411	.267	-.442							.4000
.4500									.265	-.474	.300	-.489	.368	-.498	.4500
.5000	.224	-.181	.225	-.240	.228	-.351	.228	-.446							.5000
.5500									.238	-.474	.268	-.489	.336	-.496	.5500
.6000	.188	-.203	.193	-.241	.199	-.328	.188	-.444							.6000
.6500									.219	-.472	.244	-.487	.278	-.493	.6500
.7000	.165	-.238	.126	-.239	.179	-.299	.177	-.445							.7000
.7500									.206	-.464	.246	-.440	.225	-.481	.7500
.8000	.151	-.210	.141	-.251	.167	-.286	.176	-.391							.8000
.8500									.212	-.358	.257	-.389	.206	-.425	.8500
.9000	.179	-.205	.156	-.254	.167	-.278	.175	-.319							.9000
$\alpha = -10$															
.0125	.421	-.184	.466	-.362	.493	-.425	.483	-.491							.0125
.0250	.376	-.187	.442	-.383	.481	-.437	.458	-.484	.480	-.490	.461	-.514	.444	-.517	.0250
.0500	.305	-.201	.364	-.374	.405	-.429	.453	-.468							.0500
.0750	.276	-.210	.339	-.356	.379	-.417	.406	-.457	.377	-.495	.400	-.507	.416	-.515	.0750
.1000	.283	-.173	.323	-.346	.359	-.406	.392	-.449							.1000
.1500	.269	-.102	.301	-.323		-.397	.370	-.437	.344	-.476	.379	-.495	.379	-.512	.1500
.2000	.246	-.100	.274	-.277	.302	-.386	.320	-.427							.2000
.2500	.263	-.117	.258	-.210	.282	-.376	.292	-.420	.272	-.463	.332	-.484	.351	-.504	.2500
.3000	.234	-.126	.239	-.186	.246	-.377	.277	-.422							.3000
.3500									.245	-.455	.283	-.476	.348	-.503	.3500
.4000	.217	-.157	.202	-.209	.210	-.327	.214	-.410							.4000
.4500									.215	-.451	.253	-.474	.315	-.499	.4500
.5000	.179	-.161	.173	-.218	.178	-.289	.179	-.418							.5000
.5500									.189	-.452	.215	-.474	.276	-.493	.5500
.6000	.140	-.184	.145	-.222	.149	-.285	.146	-.412							.6000
.6500									.170	-.452	.192	-.475	.213	-.488	.6500
.7000	.118	-.221	.081	-.221	.129	-.271	.125	-.369							.7000
.7500									.157	-.452	.183	-.469	.156	-.485	.7500
.8000	.103	-.192	.097	-.234	.120	-.264	.129	-.346							.8000
.8500									.152	-.343	.179	-.401	.130	-.471	.8500
.9000	.131	-.187	.107	-.242	.117	-.256	.126	-.296							.9000
$\alpha = -08$															
.0125	.361	-.096	.416	-.301	.448	-.374	.449	-.439							.0125
.0250	.324	-.149	.374	-.327	.424	-.389	.407	-.440	.456	-.445	.430	-.507	.432	-.518	.0250
.0500	.255	-.159	.315	-.315	.349	-.383	.403	-.423							.0500
.0750	.226	-.166	.282	-.297	.319	-.369	.355	-.412	.362	-.456	.366	-.480	.387	-.508	.0750
.1000	.226	-.117	.259	-.282	.310	-.358	.340	-.403							.1000
.1500	.226	-.075	.247	-.267		-.349	.318	-.390	.320	-.434	.335	-.457	.342	-.493	.1500
.2000	.198	-.074	.223	-.199	.253	-.336	.270	-.380							.2000
.2500	.211	-.080	.231	-.123	.231	-.316	.247	-.376	.292	-.421	.287	-.443	.306	-.482	.2500
.3000	.188	-.117	.190	-.136	.198	-.293	.228	-.377							.3000
.3500									.200	-.416	.236	-.441	.300	-.474	.3500
.4000	.170	-.133	.155	-.192	.164	-.232	.168	-.366							.4000
.4500									.168	-.414	.206	-.437	.268	-.469	.4500
.5000	.135	-.141	.128	-.190	.133	-.242	.134	-.375							.5000
.5500									.143	-.416	.173	-.439	.231	-.463	.5500
.6000	.100	-.163	.107	-.199	.109	-.259	.104	-.321							.6000
.6500									.123	-.417	.145	-.443	.166	-.461	.6500
.7000	.079	-.204	.042	-.198	.081	-.242	.081	-.296							.7000
.7500									.113	-.421	.138	-.446	.112	-.461	.7500
.8000	.061	-.173	.059	-.214	.073	-.233	.083	-.285							.8000
.8500									.104	-.413	.131	-.443	.082	-.462	.8500
.9000	.089	-.168	.065	-.220	.075	-.228	.081	-.268							.9000
$\alpha = -06$															
.0125	.296	-.034	.358	-.228	.398	-.312	.399	-.382							.0125
.0250	.265	-.115	.319	-.257	.370	-.332	.357	-.389	.420	-.392	.398	-.481	.421	-.511	.0250
.0500	.207	-.116	.262	-.251	.293	-.325	.344	-.374							.0500
.0750	.175	-.130	.228	-.231	.252	-.313	.295	-.362	.327	-.415	.327	-.444	.367	-.486	.0750
.1000	.185	-.083	.208	-.207	.244	-.302	.279	-.352							.1000
.1500	.177	-.046	.192	-.142		-.293	.262	-.340	.276	-.396	.286	-.420	.310	-.462	.1500
.2000	.153	-.050	.166	-.111	.204	-.268	.220	-.329							.2000
.2500	.163	-.051	.170	-.107	.180	-.225	.195	-.323	.194	-.379	.242	-.407	.251	-.450	.2500
.3000	.143	-.098	.142	-.108	.151	-.199	.179	-.325							.3000
.3500									.158	-.375	.191	-.403	.242	-.443	.3500
.4000	.128	-.107	.108	-.166	.113	-.180	.122	-.314							.4000
.4500									.120	-.373	.159	-.405	.219	-.440	.4500
.5000	.094	-.140	.084	-.164	.089	-.225	.090	-.260							.5000
.5500									.097	-.377	.121	-.411	.186	-.434	.5500
.6000	.061	-.139	.067	-.176	.063	-.227	.058	-.235							.6000
.6500									.077	-.379	.096	-.415	.120	-.432	.6500
.7000	.041	-.181	.002	-.178	.041	-.214	.043	-.248							.7000
.7500									.067	-.382	.090	-.419	.063	-.434	.7500
.8000	.023	-.150	.023	-.193	.035	-.209	.037	-.250							.8000
.8500									.064	-.378	.081	-.424	.036	-.429	.8500
.9000	.051	-.146	.024	-.201	.037	-.206	.039	-.239							.9000

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
α = -0.4															
.0125	.226	.017	.289	-.122	.333	-.211	.339	-.300	.368	-.315	.340	-.419	.396	-.463	.0125
.0250	.196	-.079	.248	-.183	.302	-.240	.288	-.313							.0250
.0500	.153	-.067	.201	-.177	.238	-.251	.273	-.299							.0500
.0750	.121	-.054	.152	-.145	.186	-.244	.254	-.286	.281	-.356	.291	-.383	.329	-.432	.0750
.1000	.130	-.062	.151	-.120	.172	-.232	.210	-.278							.1000
.1500	.129	-.022	.133	-.054		-.202	.202	-.272	.221	-.336	.223	-.361	.269	-.411	.1500
.2000	.107	-.023	.108	-.076	.155	-.152	.163	-.266							.2000
.2500	.116	-.022	.121	-.075	.127	-.114	.138	-.258	.137	-.325	.190	-.349	.193	-.399	.2500
.3000	.097	-.074	.093	-.081	.099	-.143	.125	-.235							.3000
.3500									.111	-.321	.142	-.354	.180	-.397	.3500
.4000	.080	-.080	.061	-.136	.061	-.146	.068	-.178							.4000
.4500									.057	-.319	.108	-.353	.153	-.394	.4500
.5000	.050	-.106	.042	-.137	.041	-.203	.038	-.194							.5000
.5500									.044	-.323	.067	-.365	.127	-.391	.5500
.6000	.021	-.116	.025	-.149	.017	-.197	.009	-.203							.6000
.6500									.029	-.317	.050	-.368	.071	-.389	.6500
.7000	-.001	-.159	-.036	-.154	-.003	-.185	-.005	-.223							.7000
.7500									.015	-.279	.037	-.376	.014	-.391	.7500
.8000	-.018	-.132	-.017	-.171	-.008	-.182	-.007	-.212							.8000
.8500									.005	-.257	.027	-.384	-.015	-.386	.8500
.9000	.010	-.124	-.019	-.173	-.006	-.179	-.008	-.208							.9000
α = -0.2															
.0125	.145	.071	.214	-.033	.267	-.105	.274	-.171							.0125
.0250	.129	-.030	.183	-.099	.228	-.170	.225	-.199	.305	-.223	.285	-.342	.344	-.295	.0250
.0500	.112	-.007	.145	-.065	.183	-.141	.199	-.207							.0500
.0750	.083	-.004	.088	-.046	.133	-.124	.166	-.206	.227	-.280	.245	-.305	.284	-.364	.0750
.1000	.094	-.022	.096	-.041	.110	-.105	.135	-.193							.1000
.1500	.087	.013	.086	-.023		-.070	.148	-.167	.160	-.260	.167	-.286	.127	-.346	.1500
.2000	.073	.013	.064	-.036	.103	-.065	.118	-.133							.2000
.2500	.079	.010	.074	-.035	.090	-.074	.090	-.091	.092	-.241	.148	-.279	.147	-.337	.2500
.3000	.062	-.030	.056	-.049	.053	-.092	.078	-.135							.3000
.3500									.075	-.220	.107	-.286	.143	-.339	.3500
.4000	.045	-.044	.021	-.093	.018	-.119	.029	-.127							.4000
.4500									.022	-.194	.072	-.286	.109	-.339	.4500
.5000	.015	-.073	.003	-.099	.001	-.158	-.011	-.161							.5000
.5500									-.008	-.195	.017	-.307	.075	-.340	.5500
.6000	-.009	-.083	-.009	-.117	-.023	-.156	-.028	-.171							.6000
.6500									-.017	-.195	.004	-.303	.019	-.333	.6500
.7000	-.027	-.130	-.070	-.121	-.035	-.148	-.044	-.179							.7000
.7500									-.022	-.215	-.008	-.282	-.034	-.336	.7500
.8000	-.045	-.103	-.048	-.141	-.045	-.150	-.042	-.174							.8000
.8500									-.032	-.226	-.015	-.272	-.059	-.329	.8500
.9000	-.018	-.093	-.054	-.142	-.042	-.148	-.046	-.171							.9000
α = 0.0															
.0125	.086	.121	.125	.073	.175	-.008	.193	-.031							.0125
.0250	.047	.026	.105	.018	.139	-.051	.167	-.077	.227	-.086	.217	-.239	.312	-.309	.0250
.0500	.065	.042	.090	.030	.111	-.009	.132	-.077							.0500
.0750	.045	.040	.038	.016	.073	-.011	.096	-.051	.153	-.167	.179	-.203	.223	-.281	.0750
.1000	.054	.004	.040	.021	.044	-.021	.057	-.049							.1000
.1500	.050	.050	.038	.021		-.026	.076	-.087	.093	-.124	.100	-.193	.172	-.269	.1500
.2000	.042	.045	.021	.004	.042	-.023	.059	-.061							.2000
.2500	.044	.024	.026	-.002	.042	-.034	.038	-.053	.055	-.124	.076	-.178	.085	-.260	.2500
.3000	.014	.010	.014	-.015	.012	-.057	.026	-.085							.3000
.3500									.026	-.112	.061	-.177	.088	-.262	.3500
.4000	.010	-.008	-.017	-.055	-.025	-.082	-.019	-.093							.4000
.4500									-.018	-.142	.030	-.180	.062	-.264	.4500
.5000	-.017	-.041	-.031	-.067	-.045	-.119	-.063	-.124							.5000
.5500									-.054	-.158	-.028	-.188	.025	-.267	.5500
.6000	-.043	-.056	-.043	-.088	-.062	-.126	-.074	-.124							.6000
.6500									-.068	-.167	-.045	-.199	-.027	-.252	.6500
.7000	-.061	-.105	-.099	-.093	-.077	-.116	-.085	-.143							.7000
.7500									-.077	-.192	-.062	-.224	-.075	-.244	.7500
.8000	-.071	-.088	-.079	-.116	-.081	-.120	-.085	-.143							.8000
.8500									-.075	-.188	-.067	-.243	-.101	-.239	.8500
.9000	-.048	-.067	-.081	-.118	-.081	-.120	-.084	-.139							.9000
α = 0.2															
.0125	.037	.170	.028	.161	.081	.110	.078	.093							.0125
.0250	-.012	.090	.020	.103	.045	.069	.050	.038	.129	.040	.128	-.094	.244	-.179	.0250
.0500	.018	.081	.030	.085	.040	.070	.066	.030							.0500
.0750	.013	.048	-.001	.075	.016	.045	.048	.024	.072	-.027	.091	-.067	.155	-.142	.0750
.1000	.038	.019	.089	.080	-.015	.096	.092	.001							.1000
.1500	.018	.077	-.006	.064		.024	.010	-.007	-.006	-.043	.029	-.055	.107	-.132	.1500
.2000	.012	.072	-.017	.044	-.009	.022	-.002	.009							.2000
.2500	.032	.050	-.004	.036	-.003	.010	-.009	-.005	.005	-.028	.018	-.082	.016	-.124	.2500
.3000	-.028	.054	-.016	.022	-.019	-.017	-.019	-.041							.3000
.3500									-.020	-.070	.006	-.092	.022	-.167	.3500
.4000	-.023	.021	-.056	-.022	-.063	-.040	-.051	-.047							.4000
.4500									-.052	-.090	-.010	-.108	.011	-.178	.4500
.5000	-.048	-.017	-.067	-.035	-.083	-.085	-.106	-.089							.5000
.5500									-.090	-.119	-.062	-.155	-.025	-.183	.5500
.6000	-.069	-.028	-.075	-.056	-.096	-.094	-.116	-.087							.6000
.6500									-.109	-.128	-.081	-.163	-.067	-.193	.6500
.7000	-.087	-.082	-.124	-.065	-.107	-.086	-.125	-.108							.7000
.7500									-.121	-.153	-.100	-.190	-.109	-.200	.7500
.8000	-.087	-.079	-.107	-.088	-.109	-.094	-.119	-.105							.8000
.8500									-.132	-.154	-.111	-.210	-.130	-.199	.8500
.9000	-.063	-.048	-.106	-.096	-.107	-.092	-.116	-.106							.9000

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN														X/C	
	0.050		0.200		0.350		0.500		0.700		0.825		0.950			
	U	L	U	L	U	L	U	L	U	L	U	L	U	L		
	$\alpha = 0.4$															
.0125	-.021	.243	-.087	.255	-.042	.220	-.092	.209	-.013	.180	.009	.097	.135	.056	.0125	
.0250	-.065	.155	-.098	.183	-.102	.163	-.103	.155	-.013	.180	.009	.097	.135	.056	.0250	
.0500	-.041	.133	-.062	.153	-.081	.147	-.063	.109	-.012	.104	.006	.072	.037	.010	.0500	
.0750	-.023	.083	-.061	.145	-.056	.111	-.037	.111	-.076	.053	-.046	.037	.019	-.024	.0750	
.1000	-.016	.076	-.072	.136	-.062	.108	-.041	.085	-.082	.016	-.089	-.003	-.044	-.054	.1000	
.1500	-.013	.111	-.054	.115	-.072	.086	-.085	.080	-.102	-.004	-.097	-.038	-.089	-.104	.1500	
.2000	-.020	.113	-.062	.092	-.072	.086	-.095	.076	-.110	-.039	-.096	-.042	-.094	-.108	.2000	
.2500	.013	.087	-.044	.089	-.057	.063	-.078	.039	-.136	-.072	-.129	-.099	-.112	-.127	.2500	
.3000	-.067	.092	-.045	.063	-.055	.028	-.084	.021	-.132	-.082	-.135	-.114	-.136	-.152	.3000	
.3500									-.150	-.108	-.148	-.145	-.152	-.159	.3500	
.4000	-.057	.059	-.098	.019	-.092	.005	-.088	.004	-.162	-.108	-.146	-.163	-.154	-.165	.4000	
.4500															.4500	
.5000	-.080	.020	-.102	.004	-.132	-.043	-.134	-.039							.5000	
.5500															.5500	
.6000	-.097	.012	-.108	-.019	-.136	-.054	-.161	-.043							.6000	
.6500															.6500	
.7000	-.111	-.053	-.152	-.032	-.142	-.047	-.175	-.067							.7000	
.7500															.7500	
.8000	-.106	-.050	-.134	-.058	-.141	-.058	-.159	-.069							.8000	
.8500															.8500	
.9000	-.086	-.019	-.128	-.064	-.132	-.057	-.153	-.061							.9000	
$\alpha = 0.6$																
.0125	-.077	.302	-.170	.331	-.139	.303	-.211	.290	-.159	.268	-.129	.221	-.069	.173	.0125	
.0250	-.114	.241	-.188	.254	-.188	.240	-.217	.242	-.159	.268	-.129	.221	-.069	.173	.0250	
.0500	-.097	.177	-.173	.208	-.199	.208	-.188	.201	-.158	.176	-.135	.155	-.088	.109	.0500	
.0750	-.069	.151	-.158	.197	-.184	.178	-.167	.177	-.167	.121	-.133	.100	-.108	.047	.0750	
.1000	-.052	.112	-.138	.191	-.178	.169	-.161	.146	-.167	.070	-.158	.066	-.126	-.000	.1000	
.1500	-.040	.155	-.106	.166	-.154	.129	-.178	.128	-.180	.045	-.169	.032	-.147	-.023	.1500	
.2000	-.052	.153	-.098	.139	-.154	.129	-.178	.128	-.180	.045	-.169	.032	-.147	-.023	.2000	
.2500	-.009	.130	-.086	.129	-.121	.106	-.183	.088	-.180	.045	-.169	.032	-.147	-.023	.2500	
.3000	-.099	.133	-.074	.102	-.112	.074	-.177	.071	-.190	.002	-.176	.011	-.161	-.050	.3000	
.3500									-.217	-.020	-.197	.055	-.176	-.076	.3500	
.4000	-.086	.098	-.139	.057	-.120	.047	-.149	.044	-.217	-.037	-.210	-.067	-.190	-.116	.4000	
.4500									-.207	-.069	-.218	-.091	-.198	-.143	.4500	
.5000	-.108	.064	-.136	.037	-.166	-.004	-.161	.005	-.188	-.072	-.225	-.127	-.202	-.157	.5000	
.5500															.5500	
.6000	-.120	.049	-.140	.015	-.179	-.014	-.185	-.006							.6000	
.6500															.6500	
.7000	-.136	-.022	-.178	.000	-.180	-.009	-.206	-.025							.7000	
.7500															.7500	
.8000	-.123	-.020	-.162	-.031	-.177	-.023	-.210	-.030							.8000	
.8500															.8500	
.9000	-.113	.016	-.154	-.036	-.169	-.022	-.192	-.034							.9000	
$\alpha = 0.8$																
.0125	-.142	.351	-.236	.385	-.224	.373	-.290	.348	-.248	.339	-.221	.295	-.152	.248	.0125	
.0250	-.163	.244	-.252	.317	-.263	.299	-.300	.311	-.248	.339	-.221	.295	-.152	.248	.0250	
.0500	-.147	.220	-.245	.274	-.268	.267	-.275	.285	-.240	.237	-.215	.227	-.174	.178	.0500	
.0750	-.125	.208	-.237	.242	-.254	.236	-.254	.223	-.237	.237	-.215	.227	-.174	.178	.0750	
.1000	-.098	.149	-.228	.228	-.248	.226	-.244	.203	-.237	.175	-.216	.157	-.190	.101	.1000	
.1500	-.071	.200	-.193	.208	-.206	.238	.188	.188	-.238	.121	-.226	.135	-.202	.061	.1500	
.2000	-.082	.195	-.127	.189	-.248	.169	-.239	.177	-.238	.091	-.231	.087	-.215	.035	.2000	
.2500	-.035	.167	-.122	.166	-.234	.250	-.245	.131	-.252	.046	-.236	.052	-.225	-.009	.2500	
.3000	-.123	.170	-.100	.143	-.165	.118	-.250	.121	-.269	.023	-.259	-.008	-.235	-.026	.3000	
.3500									-.276	.000	-.265	-.029	-.235	-.079	.3500	
.4000	-.115	.133	-.162	.094	-.155	.087	-.258	.086	-.280	-.032	-.274	-.059	-.243	-.114	.4000	
.4500									-.287	-.025	-.278	-.086	-.240	-.135	.4500	
.5000	-.134	.090	-.167	.073	-.185	.028	-.264	.044							.5000	
.5500															.5500	
.6000	-.146	.084	-.166	.052	-.215	.022	-.211	.032							.6000	
.6500															.6500	
.7000	-.161	.009	-.199	.032	-.211	.022	-.224	.012							.7000	
.7500															.7500	
.8000	-.145	.010	-.186	.001	-.206	.007	-.234	.012							.8000	
.8500															.8500	
.9000	-.137	.050	-.177	-.000	-.195	.012	-.235	.005							.9000	
$\alpha = 1.0$																
.0125	-.202	.410	-.303	.449	-.307	.433	-.355	.404	-.323	.404	-.313	.361	-.287	.322	.0125	
.0250	-.223	.333	-.313	.393	-.326	.362	-.365	.378	-.306	.305	-.295	.293	-.281	.249	.0250	
.0500	-.195	.289	-.305	.337	-.328	.344	-.346	.325	-.306	.305	-.295	.293	-.281	.249	.0500	
.0750	-.179	.262	-.300	.302	-.318	.296	-.327	.281	-.306	.305	-.295	.293	-.281	.249	.0750	
.1000	-.149	.216	-.297	.288	-.311	.287	-.318	.270	-.298	.243	-.284	.215	-.272	.153	.1000	
.1500	-.102	.252	-.274	.267	-.307	.253	-.307	.250	-.302	.177	-.289	.174	-.273	.124	.1500	
.2000	-.110	.241	-.261	.248	-.307	.225	-.300	.227	-.302	.177	-.289	.174	-.273	.124	.2000	
.2500	-.069	.216	-.184	.217	-.301	.198	-.301	.186	-.302	.177	-.289	.174	-.273	.124	.2500	
.3000	-.138	.216	-.136	.197	-.299	.171	-.303	.174	-.293	.141	-.291	.146	-.276	.085	.3000	
.3500									-.302	.095	-.291	.102	-.287	.044	.3500	
.4000	-.143	.180	-.181	.142	-.243	.137	-.312	.140	-.317	.074	-.306	.052	-.288	.027	.4000	
.4500									-.323	.054	-.313	.015	-.278	-.024	.4500	
.5000	-.159	.139	-.195	.120	-.217	.065	-.333	.086	-.327	.018	-.318	-.010	-.288	-.080	.5000	
.5500									-.333	.016	-.323	-.039	-.289	-.100	.5500	
.6000	-.167	.124	-.192	.096	-.236	.068	-.339	.084							.6000	
.6500															.6500	
.7000	-.184	.045	-.227	.074	-.246	.062	-.305	.058							.7000	
.7500															.7500	
.8000	-.164	.050	-.209	.037	-.232	.049	-.249	.045							.8000	
.8500															.8500	
.9000	-.160	.090	-.201	.045	-.220	.053	-.263	.052							.9000	

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a) M = 1.61 - Continued

X/C	FRACTION OF SEMISPAN														X/C	
	0.050		0.200		0.350		0.500		0.700		0.825		0.950			
	U	L	U	L	U	L	U	L	U	L	U	L	U	L		
$\alpha = 12$																
.0125	-.259	.464	-.361	.515	-.385	.497	-.417	.442	-.400	.451	-.389	.408	-.372	.376	.0125	
.0250	-.283	.395	-.365	.449	-.390	.426	-.416	.429	-.416	.429	-.382	.382	-.372	.297	.0250	
.0500	-.250	.335	-.356	.391	-.381	.402	-.398	.382	-.372	.353	-.362	.348	-.355	.220	.0500	
.0750	-.239	.314	-.349	.359	-.369	.340	-.381	.339	-.354	.293	-.346	.264	-.337	.172	.0750	
.1000	-.169	.278	-.345	.337	-.361	.336	-.374	.329	-.351	.241	-.352	.215	-.344	.139	.1000	
.1500	-.131	.293	-.391	.317	-.311	.311	-.362	.300	-.346	.200	-.344	.229	-.335	.103	.1500	
.2000	-.131	.289	-.306	.294	-.349	.275	-.355	.284	-.355	.284	-.344	.229	-.335	.078	.2000	
.2500	-.100	.267	-.253	.264	-.347	.245	-.351	.241	-.352	.215	-.344	.229	-.335	.050	.2500	
.3000	-.153	.261	-.227	.239	-.345	.216	-.351	.222	-.346	.177	-.344	.186	-.330	.036	.3000	
.3500															.3500	
.4000	-.168	.222	-.208	.188	-.347	.186	-.355	.188	-.351	.142	-.343	.149	-.334	.103	.4000	
.4500															.4500	
.5000	-.180	.169	-.226	.160	-.281	.109	-.375	.129	-.361	.119	-.359	.100	-.337	.078	.5000	
.5500															.5500	
.6000	-.188	.163	-.217	.137	-.270	.109	-.381	.129	-.365	.094	-.359	.059	-.323	.014	.6000	
.6500															.6500	
.7000	-.205	.085	-.253	.110	-.273	.102	-.386	.097	-.366	.054	-.361	.033	-.331	-.036	.7000	
.7500															.7500	
.8000	-.184	.089	-.231	.077	-.261	.086	-.366	.087	-.370	.050	-.366	-.001	-.333	-.061	.8000	
.8500															.8500	
.9000	-.180	.130	-.225	.087	-.246	.102	-.268	.088							.9000	
$\alpha = 14$																
.0125	-.309	.525	-.418	.556	-.445	.548	-.465	.473	-.450	.494	-.438	.460	-.434	.417	.0125	
.0250	-.335	.454	-.417	.507	-.445	.548	-.462	.487	-.450	.494	-.438	.460	-.434	.417	.0250	
.0500	-.311	.392	-.401	.446	-.433	.448	-.442	.435	-.427	.413	-.415	.398	-.417	.353	.0500	
.0750	-.293	.370	-.392	.414	-.418	.392	-.429	.396	-.427	.396	-.415	.398	-.417	.283	.0750	
.1000	-.200	.316	-.388	.389	-.411	.388	-.421	.382	-.407	.355	-.396	.334	-.399	.237	.1000	
.1500	-.155	.342	-.374	.382	-.400	.361	-.412	.357	-.397	.272	-.389	.295	-.389	.198	.1500	
.2000	-.155	.339	-.357	.339	-.400	.324	-.404	.332	-.397	.272	-.389	.295	-.389	.189	.2000	
.2500	-.126	.326	-.305	.315	-.393	.300	-.396	.285	-.392	.231	-.386	.241	-.381	.150	.2500	
.3000	-.176	.310	-.268	.286	-.392	.269	-.394	.277	-.392	.231	-.386	.241	-.381	.139	.3000	
.3500															.3500	
.4000	-.191	.273	-.264	.232	-.396	.235	-.395	.234	-.393	.186	-.385	.202	-.381	.189	.4000	
.4500															.4500	
.5000	-.202	.219	-.261	.205	-.382	.160	-.410	.178	-.402	.170	-.393	.155	-.381	.145	.5000	
.5500															.5500	
.6000	-.210	.209	-.250	.175	-.334	.147	-.416	.166	-.405	.148	-.396	.128	-.365	.091	.6000	
.6500															.6500	
.7000	-.226	.127	-.273	.151	-.327	.144	-.422	.138	-.406	.116	-.399	.106	-.371	.061	.7000	
.7500															.7500	
.8000	-.206	.133	-.258	.118	-.313	.134	-.420	.135	-.406	.126	-.400	.085	-.374	.092	.8000	
.8500															.8500	
.9000	-.202	.175	-.250	.137	-.287	.161	-.286	.181							.9000	
$\alpha = 16$																
.0125	-.345	.578	-.448	.589	-.473	.584	-.489	.504	-.473	.523	-.467	.508	-.474	.479	.0125	
.0250	-.360	.501	-.442	.556	-.471	.500	-.488	.528	-.473	.523	-.467	.508	-.474	.479	.0250	
.0500	-.342	.444	-.424	.494	-.458	.491	-.473	.479	-.452	.460	-.445	.460	-.459	.462	.0500	
.0750	-.326	.397	-.420	.464	-.446	.439	-.459	.441	-.436	.416	-.427	.398	-.438	.368	.0750	
.1000	-.255	.352	-.419	.443	-.439	.432	-.451	.425	-.436	.416	-.427	.398	-.438	.342	.1000	
.1500	-.172	.385	-.412	.431	-.408	.408	-.440	.401	-.436	.416	-.427	.398	-.438	.283	.1500	
.2000	-.172	.384	-.400	.379	-.428	.363	-.433	.373	-.423	.342	-.417	.361	-.425	.237	.2000	
.2500	-.151	.379	-.365	.368	-.420	.343	-.426	.323	-.423	.342	-.417	.361	-.425	.189	.2500	
.3000	-.188	.352	-.332	.331	-.419	.310	-.421	.321	-.415	.302	-.411	.310	-.416	.150	.3000	
.3500															.3500	
.4000	-.208	.318	-.284	.278	-.422	.277	-.424	.274	-.413	.261	-.409	.279	-.415	.240	.4000	
.4500															.4500	
.5000	-.220	.265	-.266	.245	-.422	.199	-.434	.219	-.421	.242	-.415	.246	-.408	.283	.5000	
.5500															.5500	
.6000	-.224	.248	-.256	.215	-.377	.181	-.439	.237	-.423	.223	-.416	.245	-.397	.242	.6000	
.6500															.6500	
.7000	-.241	.163	-.279	.190	-.363	.196	-.442	.216	-.425	.185	-.418	.267	-.400	.253	.7000	
.7500															.7500	
.8000	-.219	.175	-.262	.169	-.354	.205	-.442	.209	-.378	.264	-.419	.355	-.402	.212	.8000	
.8500															.8500	
.9000	-.216	.227	-.254	.220	-.323	.254	-.323	.275							.9000	
$\alpha = 18$																
.0125	-.384	.637	-.462	.640	-.498	.629	-.510	.536	-.502	.557	-.498	.576	-.500	.609	.0125	
.0250	-.404	.562	-.456	.606	-.497	.534	-.508	.573	-.502	.557	-.498	.576	-.500	.609	.0250	
.0500	-.381	.507	-.454	.556	-.487	.538	-.497	.525	-.489	.549	-.490	.556	-.494	.470	.0500	
.0750	-.393	.468	-.456	.528	-.477	.486	-.488	.497	-.489	.549	-.490	.556	-.494	.470	.0750	
.1000	-.379	.414	-.444	.484	-.472	.494	-.481	.483	-.474	.502	-.475	.527	-.488	.520	.1000	
.1500	-.179	.447	-.446	.486	-.465	.471	-.469	.469	-.469	.502	-.475	.527	-.488	.476	.1500	
.2000	-.187	.443	-.391	.433	-.459	.427	-.464	.451	-.460	.502	-.475	.527	-.488	.440	.2000	
.2500	-.168	.448	-.343	.430	-.451	.400	-.458	.427	-.460	.502	-.475	.527	-.488	.440	.2500	
.3000	-.203	.406	-.336	.384	-.448	.367	-.453	.421	-.453	.383	-.459	.474	-.474	.440	.3000	
.3500															.3500	
.4000	-.225	.377	-.335	.335	-.450	.338	-.453	.380	-.453	.364	-.454	.463	-.470	.368	.4000	
.4500															.4500	
.5000	-.240	.325	-.333	.305	-.418	.290	-.461	.328	-.456	.386	-.455	.469	-.457	.292	.5000	
.5500															.5500	
.6000	-.244	.299	-.345	.279	-.387	.295	-.464	.319	-.406	.420	-.342	.428	-.445	.261	.6000	
.6500															.6500	
.7000	-.260	.220	-.362	.289	-.387	.295	-.467	.294	-.327	.451	-.327	.374	-.362	.214	.7000	
.7500															.7500	
.8000	-.233	.281	-.337	.293	-.390	.300	-.390	.323	-.329	.434	-.320	.363	-.328		.8000	
.8500															.8500	
.9000	-.234	.336	-.306	.314	-.379	.361	-.331	.450							.9000	

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(a) M = 1.61 - Concluded

X/C		FRACTION OF SEMISPAN														X/C	
		0.050		0.200		0.350		0.500		0.700		0.825		0.950			
		U	L	U	L	U	L	U	L	U	L	U	L	U	L		
		$\alpha = 20$															
.0125	-.408	.685	-.467	.679	-.503	.651	-.516	.656	-.505	.683	-.498	.638	-.499	.655	.0125		
.0250	-.418	.621	-.467	.647	-.498	.684	-.515	.650	-.505	.683	-.498	.638	-.499	.655	.0250		
.0500	-.401	.561	-.467	.604	-.495	.582	-.508	.617	-.498	.614	-.496	.672	-.492	.621	.0500		
.0750	-.406	.545	-.467	.591	-.489	.543	-.503	.596	-.498	.614	-.496	.672	-.492	.621	.0750		
.1000	-.413	.473	-.456	.577	-.488	.547	-.499	.584	-.490	.590	-.488	.651	-.487	.645	.1000		
.1500	-.296	.504	-.428	.525	-.481	.487	-.483	.535	-.475	.500	-.484	.645	-.481	.535	.1500		
.2000	-.188	.500	-.369	.486	-.475	.480	-.475	.500	-.475	.500	-.484	.645	-.481	.535	.2000		
.2500	-.189	.506	-.339	.476	-.467	.482	-.471	.494	-.475	.537	-.484	.560	-.479	.490	.2500		
.3000	-.209	.456	-.328	.431	-.467	.482	-.471	.494	-.475	.537	-.484	.560	-.479	.490	.3000		
.3500															.3500		
.4000	-.238	.429	-.343	.387	-.467	.452	-.467	.456	-.473	.554	-.479	.527	-.468	.452	.4000		
.4500															.4500		
.5000	-.254	.362	-.335	.410	-.464	.363	-.475	.395	-.473	.543	-.475	.466	-.456	.387	.5000		
.5500															.5500		
.6000	-.257	.374	-.322	.378	-.433	.368	-.477	.413	-.473	.543	-.475	.466	-.456	.387	.6000		
.6500															.6500		
.7000	-.272	.319	-.331	.364	-.415	.364	-.479	.462	-.386	.501	-.415	.422	-.437	.313	.7000		
.7500															.7500		
.8000	-.241	.379	-.342	.376	-.412	.408	-.377	.522	-.347	.449	-.381	.372	-.415	.276	.8000		
.8500															.8500		
.9000	-.251	.423	-.340	.417	-.391	.495	-.373	.520	-.355	.427	-.365	.370	-.404	.225	.9000		

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b)  $M = 2.01$

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
	α=-20														
.0125	.671	-.254	.695	-.320	.675	-.322	.641	-.328							.0125
.0250	.642	-.268	.686	-.314	.697	-.322	.662	-.326	.626	-.277	.619	-.290	.519	-.283	.0250
.0500	.561	-.240	.618	-.306	.649	-.322	.676	-.325							.0500
.0750	.531	-.266	.600	-.303	.638	-.320	.626	-.326	.589	-.318	.598	-.290	.554	-.287	.0750
.1000	.521	-.265	.583	-.304	.634	-.321	.635	-.325							.1000
.1500	.491	-.265	.558	-.304	.623	-.323	.621	-.324	.577	-.309	.606	-.286	.578	-.290	.1500
.2000	.467	-.237	.521	-.304	.546	-.326	.568	-.327							.2000
.2500	.481	-.172	.518	-.303	.524	-.329	.541	-.327	.519	-.311	.577	-.282	.572	-.291	.2500
.3000	.470	-.121	.511	-.299	.488	-.328	.529	-.326							.3000
.3500									.490	-.312	.528	-.282	.584	-.291	.3500
.4000	.451	-.221	.433	-.289	.453	-.327	.464	-.327	.466	-.307	.499	-.283	.545	-.293	.4000
.4500															.4500
.5000	.392	-.192	.406	-.294	.417	-.322	.425	-.329	.436	-.308	.465	-.292	.543	-.292	.5000
.5500															.5500
.6000	.364	-.207	.372	-.293	.380	-.317	.383	-.328	.436	-.308	.465	-.292	.543	-.292	.6000
.6500															.6500
.7000	.343	-.222	.293	-.295	.345	-.311	.363	-.312	.414	-.303	.442	-.294	.505	-.293	.7000
.7500															.7500
.8000	.296	-.205	.308	-.299	.334	-.304	.360	-.301	.400	-.303	.433	-.294	.455	-.294	.8000
.8500															.8500
.9000	.331	-.205	.300	-.300	.326	-.292	.351	-.289	.393	-.298	.437	-.294	.437	-.296	.9000
α=-18															
.0125	.621	-.231	.649	-.298	.657	-.314	.613	-.334	.603	-.276	.597	-.301	.519	-.289	.0125
.0250	.584	-.248	.638	-.303	.664	-.308	.615	-.331							.0250
.0500	.508	-.240	.569	-.288	.611	-.306	.629	-.330	.554	-.320	.565	-.307	.536	-.291	.0500
.0750	.471	-.247	.546	-.283	.592	-.314	.586	-.330							.0750
.1000	.461	-.247	.525	-.280	.582	-.315	.588	-.329	.537	-.311	.568	-.303	.547	-.294	.1000
.1500	.431	-.235	.497	-.280	.514	-.318	.569	-.329	.537	-.311	.568	-.303	.547	-.294	.1500
.2000	.412	-.170	.467	-.281	.514	-.322	.516	-.329							.2000
.2500	.419	-.147	.457	-.282	.469	-.322	.489	-.328	.482	-.313	.534	-.300	.538	-.294	.2500
.3000	.414	-.110	.446	-.279	.445	-.324	.479	-.327							.3000
.3500									.446	-.313	.486	-.301	.543	-.294	.3500
.4000	.396	-.215	.383	-.263	.410	-.320	.411	-.327	.421	-.308	.452	-.297	.502	-.294	.4000
.4500															.4500
.5000	.341	-.180	.360	-.269	.373	-.317	.372	-.327	.387	-.310	.417	-.304	.499	-.294	.5000
.5500															.5500
.6000	.313	-.193	.338	-.271	.345	-.319	.335	-.327	.367	-.304	.391	-.304	.457	-.294	.6000
.6500															.6500
.7000	.294	-.210	.255	-.273	.306	-.308	.310	-.313	.347	-.301	.380	-.301	.406	-.294	.7000
.7500															.7500
.8000	.245	-.193	.271	-.271	.296	-.297	.306	-.292	.342	-.294	.377	-.299	.387	-.294	.8000
.8500															.8500
.9000	.282	-.192	.258	-.270	.288	-.286	.301	-.262							.9000
α=-16															
.0125	.564	-.205	.604	-.275	.618	-.310	.597	-.332	.579	-.279	.576	-.307	.513	-.317	.0125
.0250	.525	-.226	.576	-.284	.609	-.310	.581	-.332							.0250
.0500	.452	-.220	.509	-.282	.551	-.309	.568	-.329	.519	-.321	.535	-.315	.510	-.318	.0500
.0750	.403	-.224	.486	-.276	.524	-.308	.539	-.329							.0750
.1000	.403	-.223	.458	-.273	.516	-.305	.527	-.326	.489	-.309	.522	-.311	.507	-.319	.1000
.1500	.372	-.211	.434	-.269	.468	-.299	.462	-.318							.1500
.2000	.355	-.113	.400	-.265	.468	-.299	.462	-.318	.432	-.308	.489	-.310	.494	-.316	.2000
.2500	.356	-.139	.401	-.261	.468	-.299	.462	-.318							.2500
.3000	.361	-.088	.382	-.261	.386	-.294	.421	-.314	.396	-.313	.438	-.308	.498	-.318	.3000
.3500															.3500
.4000	.336	-.213	.314	-.261	.346	-.296	.358	-.311	.368	-.308	.404	-.305	.454	-.318	.4000
.4500															.4500
.5000	.290	-.166	.299	-.254	.316	-.298	.316	-.314	.339	-.309	.368	-.318	.448	-.318	.5000
.5500															.5500
.6000	.261	-.181	.282	-.240	.288	-.300	.286	-.316	.319	-.306	.341	-.318	.407	-.318	.6000
.6500															.6500
.7000	.243	-.202	.202	-.223	.254	-.293	.263	-.307	.301	-.308	.328	-.318	.355	-.316	.7000
.7500															.7500
.8000	.195	-.186	.221	-.218	.241	-.288	.253	-.301	.291	-.308	.320	-.315	.332	-.317	.8000
.8500															.8500
.9000	.231	-.180	.202	-.219	.237	-.280	.248	-.242							.9000
α=-14															
.0125	.507	-.180	.561	-.247	.583	-.290	.570	-.328	.560	-.274	.550	-.304	.503	-.323	.0125
.0250	.473	-.205	.525	-.261	.565	-.295	.542	-.327							.0250
.0500	.404	-.200	.460	-.261	.502	-.294	.550	-.320	.487	-.319	.505	-.312	.487	-.323	.0500
.0750	.341	-.202	.426	-.256	.465	-.290	.498	-.317							.0750
.1000	.347	-.191	.407	-.253	.453	-.288	.485	-.311	.446	-.307	.482	-.306	.471	-.323	.1000
.1500	.319	-.121	.378	-.251	.428	-.286	.459	-.305	.390	-.306	.446	-.306	.450	-.321	.1500
.2000	.305	-.122	.343	-.248	.409	-.284	.416	-.303							.2000
.2500	.303	-.134	.346	-.243	.383	-.281	.386	-.300	.355	-.306	.397	-.306	.447	-.322	.2500
.3000	.314	-.084	.326	-.237	.337	-.282	.373	-.300	.355	-.306	.397	-.306	.447	-.322	.3000
.3500															.3500
.4000	.280	-.202	.266	-.234	.296	-.282	.308	-.295	.322	-.302	.358	-.301	.406	-.321	.4000
.4500															.4500
.5000	.249	-.157	.251	-.221	.265	-.282	.269	-.301	.292	-.303	.324	-.318	.401	-.323	.5000
.5500															.5500
.6000	.217	-.171	.237	-.211	.238	-.283	.238	-.303	.274	-.300	.295	-.319	.356	-.320	.6000
.6500															.6500
.7000	.198	-.191	.160	-.202	.211	-.280	.219	-.295	.260	-.302	.285	-.320	.306	-.322	.7000
.7500															.7500
.8000	.149	-.176	.178	-.207	.193	-.278	.213	-.295	.249	-.301	.275	-.320	.285	-.320	.8000
.8500															.8500
.9000	.185	-.169	.158	-.209	.191	-.272	.203	-.251							.9000

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b)  $M = 2.01$  - Continued

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
$\alpha = -12$															
.0125	.448	-.151	.512	-.213	.543	-.261	.531	-.310	.538	-.266	.528	-.306	.498	-.326	.0125
.0250	.412	-.176	.473	-.234	.523	-.272	.501	-.308							.0250
.0500	.355	-.172	.416	-.240	.458	-.272	.501	-.299							.0500
.0750	.287	-.165	.373	-.232	.411	-.268	.447	-.295	.458	-.308	.477	-.313	.472	-.325	.0750
.1000	.294	-.156	.353	-.232	.394	-.263	.435	-.289							.1000
.1500	.270	-.120	.325	-.230	.358	-.263	.417	-.283	.408	-.296	.440	-.304	.443	-.325	.1500
.2000	.256	-.102	.292	-.228	.339	-.261	.363	-.280							.2000
.2500	.253	-.118	.292	-.227	.331	-.260	.336	-.279	.346	-.290	.405	-.302	.411	-.322	.2500
.3000	.267	-.066	.279	-.216	.280	-.264	.325	-.280							.3000
.3500									.311	-.288	.353	-.302	.405	-.321	.3500
.4000	.233	-.184	.214	-.180	.247	-.267	.262	-.277							.4000
.4500									.277	-.286	.317	-.298	.364	-.318	.4500
.5000	.205	-.142	.203	-.180	.218	-.266	.222	-.283							.5000
.5500									.250	-.290	.281	-.308	.361	-.319	.5500
.6000	.173	-.157	.191	-.198	.190	-.265	.191	-.286							.6000
.6500									.228	-.290	.253	-.311	.314	-.321	.6500
.7000	.154	-.178	.123	-.188	.169	-.264	.173	-.279							.7000
.7500									.217	-.293	.240	-.312	.263	-.318	.7500
.8000	.105	-.165	.135	-.196	.150	-.265	.169	-.279							.8000
.8500									.206	-.294	.231	-.314	.240	-.315	.8500
.9000	.145	-.157	.118	-.199	.147	-.264	.161	-.270							.9000
$\alpha = -10$															
.0125	.393	-.105	.461	-.169	.501	-.225	.490	-.270	.506		.487	-.307	.492	-.330	.0125
.0250	.347	-.121	.423	-.198	.473	-.237	.451	-.272		.451					.0250
.0500	.305	-.133	.369	-.203	.408	-.239	.441	-.266							.0500
.0750	.238	-.133	.313	-.198	.354	-.233	.386	-.259	.435	-.287	.446	-.301	.453	-.326	.0750
.1000	.245	-.121	.299	-.197	.335	-.229	.378	-.255							.1000
.1500	.230	-.100	.273	-.196		-.229	.367	-.251	.375	-.274	.391	-.288	.415	-.317	.1500
.2000	.211	-.077	.239	-.195	.300	-.229	.322	-.248							.2000
.2500	.207	-.095	.234	-.191	.280	-.227	.291	-.247	.297	-.269	.360	-.283	.362	-.312	.2500
.3000	.227	-.051	.236	-.174	.239	-.233	.275	-.251							.3000
.3500									.271	-.269	.308	-.281	.349	-.307	.3500
.4000	.189	-.147	.166	-.145	.201	-.238	.219	-.250							.4000
.4500									.224	-.270	.267	-.281	.309	-.298	.4500
.5000	.159	-.120	.154	-.162	.170	-.247	.175	-.259							.5000
.5500									.200	-.273	.227	-.288	.316	-.294	.5500
.6000	.140	-.140	.143	-.179	.143	-.240	.148	-.261							.6000
.6500									.178	-.275	.204	-.293	.270	-.289	.6500
.7000	.118	-.159	.085	-.172	.123	-.217	.130	-.262							.7000
.7500									.166	-.278	.191	-.289	.215	-.287	.7500
.8000	.069	-.148	.095	-.180	.113	-.207	.124	-.264							.8000
.8500									.156	-.276	.180	-.285	.193	-.284	.8500
.9000	.105	-.141	.079	-.183	.108	-.209	.123	-.263							.9000
$\alpha = -08$															
.0125	.347	-.050	.412	-.124	.460	-.182	.456	-.223							.0125
.0250	.288	-.059	.373	-.157	.423	-.201	.405	-.235	.474	-.216	.451	-.295	.478	-.322	.0250
.0500	.252	-.112	.322	-.166	.363	-.204	.385	-.231							.0500
.0750	.198	-.109	.259	-.162	.307	-.198	.332	-.226	.402	-.256	.418	-.277	.445	-.308	.0750
.1000	.199	-.098	.249	-.162	.286	-.193	.320	-.222							.1000
.1500	.190	-.063	.227	-.163	.249	-.197	.311	-.218	.340	-.244	.348	-.262	.385	-.296	.1500
.2000	.171	-.060	.195	-.163	.249	-.196	.275	-.216							.2000
.2500	.167	-.076	.189	-.159	.229	-.198	.252	-.217	.254	-.240	.320	-.256	.323	-.288	.2500
.3000	.189	-.044	.191	-.137	.205	-.204	.235	-.222							.3000
.3500									.234	-.244	.271	-.258	.313	-.285	.3500
.4000	.149	-.116	.127	-.120	.156	-.208	.180	-.222							.4000
.4500									.186	-.246	.229	-.259	.269	-.283	.4500
.5000	.122	-.106	.113	-.147	.128	-.217	.131	-.233	.157	-.251	.186	-.266	.266	-.274	.5000
.5500									.157	-.251	.186	-.266	.266	-.274	.5500
.6000	.105	-.124	.102	-.163	.102	-.192	.107	-.235	.139	-.254	.164	-.272	.226	-.266	.6000
.6500									.139	-.254	.164	-.272	.226	-.266	.6500
.7000	.084	-.146	.053	-.155	.084	-.181	.089	-.238							.7000
.7500									.126	-.258	.149	-.275	.175	-.263	.7500
.8000	.037	-.135	.062	-.166	.077	-.189	.085	-.238							.8000
.8500									.116	-.258	.139	-.271	.151	-.258	.8500
.9000	.071	-.127	.049	-.170	.072	-.193	.084	-.233							.9000
$\alpha = -06$															
.0125	.294	-.036	.347	-.057	.402	-.120	.402	-.172							.0125
.0250	.228	-.015	.307	-.102	.362	-.146	.354	-.188	.422	-.168	.406	-.258	.457	-.299	.0250
.0500	.193	-.088	.261	-.117	.305	-.155	.324	-.184							.0500
.0750	.153	-.076	.197	-.116	.252	-.152	.271	-.180	.349	-.217	.374	-.235	.399	-.277	.0750
.1000	.145	-.064	.190	-.120	.225	-.148	.250	-.177							.1000
.1500	.142	-.040	.171	-.126	.157	-.157	.252	-.176	.287	-.206	.292	-.225	.341	-.263	.1500
.2000	.125	-.034	.145	-.124	.193	-.162	.217	-.176							.2000
.2500	.119	-.051	.135	-.114	.176	-.159	.195	-.178	.211	-.204	.266	-.221	.273	-.257	.2500
.3000	.147	-.034	.137	-.102	.156	-.167	.184	-.186							.3000
.3500									.189	-.213	.228	-.227	.263	-.259	.3500
.4000	.105	-.095	.087	-.097	.112	-.173	.133	-.196							.4000
.4500									.144	-.211	.188	-.232	.220	-.260	.4500
.5000	.078	-.086	.070	-.133	.083	-.163	.084	-.204	.110	-.221	.140	-.242	.216	-.260	.5000
.5500									.110	-.221	.140	-.242	.216	-.260	.5500
.6000	.062	-.106	.061	-.141	.058	-.153	.063	-.207	.093	-.224	.121	-.248	.176	-.260	.6000
.6500									.093	-.224	.121	-.248	.176	-.260	.6500
.7000	.041	-.130	.018	-.136	.041	-.160	.044	-.209							.7000
.7500									.081	-.233	.104	-.255	.125	-.253	.7500
.8000	.003	-.122	.023	-.149	.036	-.175	.040	-.205							.8000
.8500									.069	-.236	.093	-.261	.101	-.246	.8500
.9000	.032	-.112	.016	-.155	.032	-.171	.038	-.184							.9000



TABLE VI.- Continued

PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b) M = 2.01 - Continued

X/C		FRACTION OF SEMISPAN														X/C
		0.050		0.200		0.350		0.500		0.700		0.825		0.950		
		U	L	U	L	U	L	U	L	U	L	U	L	U	L	
		α = -0.4														
.0125	.254	.096	.294	.014	.353	-.042	.359	-.114	.378	-.112	.363	-.219	.424	-.257	.0125	
.0250	.190	.018	.262	-.086	.313	-.086	.309	-.129	.378	-.112	.363	-.219	.424	-.257	.0250	
.0500	.148	-.058	.222	-.071	.263	-.107	.279	-.133	.299	-.174	.326	-.193	.356	-.237	.0500	
.0750	.124	-.042	.164	-.073	.214	-.111	.228	-.130	.299	-.174	.326	-.193	.356	-.237	.0750	
.1000	.113	-.030	.142	-.078	.182	-.109	.196	-.130	.299	-.174	.326	-.193	.356	-.237	.1000	
.1500	.110	-.012	.131	-.084	.182	-.118	.203	-.135	.237	-.165	.245	-.183	.302	-.225	.1500	
.2000	.101	-.012	.111	-.074	.155	-.121	.171	-.138	.237	-.165	.245	-.183	.302	-.225	.2000	
.2500	.092	-.022	.102	-.064	.138	-.119	.154	-.143	.166	-.163	.217	-.184	.228	-.221	.2500	
.3000	.118	-.019	.101	-.065	.117	-.131	.135	-.152	.166	-.163	.217	-.184	.228	-.221	.3000	
.3500									.153	-.173	.190	-.193	.224	-.225	.3500	
.4000	.075	-.069	.058	-.073	.088	-.125	.099	-.154	.114	-.176	.154	-.201	.186	-.229	.4000	
.4500									.114	-.176	.154	-.201	.186	-.229	.4500	
.5000	.051	-.064	.040	-.109	.053	-.128	.046	-.172	.073	-.190	.104	-.214	.177	-.234	.5000	
.5500									.073	-.190	.104	-.214	.177	-.234	.5500	
.6000	.039	-.085	.030	-.110	.028	-.131	.030	-.174	.055	-.195	.083	-.221	.135	-.238	.6000	
.6500									.055	-.195	.083	-.221	.135	-.238	.6500	
.7000	.018	-.111	-.007	-.108	.011	-.145	.010	-.172	.055	-.195	.083	-.221	.135	-.238	.7000	
.7500									.049	-.206	.062	-.229	.088	-.237	.7500	
.8000	-.019	-.106	-.001	-.125	.007	-.157	.006	-.154	.037	-.207	.058	-.237	.061	-.229	.8000	
.8500									.037	-.207	.058	-.237	.061	-.229	.8500	
.9000	.009	-.094	-.007	-.130	.006	-.148	.005	-.158							.9000	
α = -0.2																
.0125	.210	.142	.219	.073	.290	.019	.298	.019	.326	-.041	.208	-.154	.389	-.204	.0125	
.0250	.146	.052	.188	.001	.248	-.040	.252	-.058	.326	-.041	.208	-.154	.389	-.204	.0250	
.0500	.088	-.026	.169	-.1026	.207	.061	.237	-.074	.326	-.041	.208	-.154	.389	-.204	.0500	
.0750	.095	-.006	.125	-.030	.164	-.059	.193	-.081	.242	-.120	.269	-.133	.309	-.188	.0750	
.1000	.077	-.002	.089	-.036	.119	-.053	.148	-.080	.170	-.112	.200	-.131	.256	-.182	.1000	
.1500	.073	.011	.080	-.036	.114	-.065	.145	-.088	.170	-.112	.200	-.131	.256	-.182	.1500	
.2000	.067	.012	.069	-.033	.097	-.075	.121	-.094	.132	-.115	.167	-.135	.176	-.180	.2000	
.2500	.060	.005	.062	-.034	.092	-.070	.109	-.095	.132	-.115	.167	-.135	.176	-.180	.2500	
.3000	.079	-.003	.058	-.042	.068	-.078	.092	-.105	.107	-.129	.148	-.150	.178	-.186	.3000	
.3500									.107	-.129	.148	-.150	.178	-.186	.3500	
.4000	.040	-.043	.027	-.055	.048	-.081	.061	-.110	.075	-.141	.115	-.161	.143	-.193	.4000	
.4500									.075	-.141	.115	-.161	.143	-.193	.4500	
.5000	.018	-.041	.006	-.089	.013	-.100	.016	-.124	.044	-.154	.071	-.179	.139	-.201	.5000	
.5500									.044	-.154	.071	-.179	.139	-.201	.5500	
.6000	.008	-.065	-.002	-.092	-.010	-.109	-.009	-.122	.021	-.160	.052	-.187	.099	-.205	.6000	
.6500									.021	-.160	.052	-.187	.099	-.205	.6500	
.7000	-.012	-.091	-.035	-.090	-.028	-.125	-.021	-.122	.008	-.170	.033	-.195	.055	-.207	.7000	
.7500									.008	-.170	.033	-.195	.055	-.207	.7500	
.8000	-.043	-.091	-.029	-.111	-.030	-.131	-.027	-.126	-.005	-.171	.021	-.206	.027	-.199	.8000	
.8500									-.005	-.171	.021	-.206	.027	-.199	.8500	
.9000	-.018	-.073	-.035	-.119	-.028	-.123	-.029	-.137							.9000	
α = 0.0																
.0125	.163	.188	.154	.127	.223	.074	.230	.063	.270	.039	.257	-.079	.339	-.137	.0125	
.0250	.121	.087	.121	.090	.176	.013	.191	.002	.270	.039	.257	-.079	.339	-.137	.0250	
.0500	.044	.014	.107	.027	.151	.002	.191	-.024	.270	.039	.257	-.079	.339	-.137	.0500	
.0750	.045	.031	.079	.024	.124	.012	.164	-.022	.270	.039	.257	-.079	.339	-.137	.0750	
.1000	.047	.015	.048	.016	.088	.007	.113	-.017	.270	.039	.257	-.079	.339	-.137	.1000	
.1500	.045	.037	.035	.012	.088	-.015	.088	-.031	.270	.039	.257	-.079	.339	-.137	.1500	
.2000	.038	.037	.027	.006	.045	-.022	.066	-.034	.270	.039	.257	-.079	.339	-.137	.2000	
.2500	.039	.021	.031	-.003	.047	-.017	.078	.063	.270	.039	.257	-.079	.339	-.137	.2500	
.3000	.041	.021	.023	-.017	.032	-.036	.044	-.058	.270	.039	.257	-.079	.339	-.137	.3000	
.3500									.270	.039	.257	-.079	.339	-.137	.3500	
.4000	.011	-.013	-.005	-.033	.012	-.047	.023	-.058	.062	-.081	.097	-.108	.126	-.133	.4000	
.4500									.062	-.081	.097	-.108	.126	-.133	.4500	
.5000	-.011	-.016	-.028	-.054	-.026	-.077	-.016	-.086	.033	-.090	.074	-.115	.091	-.146	.5000	
.5500									.033	-.090	.074	-.115	.091	-.146	.5500	
.6000	.019	-.041	-.035	-.062	-.039	-.083	-.041	-.091	.004	-.109	.032	-.136	.092	-.160	.6000	
.6500									.004	-.109	.032	-.136	.092	-.160	.6500	
.7000	-.037	-.070	-.062	-.065	-.058	-.096	-.056	-.100	-.010	-.119	.014	-.144	.053	-.172	.7000	
.7500									-.010	-.119	.014	-.144	.053	-.172	.7500	
.8000	-.061	-.075	-.057	-.085	-.058	-.099	-.055	-.109	-.021	-.133	-.003	-.152	.016	-.168	.8000	
.8500									-.021	-.133	-.003	-.152	.016	-.168	.8500	
.9000	-.038	-.053	-.062	-.095	-.054	-.096	-.061	-.116	-.032	-.126	-.012	-.168	-.010	-.164	.9000	
α = 0.2																
.0125	.113	.233	.093	.181	.154	.142	.141	.136	.193	.119	.193	.017	.288	-.046	.0125	
.0250	.065	.124	.060	.110	.097	.090	.105	.072	.193	.119	.193	.017	.288	-.046	.0250	
.0500	.007	.073	.043	.106	.076	.091	.111	.064	.149	.025	.164	-.002	.197	-.040	.0500	
.0750	.006	.086	.059	.084	.074	.086	.119	.074	.149	.025	.164	-.002	.197	-.040	.0750	
.1000	.013	.042	.030	.059	.051	.058	.078	.063	.149	.025	.164	-.002	.197	-.040	.1000	
.1500	.018	.066	-.011	.058	.027	.037	.037	.014	.058	.030	.102	-.002	.164	-.054	.1500	
.2000	.014	.067	-.008	.048	-.005	.028	.017	.009	.058	.030	.102	-.002	.164	-.054	.2000	
.2500	.041	.038	.001	.029	-.006	.025	.007	.013	.046	-.010	.056	-.015	.085	-.059	.2500	
.3000	.005	.058	-.004	.015	-.008	-.001	-.010	-.016	.015	-.039	.045	-.049	.058	-.081	.3000	
.3500									.015	-.039	.045	-.049	.058	-.081	.3500	
.4000	-.030	.019	-.031	-.001	-.023	-.016	-.028	-.020	-.012	-.048	.028	-.062	.040	-.093	.4000	
.4500									-.012	-.048	.028	-.062	.040	-.093	.4500	
.5000	-.039	.013	-.065	-.020	-.047	-.051	-.055	-.057	-.040	-.066	-.010	-.096	.043	-.106	.5000	
.5500									-.040	-.066	-.010	-.096	.043	-.106	.5500	
.6000	-.045	-.013	-.063	-.032	-.065	-.058	-.069	-.056	-.056	-.074	-.026	-.109	.011	-.125	.6000	
.6500									-.056	-.074	-.026	-.109	.011	-.125	.6500	
.7000	-.062	-.048	-.085	-.040	-.092	-.069	-.086	-.072	-.063	-.095	-.043	-.119	-.022	-.129	.7000	
.7500									-.063	-.095	-.043	-.119	-.022	-.129	.7500	
.8000	-.074	-.059	-.078	-.062	-.095	-.073	-.083	-.084	-.072	-.102	-.093	-.133	-.045	-.130	.8000	
.8500									-.072	-.102	-.093	-.133	-.045	-.130	.8500	
.9000	-.051	-.037	-.082	-.073	-.085	-.072	-.091	-.01							.9000	

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b) M = 2.01 - Continued

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
$\alpha = 0.4$															
.0125	.057	.277	.030	.268	.089	.224	.050	.224	.096	.213	.107	.125	.221	.104	.0125
.0250	.021	.167	.004	.196	.028	.172	.026	.162	.059	.122	.084	.103	.128	.066	.0250
.0500	-.024	.130	-.015	.170	.005	.162	.027	.144	.059	.122	.084	.103	.128	.066	.0500
.0750	-.034	.089	-.022	.133	.006	.135	.033	.139	.059	.122	.084	.103	.128	.066	.0750
.1000	-.021	.072	-.041	.116	-.003	.104	.022	.111	.059	.122	.084	.103	.128	.066	.1000
.1500	-.006	.104	-.060	.114	.085	.085	-.015	.070	.024	.076	.062	.081	.108	.033	.1500
.2000	-.012	.099	-.055	.091	-.045	.076	-.030	.074	.024	.076	.062	.081	.108	.033	.2000
.2500	.019	.071	-.044	.066	-.053	.069	-.034	.067	.001	.043	.010	.022	.056	.021	.2500
.3000	-.019	.100	-.034	.057	-.064	.038	-.049	.033	.001	.043	.010	.022	.056	.021	.3000
.3500															.3500
.4000	-.061	.051	-.049	.034	-.063	.027	-.071	.021	-.033	.022	-.007	.003	.020	-.030	.4000
.4500															.4500
.5000	-.066	.042	-.089	.015	-.071	-.019	-.107	-.012	-.048	-.005	-.020	-.005	-.002	-.053	.5000
.5500															.5500
.6000	-.069	.017	-.089	.000	-.087	-.026	-.118	-.020	-.078	-.027	-.049	-.035	-.011	-.069	.6000
.6500															.6500
.7000	-.083	-.022	-.108	-.009	-.113	-.038	-.115	-.048	-.092	-.039	-.065	-.053	-.037	-.081	.7000
.7500															.7500
.8000	-.091	-.042	-.098	-.033	-.125	-.044	-.119	-.055	-.102	-.064	-.080	-.076	-.063	-.090	.8000
.8500															.8500
.9000	-.069	-.014	-.101	-.052	-.112	-.042	-.114	-.055	-.111	-.070	-.089	-.099	-.078	-.087	.9000
$\alpha = 0.6$															
.0125	.002	.315	-.026	.324	.026	.291	.027	.294	.015	.284	.030	.218	.138	.192	.0125
.0250	-.023	.208	-.053	.248	-.030	.239	.046	.235	.015	.284	.030	.218	.138	.192	.0250
.0500	-.058	.187	-.065	.216	-.054	.218	.037	.206	.007	.186	.014	.178	.054	.135	.0500
.0750	-.068	.126	-.069	.176	-.057	.181	.030	.193	.007	.186	.014	.178	.054	.135	.0750
.1000	-.054	.104	-.081	.168	-.061	.153	.034	.156	.032	.132	.005	.135	.023	.093	.1000
.1500	-.039	.137	-.099	.145	-.091	.125	.057	.130	.032	.132	.005	.135	.023	.093	.1500
.2000	-.037	.130	-.103	.126	-.091	.125	.057	.126	.043	.104	-.037	.087	.003	.064	.2000
.2500	-.004	.100	-.089	.099	-.099	.107	.077	.106	.043	.104	-.037	.087	.003	.064	.2500
.3000	-.040	.132	-.069	.088	-.107	.079	-.089	.073	.070	.058	-.052	.049	-.023	.020	.3000
.3500															.3500
.4000	-.080	.080	-.079	.060	-.120	.065	-.106	.061	.083	.038	-.064	.035	-.048	.005	.4000
.4500															.4500
.5000	-.087	.070	-.114	.040	-.113	.012	-.138	.029	.108	.009	-.087	-.005	-.052	-.018	.5000
.5500															.5500
.6000	-.090	.044	-.122	.023	-.119	.002	-.152	.005	.122	-.005	-.099	-.025	-.077	-.043	.6000
.6500															.6500
.7000	-.103	.001	-.133	.013	-.138	-.005	-.161	-.016	.130	-.031	-.111	-.049	-.094	-.066	.7000
.7500															.7500
.8000	-.107	-.025	-.124	-.012	-.149	-.015	-.159	-.023	.139	-.041	-.120	-.071	-.099	-.085	.8000
.8500															.8500
.9000	-.091	.008	-.126	-.028	-.148	-.015	-.143	-.025	.139	-.041	-.120	-.071	-.099	-.085	.9000
$\alpha = 0.8$															
.0125	-.057	.358	-.078	.382	-.045	.358	-.099	.358	-.058	.355	-.040	.305	.044	.273	.0125
.0250	-.079	.259	-.101	.311	-.086	.304	-.110	.308	.072	.254	-.051	.245	-.012	.210	.0250
.0500	-.090	.233	-.113	.267	-.106	.274	-.097	.260	.072	.254	-.051	.245	-.012	.210	.0500
.0750	-.098	.166	-.117	.230	-.106	.226	-.088	.251	.072	.254	-.051	.245	-.012	.210	.0750
.1000	-.084	.156	-.124	.218	-.109	.206	-.088	.208	.084	.192	-.055	.188	-.038	.153	.1000
.1500	-.070	.176	-.139	.191	-.184	.184	-.100	.189	.084	.192	-.055	.188	-.038	.153	.1500
.2000	-.058	.171	-.141	.164	-.131	.169	-.110	.177	.095	.149	-.082	.143	-.055	.104	.2000
.2500	-.025	.138	-.132	.143	-.136	.151	-.119	.152	.095	.149	-.082	.143	-.055	.104	.2500
.3000	-.055	.166	-.109	.132	-.146	.124	-.126	.121	.112	.118	-.095	.098	-.073	.076	.3000
.3500															.3500
.4000	-.105	.116	-.100	.095	-.155	.099	-.142	.104	.123	.082	-.107	.088	-.091	.040	.4000
.4500															.4500
.5000	-.105	.104	-.130	.074	-.171	.049	-.169	.057	.142	.047	-.126	.037	-.101	.022	.5000
.5500															.5500
.6000	-.108	.077	-.149	.056	-.147	.036	-.181	.037	.142	.047	-.126	.037	-.101	.022	.6000
.6500															.6500
.7000	-.122	.030	-.153	.043	-.155	.027	-.193	.020	.159	-.002	-.146	-.016	-.121	-.030	.7000
.7500															.7500
.8000	-.121	.001	-.145	.016	-.165	.018	-.195	.013	.167	-.006	-.153	-.040	-.122	-.059	.8000
.8500															.8500
.9000	-.108	.037	-.144	-.001	-.168	.015	-.195	.008							.9000
$\alpha = 1.0$															
.0125	-.091	.405	-.122	.439	-.099	.421	-.148	.409	-.118	.412	-.103	.371	-.059	.334	.0125
.0250	-.129	.328	-.143	.372	-.137	.364	-.163	.363	.118	.412	-.103	.371	-.059	.334	.0250
.0500	-.124	.270	-.150	.318	-.151	.321	-.151	.321	.123	.309	-.108	.302	-.078	.269	.0500
.0750	-.121	.204	-.154	.278	-.149	.275	-.138	.298	.123	.309	-.108	.302	-.078	.269	.0750
.1000	-.111	.209	-.160	.270	-.149	.259	-.137	.259	.129	.238	-.105	.240	-.090	.200	.1000
.1500	-.090	.219	-.164	.239	-.149	.234	-.142	.241	.129	.238	-.105	.240	-.090	.200	.1500
.2000	-.076	.216	-.174	.207	-.167	.216	-.149	.231	.136	.194	-.128	.209	-.105	.146	.2000
.2500	-.063	.176	-.167	.187	-.168	.195	-.155	.194	.136	.194	-.128	.209	-.105	.146	.2500
.3000	-.066	.202	-.151	.174	-.176	.171	-.160	.171	.148	.153	-.136	.156	-.116	.121	.3000
.3500															.3500
.4000	-.125	.152	-.123	.133	-.187	.137	-.172	.147	.159	.113	-.144	.125	-.131	.085	.4000
.4500															.4500
.5000	-.121	.140	-.144	.112	-.200	.085	-.195	.090	.174	.079	-.161	.071	-.137	.045	.5000
.5500															.5500
.6000	-.124	.119	-.164	.090	-.203	.068	-.206	.075	.184	.058	-.169	.046	-.147	.035	.6000
.6500															.6500
.7000	-.136	.061	-.168	.076	-.186	.059	-.215	.054	.188	.033	-.178	.019	-.147	.001	.7000
.7500															.7500
.8000	-.130	.027	-.159	.046	-.182	.053	-.218	.047	.195	.026	-.181	-.005	-.150	-.031	.8000
.8500															.8500
.9000	-.121	.070	-.157	.023	-.183	.048	-.221	.040							.9000

TABLE VI.- Continued  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b) M = 2.01 - Continued

X/C		FRACTION OF SEMISPAN														X/C	
		0.050		0.200		0.350		0.500		0.700		0.825		0.950			
		U	L	U	L	U	L	U	L	U	L	U	L	U	L		
		$\alpha = 12$															
.0125	-.129	.444	-.168	.500	-.157	.488	-.197	.458	-.180	.469	-.172	.429	-.151	.390	.0125		
.0250	-.165	.401	-.183	.436	-.179	.426	-.208	.424							.0250		
.0500	-.168	.317	-.187	.381	-.190	.381	-.200	.385							.0500		
.0750	-.156	.256	-.191	.338	-.186	.327	-.189	.351	-.176	.369	-.163	.357	-.154	.324	.0750		
.1000	-.153	.252	-.194	.323	-.184	.314	-.184	.313							.1000		
.1500	-.105	.265	-.202	.288	-.289	.289	-.185	.302	-.176	.297	-.152	.288	-.151	.246	.1500		
.2000	-.099	.254	-.208	.260	-.198	.268	-.189	.283							.2000		
.2500	-.088	.223	-.202	.241	-.196	.245	-.191	.242	-.179	.232	-.172	.243	-.155	.200	.2500		
.3000	-.072	.245	-.192	.223	-.202	.219	-.197	.223							.3000		
.3500									-.186	.197	-.177	.213	-.162	.170	.3500		
.4000	-.149	.196	-.159	.177	-.212	.178	-.206	.190							.4000		
.4500									-.195	.156	-.183	.173	-.174	.127	.4500		
.5000	-.142	.184	-.160	.157	-.224	.126	-.224	.131							.5000		
.5500									-.206	.123	-.194	.113	-.179	.112	.5500		
.6000	-.142	.162	-.179	.138	-.234	.109	-.233	.116							.6000		
.6500									-.214	.106	-.203	.090	-.182	.080	.6500		
.7000	-.154	.094	-.186	.120	-.234	.101	-.241	.096							.7000		
.7500									-.218	.072	-.207	.063	-.182	.041	.7500		
.8000	-.145	.064	-.175	.086	-.215	.090	-.241	.087							.8000		
.8500									-.224	.067	-.213	.032	-.186	.012	.8500		
.9000	-.138	.107	-.172	.062	-.203	.082	-.245	.082							.9000		
$\alpha = 14$																	
.0125	-.157	.521	-.208	.545	-.214	.534	-.241	.497							.0125		
.0250	-.192	.461	-.218	.486	-.225	.465	-.244	.487	-.229	.513	-.221	.474	-.211	.433	.0250		
.0500	-.190	.366	-.218	.434	-.227	.432	-.235	.442							.0500		
.0750	-.183	.312	-.220	.387	-.223	.371	-.225	.398	-.215	.420	-.210	.404	-.207	.376	.0750		
.1000	-.187	.304	-.224	.377	-.221	.362	-.222	.369							.1000		
.1500	-.180	.313	-.227	.338	-.227	.344	-.218	.354	-.210	.344	-.190	.336	-.198	.290	.1500		
.2000	-.100	.300	-.232	.305	-.227	.322	-.219	.331							.2000		
.2500	-.126	.267	-.221	.293	-.227	.290	-.220	.290	-.208	.275	-.204	.294	-.198	.246	.2500		
.3000	-.074	.289	-.210	.261	-.233	.267	-.222	.272							.3000		
.3500									-.215	.242	-.209	.256	-.198	.210	.3500		
.4000	-.181	.239	-.198	.219	-.236	.221	-.230	.232							.4000		
.4500									-.221	.194	-.215	.214	-.204	.167	.4500		
.5000	-.153	.225	-.189	.196	-.248	.167	-.248	.173	-.231	.170	-.225	.156	-.211	.154	.5000		
.5500									-.235	.143	-.228	.129	-.212	.123	.5500		
.6000	-.157	.201	-.198	.175	-.256	.149	-.254	.157							.6000		
.6500									-.237	.111	-.231	.098	-.210	.080	.6500		
.7000	-.168	.130	-.205	.154	-.262	.141	-.261	.137							.7000		
.7500									-.237	.111	-.231	.098	-.210	.080	.7500		
.8000	-.157	.100	-.192	.119	-.255	.127	-.259	.127							.8000		
.8500									-.241	.108	-.236	.067	-.213	.046	.8500		
.9000	-.151	.144	-.189	.093	-.233	.120	-.263	.119							.9000		
$\alpha = 16$																	
.0125	-.181	.578	-.240	.588	-.252	.578	-.272	.533							.0125		
.0250	-.214	.507	-.245	.533	-.257	.509	-.272	.533	-.269	.548	-.256	.510	-.249	.477	.0250		
.0500	-.208	.412	-.243	.484	-.255	.483	-.262	.494							.0500		
.0750	-.205	.373	-.243	.458	-.250	.416	-.254	.443	-.254	.467	-.247	.453	-.242	.417	.0750		
.1000	-.209	.347	-.244	.428	-.248	.413	-.249	.424							.1000		
.1500	-.205	.362	-.245	.382	-.240	.400	-.246	.404	-.246	.394	-.220	.397	-.231	.336	.1500		
.2000	-.094	.349	-.244	.351	-.248	.366	-.245	.380							.2000		
.2500	-.131	.316	-.237	.337	-.249	.336	-.244	.337	-.239	.325	-.235	.346	-.228	.292	.2500		
.3000	-.073	.333	-.227	.304	-.253	.308	-.244	.320							.3000		
.3500									-.244	.281	-.238	.293	-.228	.254	.3500		
.4000	-.202	.283	-.224	.259	-.258	.265	-.249	.272							.4000		
.4500									-.248	.237	-.241	.252	-.231	.216	.4500		
.5000	-.163	.266	-.218	.235	-.266	.200	-.263	.217							.5000		
.5500									-.256	.211	-.249	.201	-.234	.200	.5500		
.6000	-.168	.242	-.221	.211	-.273	.189	-.269	.200	-.260	.183	-.253	.165	-.234	.165	.6000		
.6500									-.259	.152	-.255	.138	-.235	.115	.6500		
.7000	-.178	.169	-.225	.190	-.278	.182	-.274	.177							.7000		
.7500									-.259	.152	-.255	.138	-.235	.115	.7500		
.8000	-.167	.140	-.217	.153	-.277	.162	-.274	.166							.8000		
.8500									-.261	.147	-.245	.109	-.235	.081	.8500		
.9000	-.161	.186	-.217	.127	-.258	.155	-.276	.153							.9000		
$\alpha = 18$																	
.0125	-.215	.634	-.276	.634	-.287	.616	-.305	.564							.0125		
.0250	-.239	.546	-.278	.596	-.289	.558	-.303	.575	-.284	.582	-.286	.550	-.296	.520	.0250		
.0500	-.232	.457	-.271	.541	-.285	.534	-.295	.551							.0500		
.0750	-.228	.431	-.268	.510	-.278	.466	-.286	.495	-.278	.519	-.276	.515	-.286	.452	.0750		
.1000	-.230	.393	-.265	.477	-.274	.472	-.282	.472							.1000		
.1500	-.239	.414	-.264	.436	-.273	.417	-.275	.428	-.269	.448	-.242	.452	-.273	.386	.1500		
.2000	-.121	.399	-.262	.406	-.273	.417	-.275	.428							.2000		
.2500	-.144	.366	-.260	.386	-.271	.383	-.272	.390	-.262	.374	-.262	.397	-.268	.336	.2500		
.3000	-.089	.380	-.254	.357	-.274	.355	-.271	.368							.3000		
.3500									-.263	.332	-.261	.343	-.266	.303	.3500		
.4000	-.216	.330	-.248	.311	-.278	.317	-.275	.318							.4000		
.4500									-.266	.284	-.263	.299	-.266	.267	.4500		
.5000	-.178	.311	-.250	.282	-.284	.243	-.285	.256							.5000		
.5500									-.272	.257	-.270	.247	-.267	.245	.5500		
.6000	-.182	.285	-.238	.254	-.286	.231	-.289	.247	-.275	.234	-.271	.209	-.262	.204	.6000		
.6500									-.274	.199	-.273	.182	-.258	.159	.6500		
.7000	-.193	.210	-.243	.232	-.286	.222	-.292	.221							.7000		
.7500									-.279	.195	-.263	.149	-.253	.127	.7500		
.8000	-.180	.181	-.232	.194	-.281	.203	-.290	.207							.8000		
.8500									-.279	.195	-.263	.149	-.253	.127	.8500		
.9000	-.176	.228	-.228	.171	-.268	.194	-.258	.200							.9000		

TABLE VI.- Concluded  
PRESSURE COEFFICIENTS FOR LINEAR TWIST WING

[From reference 1]

(b) M = 2.01 - Concluded

X/C	FRACTION OF SEMISPAN														X/C
	0.050		0.200		0.350		0.500		0.700		0.825		0.950		
	U	L	U	L	U	L	U	L	U	L	U	L	U	L	
	$\alpha=20$														
.0125	-.230	.683	-.293	.676	-.305	.654	-.316	.589	-.294	.610	-.294	.582	-.306	.534	.0125
.0250	-.260	.587	-.294	.644	-.305	.603	-.317	.617							.0250
.0500	-.244	.503	-.288	.586	-.302	.585	-.307	.595							.0500
.0750	-.233	.491	-.282	.552	-.295	.522	-.302	.541	-.290	.566	-.291	.556	-.299	.498	.0750
.1000	-.244	.465	-.279	.528	-.292	.525	-.298	.520							.1000
.1500	-.258	.470	-.276	.491	-.292	.510	-.294	.498	-.285	.498	-.297	.497	-.289	.397	.1500
.2000	-.198	.457	-.275	.460	-.288	.468	-.290	.483							.2000
.2500	-.149	.421	-.274	.434	-.288	.436	-.288	.439	-.280	.419	-.274	.443	-.283	.387	.2500
.3000	-.109	.429	-.277	.403	-.290	.405	-.287	.414							.3000
.3500									-.277	.374	-.277	.388	-.280	.348	.3500
.4000	-.223	.381	-.275	.356	-.293	.372	-.288	.367	-.280	.332	-.280	.350	-.279	.322	.4000
.4500															.4500
.5000	-.190	.358	-.269	.327	-.296	.278	-.297	.302	-.283	.306	-.282	.291	-.279	.288	.5000
.5500															.5500
.6000	-.198	.331	-.266	.294	-.297	.278	-.300	.296	-.287	.277	-.280	.252	-.277	.243	.6000
.6500															.6500
.7000	-.204	.252	-.273	.271	-.296	.264	-.303	.266	-.285	.242	-.284	.221	-.275	.199	.7000
.7500															.7500
.8000	-.191	.226	-.270	.229	-.292	.248	-.299	.248	-.286	.234	-.281	.191	-.273	.167	.8000
.8500															.8500
.9000	-.187	.274	-.266	.215	-.271	.238	-.242	.241							.9000

TABLE VII

## SECTION AERODYNAMIC CHARACTERISTICS FOR CAMBERED AND TWISTED WING

(a)  $M = 1.61$ 

$\alpha$ , deg	$y/b$ 2						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-.991	-1.018	-.921	-.768	-.506	-.438	-.276
-18	-.878	-.910	-.852	-.705	-.435	-.382	-.224
-16	-.778	-.803	-.796	-.646	-.393	-.351	-.194
-14	-.673	-.695	-.734	-.603	-.355	-.312	-.174
-12	-.566	-.590	-.651	-.538	-.319	-.282	-.155
-10	-.468	-.487	-.545	-.482	-.283	-.252	-.133
-08	-.372	-.392	-.435	-.416	-.244	-.221	-.111
-06	-.275	-.275	-.319	-.334	-.194	-.182	-.089
-04	-.184	-.183	-.210	-.250	-.146	-.143	-.076
-02	-.090	-.093	-.112	-.162	-.098	-.104	-.067
00	.001	.012	-.011	-.057	-.043	-.061	-.045
02	.102	.119	.093	.050	.019	-.010	-.016
04	.200	.223	.196	.157	.090	.049	.019
06	.292	.319	.299	.262	.158	.113	.063
08	.398	.429	.412	.364	.228	.189	.117
10	.493	.527	.498	.451	.272	.246	.155
12	.592	.626	.589	.516	.322	.291	.192
14	.692	.732	.686	.590	.370	.343	.231
16	.811	.850	.798	.681	.424	.417	.294
18	.947	.990	.912	.785	.509	.476	.315
20	1.086	1.139	1.036	.902	.552	.494	.319
$C_m$							
-20	-.393	-.116	.139	.303	.403	.401	.302
-18	-.369	-.133	.121	.272	.344	.345	.246
-16	-.338	-.139	.109	.244	.309	.315	.211
-14	-.307	-.141	.095	.227	.280	.277	.188
-12	-.273	-.140	.074	.198	.252	.248	.167
-10	-.242	-.135	.039	.174	.224	.220	.142
-08	-.213	-.129	.006	.144	.192	.192	.117
-06	-.181	-.117	-.019	.107	.152	.155	.092
-04	-.147	-.110	-.042	.070	.113	.120	.077
-02	-.116	-.099	-.059	.030	.075	.084	.068
00	-.086	-.085	-.073	-.017	.030	.044	.044
02	-.052	-.072	-.086	-.064	-.021	-.001	.012
04	-.019	-.058	-.097	-.109	-.079	-.056	-.026
06	.009	-.045	-.107	-.148	-.137	-.115	-.074
08	.044	-.030	-.119	-.182	-.199	-.183	-.133
10	.074	-.017	-.122	-.211	-.237	-.232	-.173
12	.104	-.003	-.126	-.227	-.276	-.271	-.213
14	.137	.010	-.139	-.252	-.312	-.317	-.255
16	.172	.019	-.160	-.285	-.350	-.383	-.322
18	.209	.030	-.173	-.325	-.418	-.430	-.344
20	.254	.045	-.195	-.374	-.454	-.446	-.347

TABLE VII.- Concluded

## SECTION AERODYNAMIC CHARACTERISTICS FOR CAMBERED AND TWISTED WING

(b)  $M = 2.01$ 

$\alpha$ , deg	$y/\frac{b}{2}$						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-.836	-.786	-.690	-.574	-.427	-.328	-.223
-18	-.755	-.740	-.657	-.544	-.404	-.307	-.208
-16	-.670	-.670	-.605	-.504	-.375	-.281	-.192
-14	-.581	-.581	-.550	-.460	-.344	-.260	-.175
-12	-.489	-.493	-.490	-.412	-.313	-.239	-.160
-10	-.411	-.414	-.423	-.363	-.282	-.222	-.148
-08	-.331	-.332	-.351	-.311	-.245	-.200	-.138
-06	-.247	-.251	-.284	-.243	-.209	-.177	-.128
-04	-.165	-.164	-.192	-.184	-.171	-.148	-.113
-02	-.049	-.043	-.070	-.087	-.102	-.102	-.084
00	-.011	-.005	-.028	-.047	-.078	-.085	-.073
02	.062	.070	.059	.028	-.025	-.055	-.052
04	.148	.158	.134	.097	.035	-.000	-.024
06	.221	.233	.205	.170	.091	.041	.013
08	.305	.318	.285	.237	.149	.097	.052
10	.387	.394	.363	.307	.198	.148	.094
12	.476	.486	.429	.378	.257	.196	.045
14	.566	.576	.509	.447	.309	.246	.058
16	.658	.668	.582	.509	.358	.293	.069
18	.747	.754	.653	.568	.404	.323	.081
20	.843	.847	.734	.632	.445	.359	.087
$C_m$							
-20	-.335	-.104	.083	.209	.293	.294	.243
-18	-.314	-.103	.077	.196	.276	.274	.226
-16	-.290	-.107	.067	.178	.254	.249	.208
-14	-.262	-.110	.055	.159	.230	.229	.189
-12	-.234	-.112	.041	.138	.207	.210	.172
-10	-.208	-.110	.025	.116	.184	.193	.158
-08	-.183	-.106	.010	.092	.157	.172	.147
-06	-.156	-.099	-.006	.061	.130	.150	.137
-04	-.127	-.092	-.026	.035	.103	.123	.119
-02	-.087	-.078	-.049	-.006	.051	.081	.086
00	-.075	-.074	-.056	-.024	.032	.065	.074
02	-.050	-.064	-.068	-.058	-.008	.036	.050
04	-.023	-.055	-.081	-.085	-.052	-.015	.019
06	.001	-.046	-.088	-.112	-.091	-.053	-.023
08	.028	-.036	-.100	-.134	-.129	-.103	-.063
10	.054	-.024	-.108	-.158	-.160	-.146	-.108
12	.081	-.014	-.112	-.181	-.196	-.186	-.054
14	.112	-.003	-.115	-.200	-.230	-.229	-.067
16	.142	.006	-.120	-.218	-.261	-.270	-.079
18	.169	.015	-.128	-.238	-.291	-.296	-.091
20	.200	.023	-.137	-.261	-.317	-.327	-.097

TABLE VIII  
SECTION AERODYNAMIC CHARACTERISTICS FOR REFLEX CAMBERED WING

(a)  $M = 1.61$

$\alpha$ , deg	$y/\frac{b}{2}$						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-.972	-1.036	-.947	-.796	-.609	-.484	-.302
-18	-.862	-.932	-.882	-.738	-.563	-.442	-.274
-16	-.748	-.826	-.806	-.676	-.511	-.400	-.260
-14	-.638	-.713	-.711	-.601	-.453	-.352	-.234
-12	-.538	-.604	-.614	-.528	-.395	-.296	-.192
-10	-.436	-.498	-.495	-.449	-.329	-.244	-.168
-08	-.340	-.396	-.380	-.355	-.273	-.197	-.134
-06	-.250	-.292	-.266	-.239	-.199	-.163	-.108
-04	-.155	-.193	-.168	-.142	-.112	-.095	-.064
-02	-.060	-.091	-.007	.027	-.026	-.015	-.009
00	.036	.013	.044	.055	.064	.063	.041
02	.121	.110	.130	.145	.149	.113	.081
04	.206	.203	.231	.232	.204	.163	.116
06	.297	.298	.324	.310	.261	.211	.150
08	.384	.395	.412	.389	.318	.255	.184
10	.481	.499	.507	.468	.380	.298	.211
12	.580	.596	.603	.541	.431	.335	.235
14	.706	.729	.728	.632	.477	.377	.252
16	.844	.865	.848	.716	.497	.356	.228
18	.950	.955	.937	.765	.512	.355	.222
20	1.071	1.124	1.008	.811	.541	.378	.233
$C_m$							
-20	-.295	-.060	.193	.345	.444	.444	.331
-18	-.266	-.063	.178	.321	.411	.406	.301
-16	-.232	-.062	.161	.295	.375	.368	.286
-14	-.199	-.056	.138	.262	.333	.324	.259
-12	-.168	-.049	.114	.231	.289	.272	.210
-10	-.135	-.037	.085	.195	.240	.222	.185
-08	-.105	-.026	.062	.151	.199	.179	.148
-06	-.076	-.012	.044	.095	.136	.146	.118
-04	-.047	-.001	.031	.055	.073	.081	.067
-02	-.017	.009	.019	-.003	.013	.011	.009
00	.017	.025	.009	-.015	-.046	-.056	-.043
02	.044	.040	.001	-.044	-.098	-.098	-.086
04	.073	.052	-.009	-.070	-.135	-.141	-.122
06	.105	.065	-.017	-.096	-.172	-.182	-.159
08	.134	.078	-.026	-.123	-.208	-.220	-.196
10	.168	.092	-.036	-.148	-.252	-.258	-.226
12	.201	.104	-.047	-.175	-.288	-.293	-.252
14	.235	.114	-.068	-.210	-.319	-.332	-.271
16	.281	.134	-.088	-.249	-.334	-.312	-.246
18	.322	.157	-.107	-.273	-.344	-.311	-.239
20	.370	.149	-.123	-.292	-.366	-.333	-.251

TABLE VIII.- Concluded

## SECTION AERODYNAMIC CHARACTERISTICS FOR REFLEX CAMBERED WING

(b)  $M = 2.01$ 

$\alpha$ , deg	$y/\frac{b}{2}$						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-.823	-.815	-.713	-.593	-.447	-.356	-.263
-18	-.732	-.747	-.659	-.550	-.412	-.320	-.236
-16	-.650	-.675	-.601	-.499	-.375	-.295	-.210
-14	-.559	-.588	-.521	-.448	-.337	-.259	-.184
-12	-.477	-.497	-.451	-.395	-.297	-.227	-.159
-10	-.395	-.401	-.381	-.336	-.251	-.194	-.134
-08	-.311	-.316	-.291	-.269	-.204	-.156	-.108
-06	-.223	-.229	-.201	-.181	-.150	-.121	-.081
-04	-.154	-.159	-.140	-.117	-.103	-.087	-.057
-02	-.069	-.075	-.059	-.041	-.043	-.044	-.027
00	-.003	.004	.017	.031	.015	.002	.006
02	.079	.086	.100	.107	.071	.055	.046
04	.152	.162	.170	.177	.132	.101	.085
06	.226	.235	.250	.237	.192	.147	.113
08	.304	.321	.326	.311	.243	.183	.145
10	.396	.403	.387	.377	.288	.220	.168
12	.478	.485	.472	.434	.330	.247	.188
14	.565	.569	.544	.488	.370	.272	.204
16	.651	.647	.606	.539	.404	.294	.219
18	.747	.739	.687	.591	.434	.313	.231
20	.840	.832	.737	.639	.462	.327	.240
$C_m$							
-20	-.260	-.047	.132	.249	.326	.328	.292
-18	-.234	-.047	.119	.229	.299	.294	.262
-16	-.210	-.046	.105	.204	.269	.270	.233
-14	-.182	-.045	.087	.180	.240	.236	.204
-12	-.155	-.040	.071	.156	.209	.205	.174
-10	-.128	-.034	.055	.131	.175	.174	.147
-08	-.102	-.028	.037	.100	.139	.139	.118
-06	-.075	-.024	.022	.060	.100	.106	.087
-04	-.051	-.013	.013	.033	.066	.075	.061
-02	-.023	-.002	.003	.004	.022	.036	.028
00	.000	.007	-.006	-.022	-.018	-.005	-.009
02	.027	.019	-.014	-.048	-.056	-.052	-.052
04	.054	.027	-.023	-.071	-.094	-.089	-.092
06	.081	.039	-.029	-.088	-.131	-.128	-.121
08	.106	.049	-.036	-.112	-.164	-.157	-.155
10	.139	.062	-.040	-.133	-.192	-.189	-.180
12	.164	.072	-.049	-.151	-.220	-.212	-.202
14	.197	.082	-.056	-.170	-.248	-.235	-.219
16	.224	.091	-.062	-.186	-.271	-.254	-.235
18	.255	.100	-.075	-.206	-.291	-.271	-.249
20	.286	.102	-.080	-.224	-.311	-.284	-.259



TABLE IX  
SECTION AERODYNAMIC CHARACTERISTICS FOR FLAT WING

[From reference 4]

(a)  $M = 1.61$

$\alpha$ , deg	$y/\frac{b}{2}$					
	.05	.20	.35	.50	.70	.90
$C_n$						
-20	-1.033	-1.087	-1.019	-.888	-.652	-.346
-18	-.889	-.935	-.903	-.791	-.611	-.388
-16	-.770	-.802	-.794	-.686	-.532	-.375
-14	-.653	-.678	-.684	-.611	-.463	-.323
-12	-.554	-.581	-.588	-.533	-.411	-.274
-10	-.465	-.489	-.494	-.468	-.365	-.238
-08	-.367	-.388	-.392	-.376	-.309	-.202
-06	-.272	-.286	-.295	-.283	-.241	-.163
-04	-.166	-.179	-.187	-.179	-.151	-.113
-02	-.084	-.094	-.096	-.091	-.077	-.054
00	.000	.000	.000	.000	.000	.000
02	.084	.094	.096	.091	.077	.054
04	.166	.179	.187	.179	.151	.113
06	.272	.286	.295	.283	.241	.163
08	.367	.388	.392	.376	.309	.202
10	.465	.489	.494	.468	.365	.238
12	.554	.581	.588	.533	.411	.274
14	.653	.678	.684	.611	.463	.323
16	.770	.802	.794	.686	.532	.375
18	.889	.935	.903	.791	.611	.388
20	1.033	1.087	1.019	.888	.652	.346
$C_m$						
-20	-.319	-.079	.174	.356	.458	-.354
-18	-.284	-.082	.143	.313	.432	-.394
-16	-.254	-.086	.115	.262	.375	-.382
-14	-.220	-.085	.085	.230	.323	-.330
-12	-.187	-.076	.066	.194	.284	-.278
-10	-.158	-.066	.050	.169	.253	-.241
-08	-.125	-.052	.037	.130	.213	-.204
-06	-.094	-.039	.027	.095	.164	-.164
-04	-.059	-.025	.015	.058	.100	-.113
-02	-.031	-.013	.009	.029	.051	-.054
00	.000	.000	.000	.000	.000	.000
02	.031	.013	-.009	-.029	-.051	.054
04	.059	.025	-.015	-.058	-.100	.113
06	.094	.039	-.027	-.095	-.164	.164
08	.125	.052	-.037	-.130	-.213	.204
10	.158	.066	-.050	-.169	-.253	.241
12	.187	.076	-.066	-.194	-.284	.278
14	.220	.085	-.085	-.230	-.323	.330
16	.254	.086	-.115	-.262	-.375	.382
18	.284	.082	-.143	-.313	-.432	.394
20	.319	.079	-.174	-.356	-.458	.354

TABLE IX.- Concluded

## SECTION AERODYNAMIC CHARACTERISTICS FOR FLAT WING

[From reference 4]

(b)  $M = 2.01$ 

$\alpha$ , deg	$y/\frac{b}{2}$					
	.05	.20	.35	.50	.70	.90
$C_n$						
-20	-.830	-.814	-.732	-.615	-.463	-.301
-18	-.741	-.742	-.668	-.564	-.427	-.279
-16	-.648	-.661	-.604	-.519	-.392	-.254
-14	-.558	-.567	-.538	-.464	-.349	-.226
-12	-.475	-.485	-.464	-.406	-.309	-.200
-10	-.394	-.407	-.396	-.351	-.265	-.174
-08	-.310	-.321	-.312	-.282	-.216	-.143
-06	-.232	-.240	-.232	-.213	-.166	-.109
-04	-.151	-.159	-.156	-.143	-.114	-.076
-02	-.074	-.078	-.073	-.069	-.056	-.038
00	.000	.000	.000	.000	.000	.000
02	.074	.078	.073	.069	.056	.038
04	.151	.159	.156	.143	.114	.076
06	.232	.240	.232	.213	.166	.109
08	.310	.321	.312	.282	.216	.143
10	.394	.407	.396	.351	.265	.174
12	.475	.485	.464	.406	.309	.200
14	.558	.567	.538	.464	.349	.226
16	.648	.661	.604	.519	.392	.254
18	.741	.742	.668	.564	.427	.279
20	.830	.814	.732	.615	.463	.301
$C_m$						
-20	-.276	-.075	.101	.231	.322	-.306
-18	-.248	-.073	.090	.210	.297	-.283
-16	-.217	-.068	.083	.196	.272	-.257
-14	-.189	-.064	.072	.174	.241	-.228
-12	-.162	-.058	.059	.151	.213	-.202
-10	-.135	-.050	.048	.130	.182	-.175
-08	-.108	-.041	.035	.103	.147	-.144
-06	-.081	-.031	.026	.076	.112	-.110
-04	-.054	-.020	.016	.050	.077	-.076
-02	-.028	-.010	.007	.024	.038	-.038
00	.000	.000	.000	.000	.000	.000
02	.028	.010	-.007	-.024	-.038	.038
04	.054	.020	-.016	-.050	-.077	.076
06	.081	.031	-.026	-.076	-.112	.110
08	.108	.041	-.035	-.103	-.147	.144
10	.135	.050	-.048	-.130	-.182	.175
12	.162	.058	-.059	-.151	-.213	.202
14	.189	.064	-.072	-.174	-.241	.228
16	.217	.068	-.083	-.196	-.272	.257
18	.248	.073	-.090	-.210	-.297	.283
20	.276	.075	-.101	-.231	-.322	.306

L-3000

TABLE X  
SECTION AERODYNAMIC CHARACTERISTICS FOR CAMBERED WING

(a)  $M = 1.61$

$\alpha$ , deg	$y/\frac{b}{2}$					
	.05	.20	.35	.50	.70	.90
$C_n$						
-20	-.996	-1.036	-.934	-.783	-.576	-.323
-18	-.877	-.900	-.838	-.682	-.505	-.283
-16	-.762	-.767	-.752	-.600	-.421	-.232
-14						
-12	-.557	-.543	-.586	-.457	-.310	-.150
-10	-.446	-.430	-.483	-.382	-.256	-.129
-08	-.350	-.329	-.361	-.317	-.206	-.115
-06	-.252	-.228	-.249	-.231	-.150	-.087
-04	-.159	-.128	-.149	-.144	-.094	-.055
-02	-.068	-.033	-.046	-.054	-.034	-.021
00	.029	.071	.061	.055	.040	.023
02	.116	.166	.159	.152	.118	.074
04	.214	.271	.266	.256	.200	.139
06	.308	.363	.360	.341	.271	.189
08	.421	.468	.464	.439	.345	.238
10	.518	.565	.558	.505	.402	.276
12	.622	.666	.648	.579	.461	.326
14	.726	.766	.739	.655	.518	.384
16	.841	.883	.842	.743	.607	.424
18	.965	1.009	.955	.859	.674	.420
20	1.125	1.175	1.105	.967	.692	.431
$C_m$						
-20	-.394	-.106	.148	.313	.407	.328
-18	-.370	-.121	.124	.265	.357	.288
-16	-.341	-.135	.103	.226	.291	.236
-14						
-12	-.276	-.139	.059	.157	.205	.146
-10	-.240	-.135	.030	.124	.165	.125
-08	-.211	-.127	-.007	.098	.129	.111
-06	-.178	-.118	-.034	.060	.089	.081
-04	-.143	-.106	-.053	.022	.048	.049
-02	-.114	-.095	-.069	-.018	.006	.015
00	-.084	-.082	-.084	-.066	-.049	-.032
02	-.057	-.070	-.096	-.106	-.106	-.084
04	-.025	-.056	-.107	-.145	-.164	-.151
06	.005	-.043	-.116	-.174	-.212	-.200
08	.042	-.030	-.125	-.208	-.260	-.248
10	.078	-.017	-.133	-.223	-.297	-.286
12	.107	-.002	-.138	-.247	-.336	-.336
14	.143	.010	-.151	-.276	-.372	-.397
16	.176	.017	-.170	-.310	-.438	-.433
18	.206	.019	-.192	-.359	-.481	-.428
20	.243	.018	-.218	-.394	-.489	-.413

TABLE X.- Concluded

## SECTION AERODYNAMIC CHARACTERISTICS FOR CAMBERED WING

(b)  $M = 2.01$ 

$\alpha$ , deg	$y/\frac{b}{2}$					
	.05	.20	.35	.50	.70	.90
$C_n$						
-20	-.795	-.774	-.674	-.537	-.390	-.234
-18	-.736	-.705	-.618	-.495	-.360	-.215
-16	-.648	-.624	-.566	-.452	-.330	.000
-14	-.552	-.533	-.501	-.398	-.294	-.178
-12	-.469	-.445	-.434	-.347	-.257	-.166
-10	-.376	-.365	-.366	-.294	-.222	-.151
-08	-.298	-.281	-.301	-.238	-.188	-.129
-06	-.222	-.207	-.219	-.178	-.147	-.103
-04	-.143	-.123	-.136	-.109	-.101	-.074
-02	-.073	-.044	-.056	-.041	-.053	-.045
00	.009	.034	.025	.030	-.001	-.012
02	.090	.115	.104	.102	.056	.022
04	.173	.198	.187	.173	.117	.071
06	.250	.270	.257	.238	.172	.113
08	.337	.355	.337	.308	.231	.158
10	.421	.435	.415	.374	.288	.199
12	.521	.525	.494	.447	.343	.236
14	.611	.612	.561	.505	.391	.267
16	.697	.701	.636	.565	.435	.296
18	.803	.794	.714	.629	.481	.322
20	.895	.872	.784	.685	.515	.342
$C_m$						
-20	-.339	-.096	.079	.190	.266	.235
-18	-.316	-.100	.066	.172	.243	.214
-16	-.274	-.108	.055	.153	.221	.000
-14	-.258	-.112	.041	.130	.193	.176
-12	-.232	-.111	.027	.109	.166	.163
-10	-.203	-.107	.013	.086	.141	.148
-08	-.177	-.102	-.002	.061	.116	.125
-06	-.150	-.095	-.019	.035	.086	.097
-04	-.123	-.086	-.037	.005	.052	.068
-02	-.100	-.078	-.052	-.025	.016	.037
00	-.073	-.069	-.066	-.056	-.022	.003
02	-.044	-.058	-.077	-.085	-.065	-.033
04	-.018	-.049	-.088	-.112	-.108	-.083
06	.007	-.040	-.095	-.134	-.144	-.124
08	.036	-.032	-.106	-.158	-.181	-.167
10	.063	-.023	-.114	-.179	-.218	-.207
12	.101	-.013	-.119	-.200	-.253	-.244
14	.123	-.003	-.121	-.216	-.284	-.275
16	.150	.008	-.128	-.237	-.312	-.304
18	.183	.017	-.137	-.259	-.343	-.330
20	.211	.024	-.147	-.279	-.366	-.351

TABLE XI  
SECTION AERODYNAMIC CHARACTERISTICS FOR LINEAR TWIST WING

[From reference 4]

(a)  $M = 1.61$

$\alpha$ , deg	$y/\frac{b}{2}$						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-1.106	-1.165	-1.071	-.906	-.641	-.454	-.288
-18	-.949	-.989	-.956	-.827	-.601	-.453	-.285
-16	-.823	-.852	-.859	-.756	-.557	-.430	-.289
-14	-.707	-.730	-.764	-.678	-.506	-.420	-.298
-12	-.605	-.632	-.653	-.610	-.471	-.387	-.282
-10	-.502	-.534	-.556		-.428	-.359	-.267
-08	-.405	-.431	-.458	-.451	-.392	-.329	-.245
-06	-.313	-.334	-.364	-.361	-.339	-.291	-.220
-04	-.212	-.231	-.261	-.263	-.257	-.241	-.186
-02	-.116	-.130	-.160	-.166	-.174	-.180	-.150
00	-.024	-.038	-.066	-.075	-.097	-.111	-.104
02	.049	.049	.017	.006	-.024	-.051	-.057
04	.142	.147	.114	.102	.051	.018	-.006
06	.236	.242	.217	.195	.132	.087	.035
08	.326	.332	.307	.285	.209	.143	.073
10	.428	.441	.410	.384	.277	.197	.113
12	.522	.541	.507	.467	.333	.245	.148
14	.625	.643	.617	.547	.398	.298	.195
16	.712	.729	.707	.623	.458	.370	.254
18	.844	.879	.822	.722	.545	.428	.259
20	.966	.969	.939	.827	.607	.467	.297
$C_m$							
-20	-.323	-.079	.189	.368	.454	.410	.316
-18	-.288	-.084	.154	.336	.430	.409	.312
-16	-.263	-.089	.128	.298	.401	.392	.317
-14	-.228	-.090	.102	.255	.355	.380	.327
-12	-.195	-.079	.073	.224	.327	.346	.308
-10	-.161	-.068	.058		.296	.321	.291
-08	-.129	-.054	.044	.156	.272	.294	.267
-06	-.100	-.041	.034	.121	.234	.260	.238
-04	-.067	-.027	.024	.085	.172	.214	.201
-02	-.034	-.015	.015	.053	.115	.158	.161
00	-.004	-.001	.005	.023	.064	.097	.110
02	.023	.010	-.001	-.004	.015	.045	.060
04	.055	.024	-.010	-.037	-.033	-.014	.007
06	.086	.038	-.020	-.067	-.087	-.075	-.038
08	.117	.050	-.028	-.099	-.143	-.125	-.078
10	.152	.065	-.039	-.138	-.190	-.173	-.121
12	.183	.076	-.053	-.170	-.229	-.216	-.160
14	.215	.085	-.074	-.205	-.277	-.265	-.213
16	.241	.090	-.097	-.240	-.321	-.336	-.277
18	.273	.079	-.127	-.284	-.385	-.382	-.278
20	.302	.072	-.156	-.333	-.427	-.416	-.323

TABLE XI.- Concluded  
SECTION AERODYNAMIC CHARACTERISTICS FOR LINEAR TWIST WING

[From reference 4]

(b)  $M = 2.01$

$\alpha$ , deg	$y/\frac{b}{2}$						
	.05	.20	.35	.50	.70	.825	.95
$C_n$							
-20	-.869	-.879	-.792	-.676	-.559	-.391	-.282
-18	-.771	-.792	-.745	-.628	-.466	-.373	-.269
-16	-.675	-.686	-.671	-.575	-.437	-.356	-.261
-14	-.587	-.602	-.605	-.527	-.408	-.335	-.248
-12	-.500	-.518	-.538	-.476	-.374	-.312	-.234
-10	-.410	-.431	-.450	-.416	-.334	-.279	
-08	-.331	-.355	-.375	-.354	-.294	-.249	-.192
-06	-.241	-.268	-.290	-.278	-.244	-.214	-.168
-04	-.172	-.190	-.222	-.212	-.197	-.179	-.145
-02	-.095	-.116	-.139	-.140	-.143	-.139	-.118
00	-.023	-.035	-.063	-.072	-.086	-.095	-.086
02	.048	.042	.012	.010	-.026	-.046	-.051
04	.125	.124	.092	.074	.032	.004	-.015
06	.196	.199	.173	.144	.084	.046	.014
08	.274	.278	.249	.215	.139	.091	.045
10	.351	.352	.324	.278	.186	.132	.074
12	.438	.440	.402	.346	.239	.174	.107
14	.524	.517	.479	.405	.283	.210	.133
16	.601	.594	.544	.459	.327	.243	.157
18	.688	.677	.609	.514	.370	.277	.184
20	.778	.762	.671	.561	.405	.305	.204
$C_m$							
-20	-.284	-.079	.114	.258	.389	.353	.311
-18	-.253	-.074	.105	.236	.328	.336	.295
-16	-.222	-.076	.092	.213	.307	.320	.287
-14	-.193	-.069	.082	.195	.285	.301	.271
-12	-.165	-.062	.071	.176	.261	.280	.256
-10	-.136	-.053	.053	.153	.231	.249	
-08	-.109	-.044	.042	.128	.202	.222	.208
-06	-.078	-.032	.031	.098	.167	.190	.182
-04	-.055	-.023	.023	.072	.135	.158	.157
-02	-.029	-.012	.013	.046	.097	.123	.128
00	-.005	-.003	.004	.022	.058	.084	.093
02	.021	.007	-.003	-.010	.017	.041	.055
04	.048	.020	-.011	-.027	-.022	-.003	.016
06	.073	.029	-.021	-.053	-.057	-.040	-.015
08	.100	.039	-.029	-.080	-.094	-.079	-.048
10	.127	.049	-.039	-.103	-.126	-.115	-.079
12	.155	.059	-.050	-.129	-.163	-.152	-.115
14	.185	.066	-.063	-.151	-.193	-.184	-.144
16	.209	.070	-.074	-.172	-.225	-.214	-.170
18	.236	.077	-.082	-.192	-.255	-.245	-.200
20	.265	.078	-.092	-.209	-.281	-.271	-.221

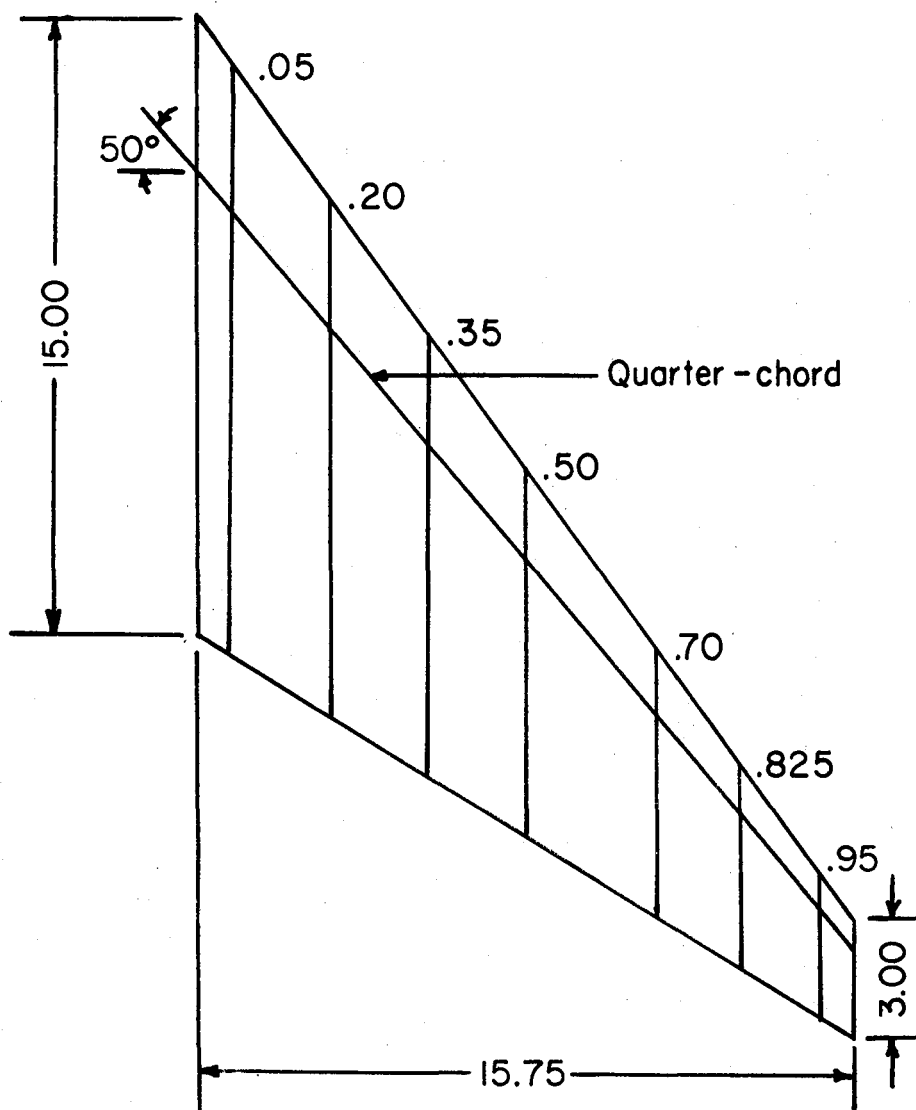


Figure 1.- Plan view of wings showing orifice stations. (Lengths are given in inches; stations are given in fractions of semispan.)